

**BHUBANANANDA ODISHA SCHOOL OF ENGINEERING, CUTTACK**

**DEPARTMENT OF CIVIL ENGINEERING**



**LECTURE NOTE ON: STRENGTH OF MATERIALS**

**(TH-1) 3<sup>TH</sup> SEMESTER**

**PREPARED BY: DR. SANTOSH KUMAR NAYAK**

**SR. LECTURER IN CIVIL ENGINEERING DEPT**

**BOSE, CUTTACK**

# Simple stresses and strains

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## Simple Stresses and strains

### Stress:

Stress of a material is defined as the resistance offered by the material per unit cross sectional area of the member subjected to external force.

Mathematically, stress is expressed as

$$\text{Stress} = \frac{\text{Resistance}}{\text{Cross sectional area}} = \frac{\text{Force}}{C/S \text{ area}}$$

$$\sigma = \frac{R}{A} = \frac{F}{A}$$

Where,

$F$  = externally applied force, and

$A$  = Cross sectional area of the member

Unit of stress in S.I system of units is  $\text{N}/\text{mm}^2$ .

Depending upon the nature of the force, stresses are of various types. Force acting on a structural member may be *perpendicular* to the plane or *tangential* to the plane. The force may tend to bend the plane about any of the axis on the plane or it may twist the plane about the axis perpendicular to the plane.

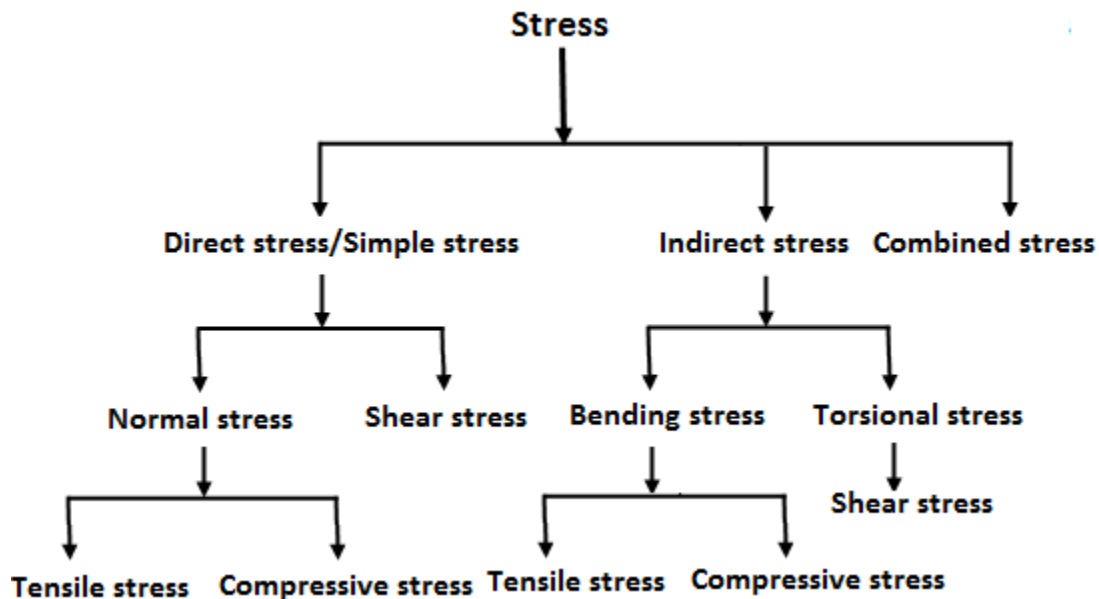
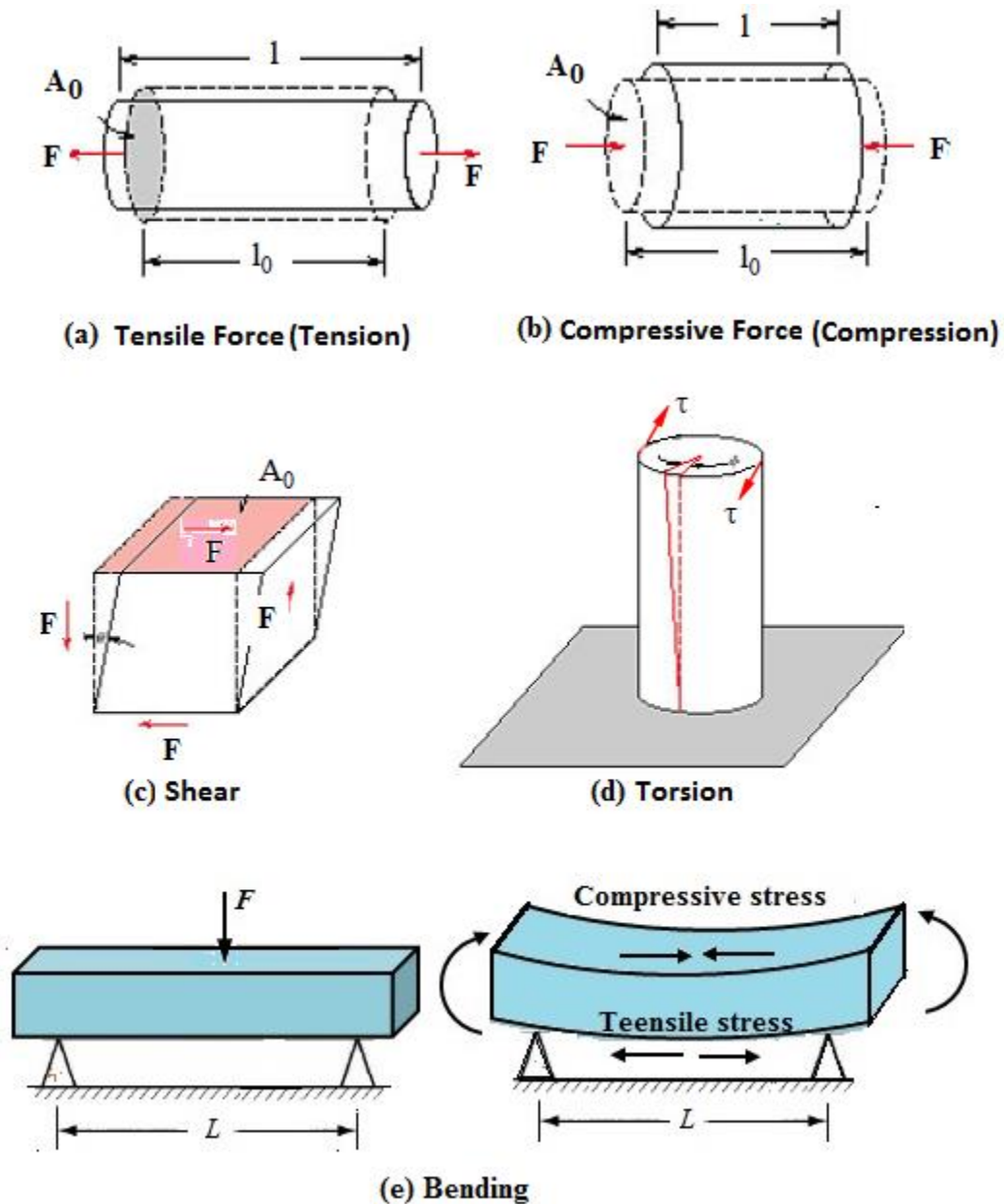


Fig. Stress Types

# Simple stresses and strains

**Strain:** *Strain* is the measure of deformation produced in the structural member by the applied load.

It is expressed as the ratio of change in dimension to the original dimension of the member before application of load. It is denoted by ' $\epsilon$ ' (Greek small letter epsilon). Strain is a dimensionless quantity as it is the ratio of same units.



**Figure. Types of Loading**

## Simple stresses and strains

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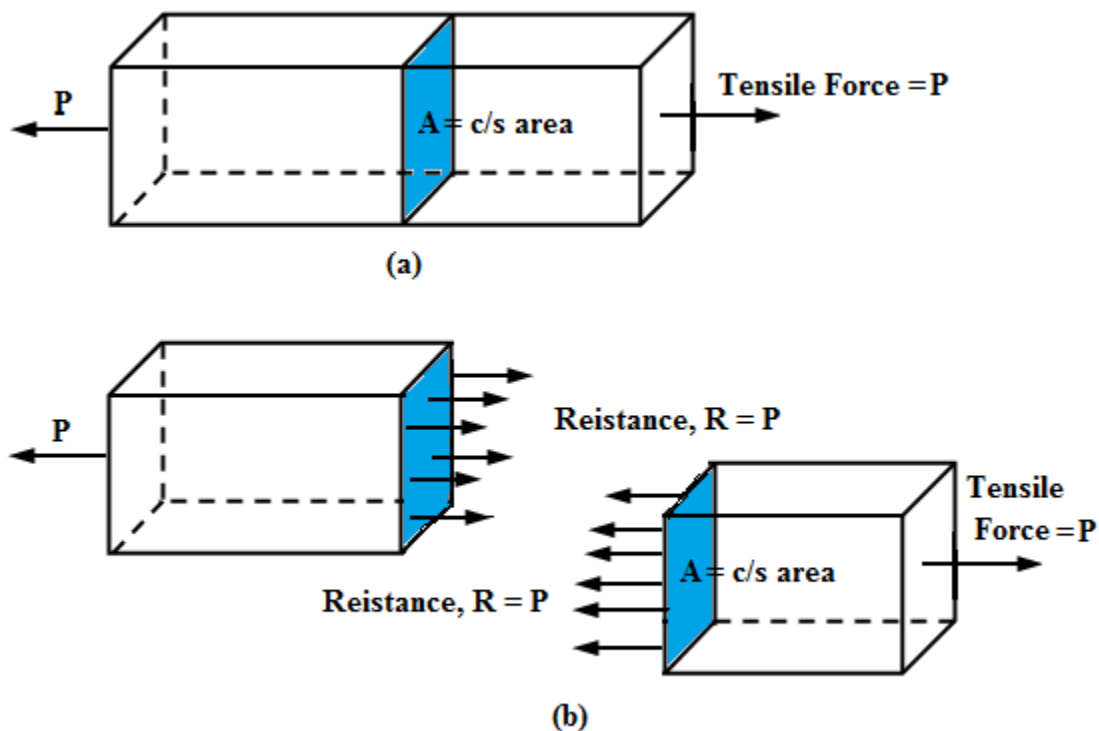
The dimension may be length, area, volume or the change in angle. The nomenclature of the strain follows according to the dimension.

**Direct Stress:** Direct stress also called *simple stress* is induced due to direct loading condition on the plane. It is of two types.

- a. Normal stress
- b. Tangential stress or shear stress

**Normal stress:** Stress developed on a plane subjected to axial loading (load perpendicular to the plane) is called *normal stress*. Normal stress is of two types.

- a. Tensile stress
- b. Compressive stress



**Fig. Tensile stress**

**Tensile stress:** The stress induced on a plane of a body subjected to axial tensile load (two equal and opposite pulls) is known as *tensile stress*.

## Simple stresses and strains

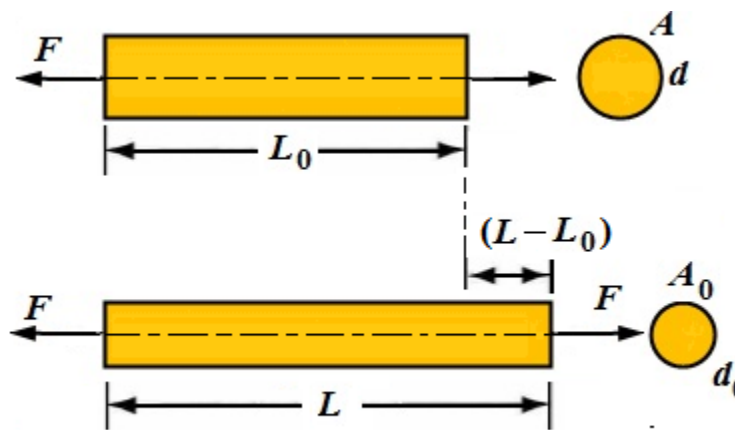
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When a body of uniform cross section e.g., a steel rod is subjected to an axial pull or tensile force, it has a tendency to elongate. The stress induced at any cross section of the rod is known as *tensile stress*.

It is the resistance of the material of the member per unit area of cross section normal to the load.

$$\text{Tensile stress} = \frac{\text{Resistance}}{\text{Original cross sectional area}} = \frac{\text{Tensile Force}}{\text{C/S area}}$$

$$\sigma = \frac{R}{A_0} = \frac{F}{A_0}$$



**Fig. Axial Tensile Load**

**Tensile strain:** It is the ratio of increase in length to the original length of the member subjected to axial tensile force.

$$\epsilon_l = \frac{\delta L}{L_0}$$

Where,

$L_0$  = Original length of the member before being subjected to load

$L$  = Final length due to the applied load

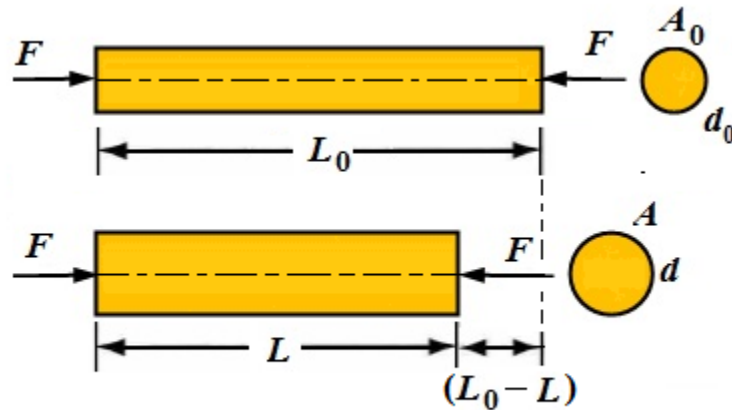
$\delta L = L - L_0$  = increase in length

**Compressive stress:** When a structural member is subjected to compressive force, it has a tendency to decrease in length. The stress induced in the member by virtue of the resistance offered by the material of the member is known as *compressive stress*.

*The stress induced in the material of the member by virtue of the resistance offered by it against decrease in length due to axial compressive load is known as compressive stress.*

## Simple stresses and strains

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**Fig. Axial Compressive Load**

$$\text{Compressive stress} = \frac{\text{Resistance}}{\text{Original cross sectional area}} = \frac{\text{Compressive Force}}{C/S \text{ area}}$$

$$\sigma = \frac{R}{A_0} = \frac{F}{A_0}$$

**Compressive strain:** It is the ration of decrease in length to the original length of the member subjected to axial compressive force.

**Lateral strain:** It is the ratio of change in lateral dimension to the original lateral dimension.  
Strain

$$\text{Lateral strain, } \epsilon_d = \frac{\text{Change in lateral dimension}}{\text{Original lateral dimension}} = \frac{\delta d}{d} = \frac{d - d_0}{d_0}$$

Strains in the direction transverse to the direction of load are called **lateral strains**. Lateral strains have a nature/sense opposite to that of the linear or primary strain. Lateral strains are also called **transverse strains** or **secondary strains**.

**Elongation:** Increase in the dimension, length, breath or depth (height) of the structural member on account of externally applied load is known as elongation.

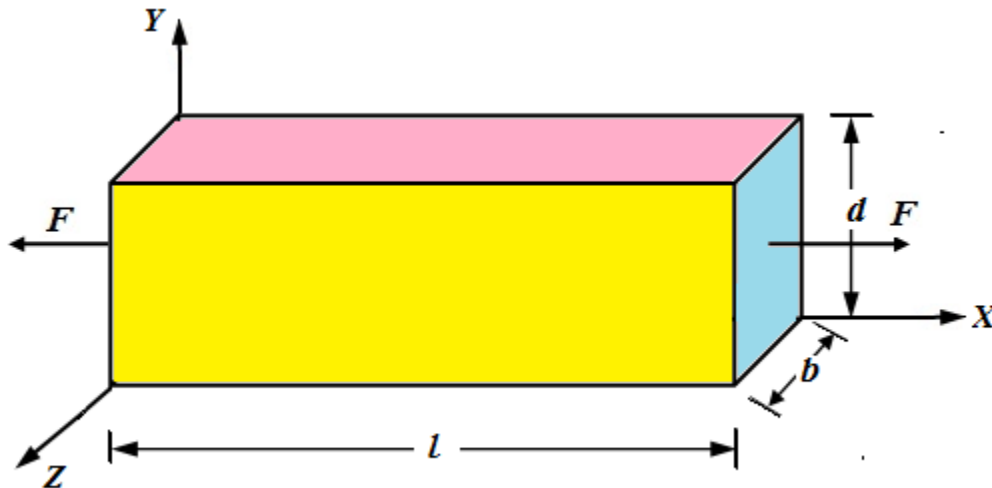
**Contraction:** Decrease in the dimension of structural member on account of externally applied load is known as contraction.

## Simple stresses and strains

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**Poisson's ratio:** The ratio of lateral strain to the linear strain is known as *Poisson's ratio*. It is denoted by 'ν' (Greek small letter Nu). Within elastic limit, in most material, this ratio remains fairly constant.

$$\text{Poisson's ratio, } \nu = \frac{\text{Lateral strain}}{\text{Linear strain}} = \frac{\epsilon_d}{\epsilon_l}$$



**Fig. Axial load along X-axis**

The value of  $\nu$  for most materials varies from 0.2 to 0.33. Poisson's ratio for steel is 0.3 and that for concrete is 0.15. Poisson's ratio cannot be 0.5 for any material.

Let us consider a bar of length  $l$ , breadth  $b$  and depth  $d$  subjected to axial tension of  $F$  along X-direction as shown in the figure.

Let strains along X, Y and Z directions be  $\epsilon_x$ ,  $\epsilon_y$  and  $\epsilon_z$  respectively.

Mathematically,

$$\epsilon_y = -\nu\epsilon_x$$

$$\epsilon_z = -\nu\epsilon_x$$

Negative sign indicates that the strain in the Y and Z directions are compressive, i.e., opposite to the strain in X direction.

## Simple stresses and strains

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### Volumetric strain:

It is the ratio of change in volume to original volume of the body (structural member) subjected to loading.

$$\epsilon_v = \frac{\Delta V}{V}$$

$\epsilon_v$  = volumetric strain

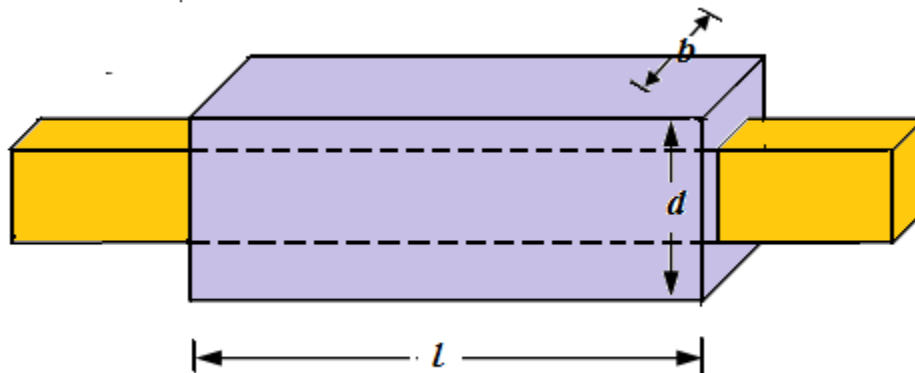
Where,

$V$  = original volume of the body

$\Delta V$  = Change in volume (increase or decrease)

Volumetric strain is the sum of strains in three mutually perpendicular directions.

### Volumetric strain of a rectangular bar (cuboid):



**Fig.**

Let us consider a rectangular bar of length  $l$ , breadth  $b$  and depth  $d$  is subjected to axial tension  $P_1$ ,  $P_2$  and  $P_3$  along X, Y and Z directions respectively as shown in the figure.

Let,  $\delta l$  = change in length

$\delta b$  = change in breadth

$\delta d$  = change in depth

Original volume,  $V = lbd$

Final volume =  $(l + \delta l)(b + \delta b)(l + \delta d)$

$$= lbd + lb \delta d + ld \delta b + bd \delta l + l \delta b \delta d + b \delta l \delta d + d \delta l \delta b + \delta l \delta b \delta d$$

$$= lbd + lb \delta d + ld \delta b + bd \delta l \quad (\text{neglecting product of small quantities})$$



## Simple stresses and strains

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Change in volume,  $\delta V = \text{Final volume} - \text{Initial volume}$

$$= (lbd + lb \delta d + ld \delta b + bd \delta l) - lbd$$

$$= lb \delta d + ld \delta b + bd \delta l$$

$$\text{Volumetric strain} = \frac{\text{Change in volume}}{\text{Original volume}} = \frac{\delta V}{V} = \frac{lb \delta d + ld \delta b + bd \delta l}{lbd}$$

$$= \frac{\delta l}{l} + \frac{\delta b}{b} + \frac{\delta d}{d}$$

$$= \epsilon_x + \epsilon_y + \epsilon_z$$

### Volumetric strain of a cylindrical rod:

Let us consider a cylindrical of length  $l$  and diameter  $d$  as shown in the figure subjected to axial pulls. Let the change in length and diameter is  $\delta l$  and  $\delta d$  respectively.

$$\text{Original volume} = \frac{\pi d^2 l}{4}$$

$$\text{Final volume} = \frac{\pi}{4} (d + \delta d)^2 (l + \delta l) = \frac{\pi}{4} (d^2 l + d \delta l + 2ld \delta d)$$

Ignoring products and higher powers of smaller quantities,

$$\text{Change in volume} = \delta V = \frac{\pi}{4} (d^2 \delta l + 2ld \delta d)$$

$$\text{Volumetric strain} = \epsilon_v = \frac{\text{Change in volume}}{\text{Original volume}} = \frac{\delta V}{V}$$

$$= \frac{d^2 \delta l + 2ld \delta d}{d^2 l} = \frac{\delta l}{l} + 2 \frac{\delta d}{d}$$

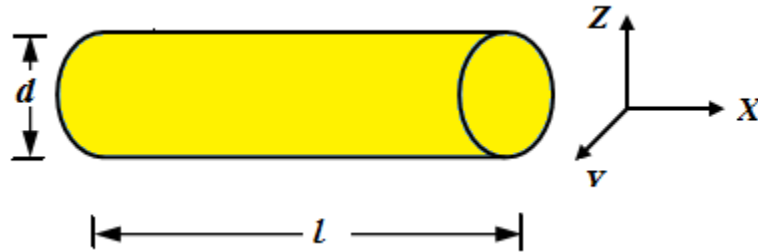
$$\epsilon_v = \epsilon_x + \epsilon_y + \epsilon_z$$

$$\text{Since,} \quad \epsilon_y = \epsilon_z = \frac{\delta d}{d}$$

Volumetric strain = Strain of the length + Twice the strain of the diameter.

## Simple stresses and strains

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In general, for any shape volumetric strain may be taken as the sum of the strains in three mutually perpendicular directions.

### Volumetric strain of a sphere:

Let us consider a cylindrical of diameter  $d$  as shown in the figure. Let the change in length and diameter is  $\delta d$ .

$$\text{Original volume, } V = \frac{\pi d^3}{6}$$

$$\begin{aligned} \text{Final volume} &= \frac{\pi}{6} (d + \delta d)^3 \\ &= \frac{\pi}{6} (d^3 + 3d^2 \delta d) \quad \text{ignoring the higher powers of } \delta d \end{aligned}$$

$$\text{Change in volume} = \delta V = \frac{\pi}{6} (3d^2 \delta d)$$

$$\text{Volumetric strain} = \epsilon_v = \frac{\text{Change in volume}}{\text{Original volume}} = \frac{\delta V}{V}$$

$$= \frac{3d^2 \delta d}{d^3} = \frac{3\delta d}{d} = 3\epsilon_d$$

Volumetric strain = Three times the strain in the diameter.

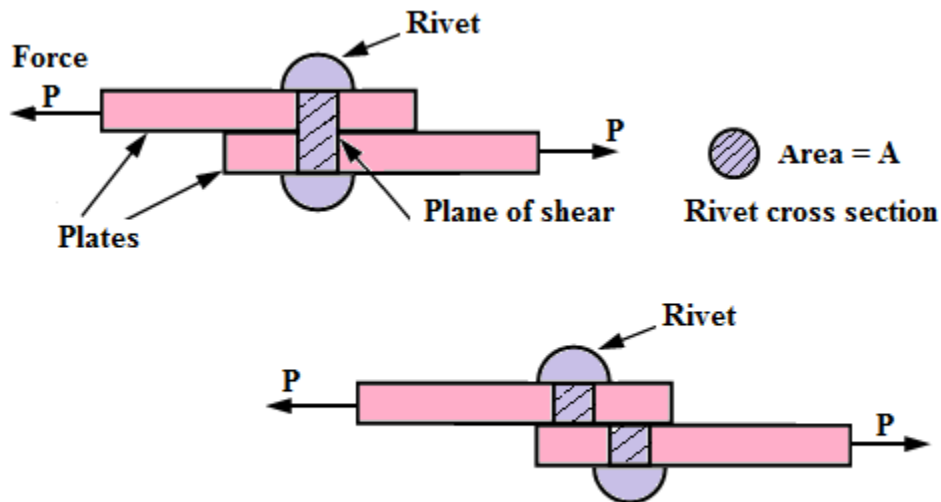
**Shear Stress:** Stress induced/developed on a plane subjected to tangential force (force parallel to the plane) is called *shear stress*. A material is said to be in a state of *simple shear* if it is subjected to only shearing stress.

## Simple stresses and strains

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If two equal and opposite parallel force act on a body, then there is a tendency of one part of the body to slide over the other. The stress induced in the body is known as shear stress. Shear stress is tangential to the area over which it acts.

A shearing stress alters only the shape of the body, leaving the volume unchanged.



**Fig. Shear stress**

Let us consider a rectangular block of length  $l$ , width  $b$  and depth  $d$  is fixed to the surface at its bottom face as shown in the figure.

$F$  = Force applied tangentially along surface AB and is called shear force

For equilibrium of the block the surface at CD will offer equal and opposite tangential reaction  $R = F$ .

Let the block be consists of two parts 1 and 2 separated by a section XX.

Consider the equilibrium of part 1. In order that the part 1 doesn't move from left to right, the part 2 will offer a resistance along the section XX such that  $R = F$ . Similarly, for equilibrium of part 2, part 1 will offer a resistance  $R$  along the section XX such that  $R = F$ .

The resistance  $R$  along the section XX is called shear resistance.

## Simple stresses and strains

$$\text{Shear stress, } \tau = \frac{\text{Shear resistance}}{\text{Shear area}} = \frac{R}{A} = \frac{F}{A}$$

$$\tau = \frac{F}{l \times b}$$

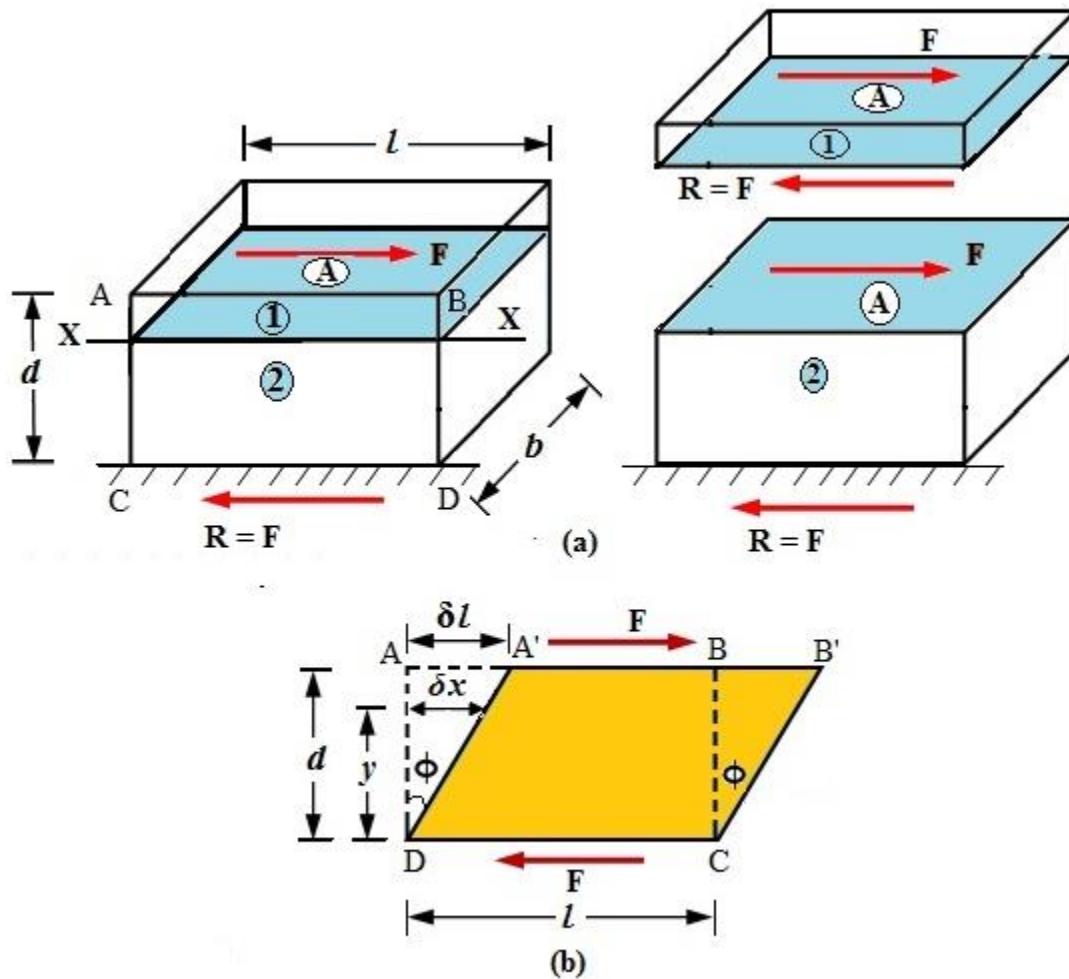


Fig. Shear (a) stress and (b) strain

### Shear deformation:

When the block doesn't fail in shear, it undergoes shear deformation as shown in figure (b). Since the bottom of the block is fixed, the block takes the shape of A'B'CD. The block undergoes distortion of  $\angle ADA' = \phi$ .

Let us think the block comprises of a number of thin horizontal layers. Each layer undergoes horizontal displacement in proportion to its distance from the lower face of the block. The ratio

## Simple stresses and strains

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of the horizontal displacement of the layer to distance of the layer from the bottom face is known as shear strain.

$$\text{Shear strain} = \frac{\delta l}{d}$$

$$\tan \phi = \frac{\delta l}{d}$$

$$\text{When } \phi \text{ is very small, } \phi \cong \tan \phi = \frac{\delta l}{d}$$

**Elastic Limit:** The limit of stress within which a structural member of any material under stress regains its original shape and size after removal of loading is called *elastic limit* of the material.

*The maximum stress which the material can withstand without causing permanent deformation after the removal of load is called elastic limit.*

If the stress exceeds the elastic limit, the member will not regain its original configuration. A residual strain called *permanent set* remains in the member.

**Limit of proportionality:** The limiting value of stress up to which stress is proportional to the strain is known as *limit of proportionality*.

In case of steel, a well pronounced proportionality limit can be found from tensile test. Some materials have a very small value of proportionality limit, others may not show such limit.

**Hooke's Law:** In 1678, English Mathematician Robert Hooke observed that there is a definite relationship between the elastic deformation (measured as *strain*) and the stress intensity causing it. He offered his observation in the form of a law called Hooke's law.

*Hooke's law states that when a material is loaded up to certain limit of stress intensity within elastic limit, the stress is directly proportional the strain in the material.*

$$\sigma \propto \varepsilon$$

Mathematically,

$$\Rightarrow \frac{\sigma}{\varepsilon} = E \text{ (Constant of proportionality)}$$

Where,  $\sigma$  is stress and  $\varepsilon$  is strain in the material of the member. The constant of proportionality,  $E$  is known as *modulus of elasticity or elastic modulus*. The elastic modulus for normal stress and

## Simple stresses and strains

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strain is also called Young's modulus. The modulus of elasticity is a measure of stiffness of the material. It has the same unit as stress. The slope of the stress-strain diagram in the linearly elastic region gives the value of **modulus of elasticity** ( $E$ ). Modulus of elasticity of steel is equal to 210 GPa, aluminum is 73 GPa and plastic is from 1 to 1.4 GPa. (1 GPa = 1000 N/mm<sup>2</sup>)

If  $P$  is the axial force acting in a prismatic member of sectional area  $A$  and length  $l$ ,

$$\begin{aligned} \text{Strain, } \quad \frac{\delta l}{l} &= \epsilon \Rightarrow \delta l = \epsilon l \\ \Rightarrow \delta l &= \frac{\sigma l}{E} = \frac{Pl}{AE} \end{aligned}$$

The product  $AE$  is called the **axial rigidity** or **axial stiffness** of the member.

For axial member, force necessary to produce one unit deformation (deflection) is known as **axial stiffness** and is denoted by  $K$ . Hence,  $\frac{AE}{L}$  is the axial stiffness for axially loaded member of length  $L$ . Reciprocal of stiffness,  $\frac{1}{K}$  is known as flexibility. More is the value of  $K$ , more is the stiffness and less is the flexibility and vice versa.

**Modulus of rigidity:** The ratio of shearing stress to corresponding shear strain within elastic limit is a constant known as modulus of rigidity or shear modulus. It is denoted by letter  $G$ . It has same unit as shear stress.

$$\text{Modulus of rigidity, } G = \frac{\tau}{\phi}$$

Where,

$$\begin{aligned} \tau &= \text{Shear stress} \\ \text{and } \phi &= \text{Shear strain} \end{aligned}$$

**Bulk modulus:** When a body is subjected to identical,  $\sigma$  (equal and like) stresses in three mutually perpendicular directions, it undergoes uniform changes in three directions without undergoing distortion of shape. In such case, the ratio of normal stress to volumetric strain is called **Bulk modulus**. It is denoted by letter  $K$ .

## Simple stresses and strains

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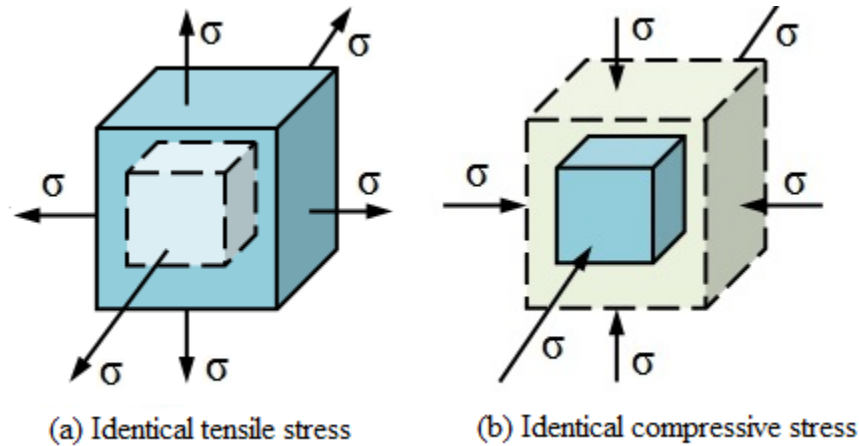


Fig. Stresses in three mutually perpendicular directions

**Bulk modulus** may be defined as the ratio of identical pressure acting in three mutually perpendicular directions to corresponding volumetric strain.

The element in figure (a) is subjected to identical tensile stress in three mutually perpendicular directions and the element in figure (b) is subjected to identical compressive stresses. Hydrostatic pressure acting on a submerged body is an ideal example of identical stresses in three mutually perpendicular directions.

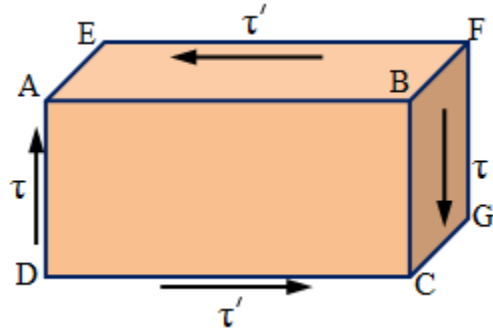
Mathematically, 
$$K = \frac{\sigma}{\epsilon_v}$$

### Complimentary shear stress:

A set of shear stresses acting across a plane is always accompanied by a balancing set of transverse shear stresses of same intensity across a plane normal to the previous plane. The balancing stress is called **complimentary shear stress**.

Let us consider a rectangular block ABCD of unit thickness perpendicular to the plane of the paper subjected to a shearing stress,  $\tau$  alongside AB and CD as displayed in the figure. The forces acting on these two faces are each equal to  $\tau \cdot (AB \cdot 1) = \tau \cdot (CD \cdot 1)$ . These two equal, opposite and parallel forces will form an anti-clockwise couple of magnitude =  $\tau \times AB \times AD$ .

## Simple stresses and strains



If the block is in equilibrium, a restoring (clockwise) couple of equal magnitude has to be developed by virtue of material resistance of the block. For this to happen, shear stress of intensity  $\tau'$  must be set upon the faces AD and BC.

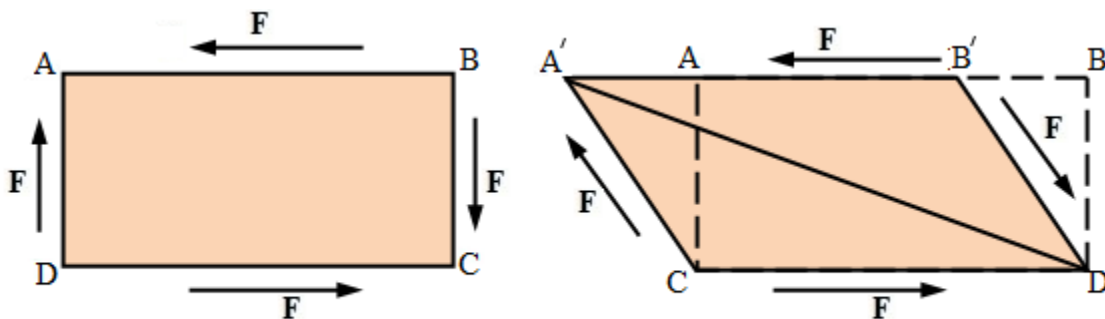
The forces acting on these faces are each equal to  $\tau'.AD$ . These two equal, opposite and parallel forces will form a restoring couple (clockwise) of magnitude =  $\tau' \times AD \times AB$ .

For, equilibrium equating the moment of two couples (acting and restoring), we have

$$\begin{aligned}\tau'.AD.AB &= \tau.AB.AD \\ \Rightarrow \tau' &= \tau\end{aligned}$$

$\tau'$  is called the complimentary shear stress.

Hence, a set of shear stresses is always accompanied by a complimentary set of shear stresses of equal intensity.



**Fig. Complimentary shear stress**



## Simple stresses and strains

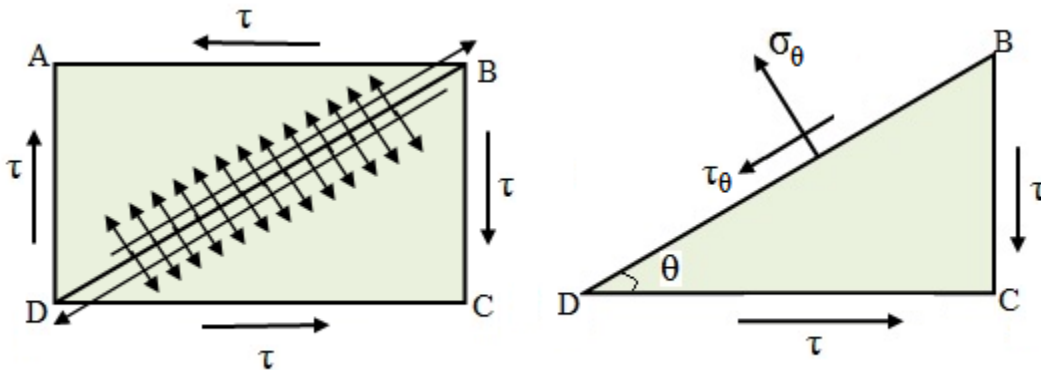
As a result of the two couples formed due to applied shear forces and the shear forces arising due to complimentary shear stresses, the diagonal AC of the block will be subjected to tension while the diagonal BD will be subjected to compression.

### Diagonal tension and compression:

Consider an elemental rectangular block ABCD with unit thickness perpendicular to the plane of the figure. Let the element be in the state of simple shear with shear stress intensity of  $\tau$  across the surfaces as shown in the figure.

Consider the plane through the diagonal BC which makes an angle of  $\theta$  with face CD. Consider the equilibrium of triangular wedge BCD. The wedge is subjected to the following forces.

- Force along BC =  $\tau \cdot BC$  ( $\downarrow$ )
- Force along DC =  $\tau \cdot DC$  ( $\rightarrow$ )
- Force normal to the plane DB =  $\sigma_\theta \cdot BD$
- Force tangential to the plane DB =  $\tau_\theta \cdot BD$



**Fig. Diagonal**

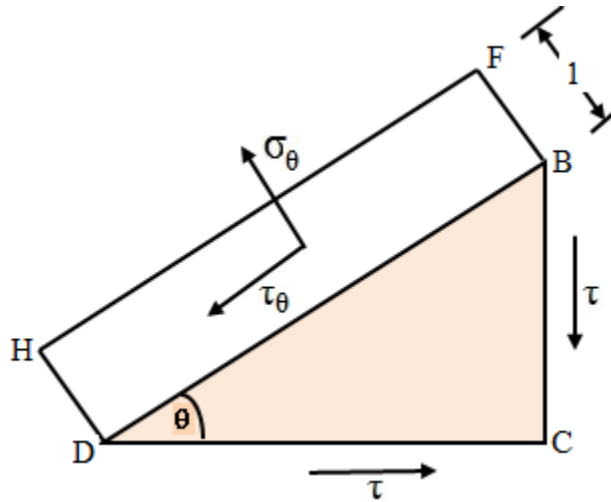
Resolving the forces normal to the plane BD and along the plane BD, we have

$$\sigma_\theta (BD \cdot 1) = \tau (BC \cdot 1) \cos \theta + \tau (DC \cdot 1) \sin \theta$$

$$\Rightarrow \sigma_\theta \cdot BD = \tau \cdot BD \sin \theta \cos \theta + \tau \cdot BD \cos \theta \sin \theta .$$

$$\Rightarrow \sigma_\theta = \tau \cdot 2 \sin \theta \cos \theta = \tau \sin 2\theta$$

## Simple stresses and strains



$$\tau_\theta(BD.1) = -\tau(BC.1)\sin\theta + \tau(DC.1)\cos\theta$$

Again, 
$$\Rightarrow \tau_\theta \cdot BD = -\tau \cdot BD \sin\theta \sin\theta + \tau \cdot B \cos\theta \cos\theta$$

$$\Rightarrow \tau_\theta = \tau(\cos^2\theta - \sin^2\theta) = \tau \cos 2\theta$$

Hence, normal and tangential stresses on plane BD are

$$\sigma_\theta = \tau \sin 2\theta$$

and 
$$\tau_\theta = \tau \cos 2\theta$$

For plane of maximum normal stress,

$$\sin 2\theta = \pm 1$$

$$\text{i.e., } 2\theta = \pm 90^\circ \text{ i.e., } \theta = \pm 45^\circ$$

$$\text{when } \theta = 45^\circ, \sigma_\theta = +\tau \text{ (+ve sign indicates that normal stress is tensile)}$$

$$\text{when } \theta = -45^\circ, \sigma_\theta = -\tau \text{ (-ve sign indicates normal stress compressive)}$$

Corresponding to  $\theta = \pm 45^\circ$ , tangential stress  $\tau_\theta = 0$

Hence, the planes carrying maximum normal stress don't carry any shear stress.

For plane of maximum shear stress,

$$\cos 2\theta = \pm 1$$

$$\text{i.e., } 2\theta = 0 \text{ or } 180^\circ \text{ i.e., } \theta = 0 \text{ or } 90^\circ$$

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Planes corresponding to maximum shear stress, the normal stresses are zero.

When an element is in the state of simple shear, the maximum direct stresses are induced on mutually perpendicular planes which are at  $45^\circ$  to the planes of pure shear. One of the maximum direct stresses is tensile while the other is compressive. The direct maximum tensile and maximum compressive stresses are of the same magnitude as the shear stress on the planes of pure shear.

### Relationship between modulus of elasticity and modulus of rigidity:

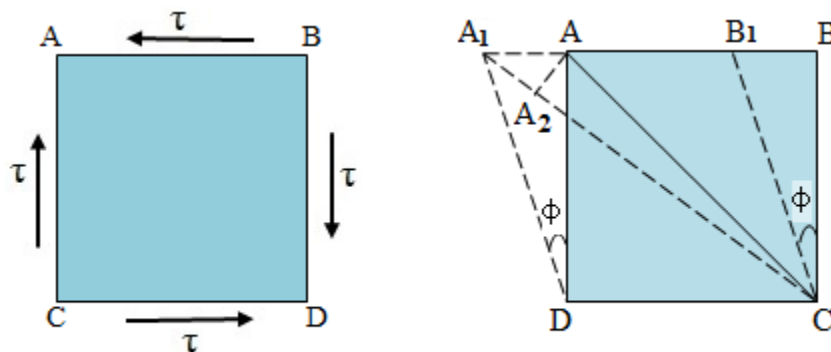
Consider a square block ABCD of side  $a$  and thickness unity perpendicular to the plane of the paper. Let the block be in the state of simple shear (pure shear) with shear stress intensity of  $\tau$  across the surfaces as shown in the figure. The block will undergo distortion of shape due to the system of stresses.

It is evident from the stress components that the diagonal AC will be elongated and the diagonal BD will be shortened.

The increase in length of the diagonal AC can be computed by considering the diagonal tensile and compressive stresses on AC and BD respectively. It is known that the diagonal tensile and compressive stresses in a square block are each equal to  $\tau$ .

Strain in the diagonal AC = Strain in AC due to diagonal tensile stress on the plane BD + Strain in length AC due to diagonal compressive stress on plane AC.

$$\text{Strain in AC} = \frac{\tau}{E} + \frac{\nu \cdot \tau}{E} = \frac{\tau}{E} (1 + \nu) \quad (1)$$



**Fig. Square block with simple shear**

## Simple stresses and strains

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Strain in the diagonal AC can also be computed from the distorted geometry of the block.

Let the geometry ABCD deform to the position  $A_1B_1CD$  through the angle  $\phi$ .

Increase in length of the diagonal  $AC = A_1C - AC$ .

Let  $AA_2$  be perpendicular to  $A_1C$ . Since the angle  $ACA_2$  is very small,  $AC = A_2C$

Hence, increase in length of the diagonal  $AC = A_1C - A_2C = A_1A_2 = AA_1 \cos AA_1A_2$ . But the angle  $AA_1A_2$  is equal to  $BAC = 45^\circ$ .

Hence, increase in length of the diagonal AC

$$= AA_1 \cos 45^\circ = \frac{AA_1}{\sqrt{2}}$$

$$\begin{aligned} \text{Shear strain, } \phi &= \frac{AA_1}{AD} = \frac{AA_1}{a} \\ \Rightarrow AA_1 &= a\phi \end{aligned}$$

Increase in length of the diagonal  $AC = a\sqrt{2}$ .

Length of the diagonal  $AC = a\sqrt{2}$

$$\text{Strain of the diagonal } AC = \frac{\text{Increase in length}}{\text{Original length}} = \frac{a\phi}{\sqrt{2}} \cdot \frac{1}{a\sqrt{2}} = \frac{\phi}{2} \quad (2)$$

From Equation (1) and (2), we have

$$\frac{\phi}{2} = \frac{\tau}{E}(1+\nu) \Rightarrow \frac{\tau}{2G} = \frac{\tau}{E}(1+\nu)$$

$$\therefore E = 2G(1+\nu) \quad (3)$$

### Relationship between Young's modulus and bulk modulus:

Consider a cube of side 'a' subjected to direct tensile stress of intensity  $\sigma$  in three mutually perpendicular directions. Let E be the Young's modulus and  $\nu$ , the Poisson's ratio.

Let us consider the strain in one edge AB of the cube.

## Simple stresses and strains

Strain in AB due to tensile stress in the X-direction

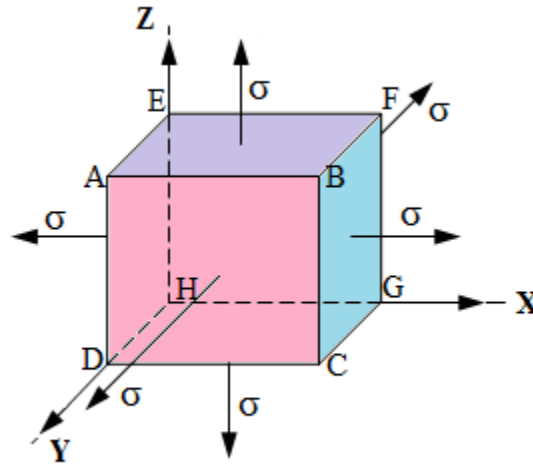


Fig.

$$\text{Strain in AB due to tensile stress in X - direction} = \frac{\sigma}{E}$$

$$\text{Strain in AB due to tensile stress in Y - direction} = -\frac{\nu\sigma}{E}$$

$$\text{Strain in AB due to tensile stress in Z - direction} = -\frac{\nu\sigma}{E}$$

$$\text{Total Strain in AB, } \varepsilon_x = \frac{\sigma}{E} - \frac{\nu\sigma}{E} - \frac{\nu\sigma}{E} = \frac{\sigma}{E}(1 - 2\nu)$$

Similarly strains in the other two edges are each equal to  $\varepsilon_y = \varepsilon_z = \frac{\sigma}{E}(1 - 2\nu)$

$$\text{Volumetric strain, } \varepsilon_v = \varepsilon_x + \varepsilon_y + \varepsilon_z = \frac{3\sigma}{E}(1 - 2\nu)$$

$$\text{Bulk modulus, } K = \frac{\text{Normal stress}}{\text{Volumetric strain}} = \frac{\sigma}{\frac{3\sigma}{E}(1 - 2\nu)}$$

$$\Rightarrow E = 3K(1 - 2\nu) \quad (4)$$

The relationship between Young's modulus and Bulk modulus is given by the above equation.

Further from equation (3), we have

## Simple stresses and strains

---

$$E = 2G(1 + \nu)$$
$$\Rightarrow 2\nu = \frac{E}{G} - 2$$

From equation (4), we have

$$E = 3K(1 - 2\nu)$$
$$\Rightarrow E = 3K \left\{ 1 - \left( \frac{E}{G} - 2 \right) \right\}$$
$$\Rightarrow E = 3K \left( 3 - \frac{E}{G} \right) = \frac{9KG - 3EK}{G}$$
$$\Rightarrow EG = 9KG - 3EK$$
$$\Rightarrow E(3K + G) = 9KG$$
$$\Rightarrow E = \frac{9KG}{3K + G}$$

The above equation is the relationship between three elastic moduli.

For isotropic and homogeneous material, there are four elastic constants, namely  $E$ ,  $G$ ,  $K$  and  $\nu$ . Out of the four constants two are independent constants and other two can be obtained by using the relationship between them.

### **Deformation of prismatic bar due to uniaxial loading:**

Consider a prismatic bar subjected to uniaxial tensile load as shown in the figure.

Let

$P$  = Load acting on the bar

$l$  = Original length of the bar

$A$  = Cross-sectional area of the bar

$\sigma$  = Stress induced in the bar

$E$  = Young's modulus of the material of the bar

$\varepsilon$  = Strain in the bar

$\delta l$  = Deformation or change in length of the bar

We know that,

## Simple stresses and strains

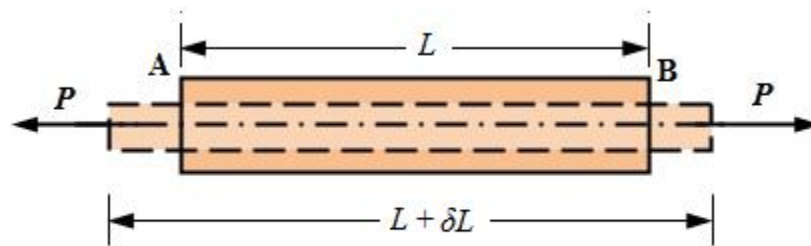


Fig. Uniaxial load (Tensile)

$$\text{Stress, } \sigma = \frac{P}{A}; \quad \text{Strain, } \epsilon = \frac{\sigma}{E} = \frac{P}{AE}$$

$$\text{and deformation, } \delta l = \epsilon \cdot E = \frac{\sigma \cdot l}{E} = \frac{Pl}{AE}$$

### Deformation of prismatic bar under self weight:

Consider a freely hanging prismatic bar  $AB$  of length  $L$ , cross sectional area  $A$  and weight  $W$  as displayed in the figure. Let  $\omega$  = Specific weight of the material of the bar.

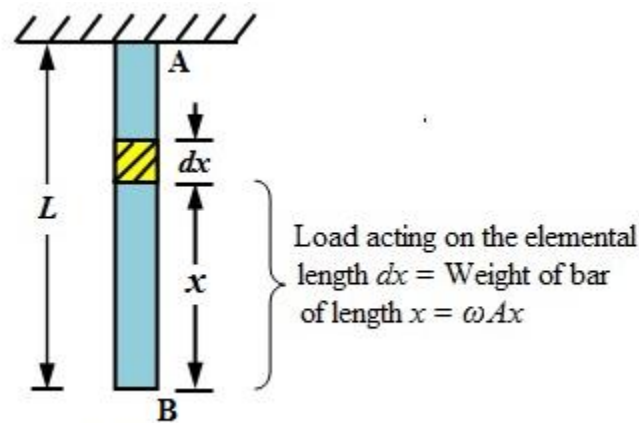


Fig.

Let us consider an elementary length,  $dx$  of the bar at a height of  $x$  from the bottom  $B$  of the bar. Weight of the portion of the bar below  $XX$  act as tensile force on the elementary length. Weight of the portion below  $XX$ ,  $P = \omega Ax$ .

Elongation of the elementary length  $dx$  due to the weight of the bar of length  $x$

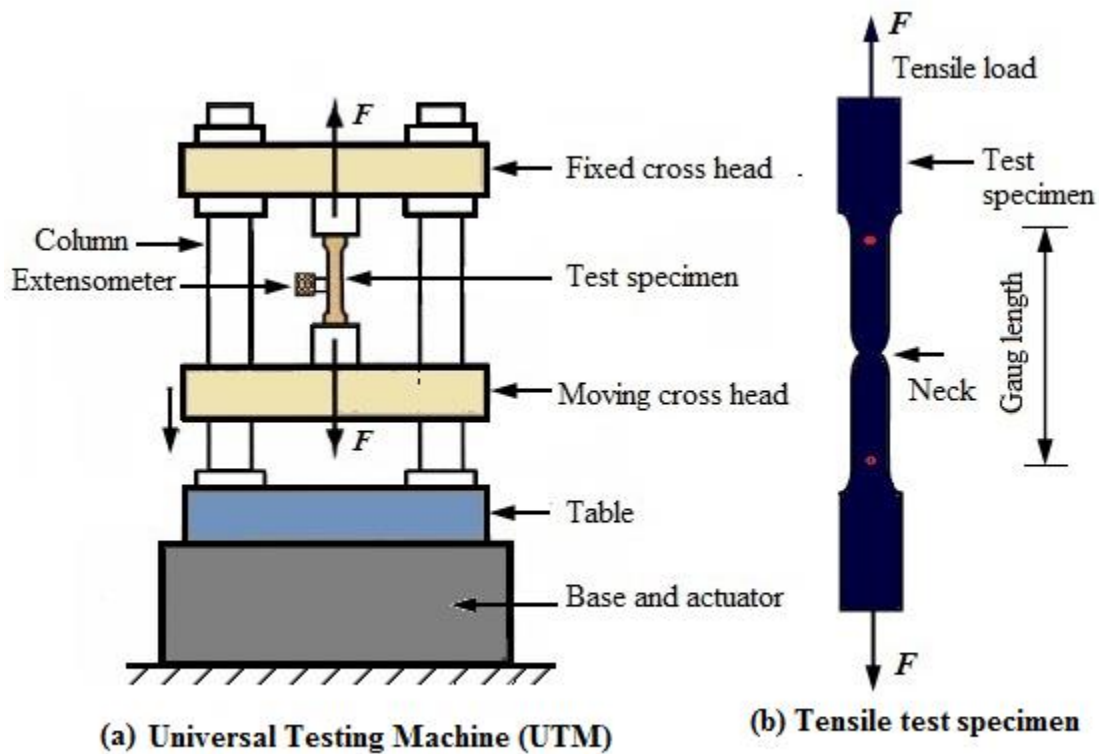
## Simple stresses and strains

$$\begin{aligned}\delta(dx) &= \frac{P \cdot dx}{AE} = \frac{(\omega Ax) dx}{AE} \\ &= \frac{\omega x dx}{E}\end{aligned}$$

Total elongation of the bar due to self weight,

$$\begin{aligned}\int_0^L \delta(dx) &= \int_0^L \frac{\omega x dx}{E} = \frac{\omega}{E} \int_0^L x dx \\ &= \frac{\omega}{E} \left[ \frac{x^2}{2} \right]_0^L = \frac{\omega L^2}{2E} \\ &= \frac{(\omega LA)L}{2AE} \\ \Rightarrow \delta L &= \frac{WL}{2AE}\end{aligned}$$

**Tensile Test:**



**Fig. Tensile Test Setup**



## Simple stresses and strains

The **tension test** is the most common method for determining the *mechanical properties* of materials, such as strength, ductility, toughness, elastic modulus, and strain hardening capability. In a tension test, a specimen of standard dimension is subjected to a continually increasing uniaxial tensile force while simultaneous observation of elongation is taken by means of extensometer fitted to the specimen. The results of the test are plotted as **stress-strain diagram**. A standard tensile test set up with the representative (a) UTM (Universal Testing Machine) and (b) tensile test specimen is shown in the figure.

A **stress-strain diagram** is a diagram in which corresponding values of stress and strain are plotted against each other. The values of the stress are plotted as ordinates in vertical (Y-axis) and values of strain as abscissas in horizontal axis (X-axis).

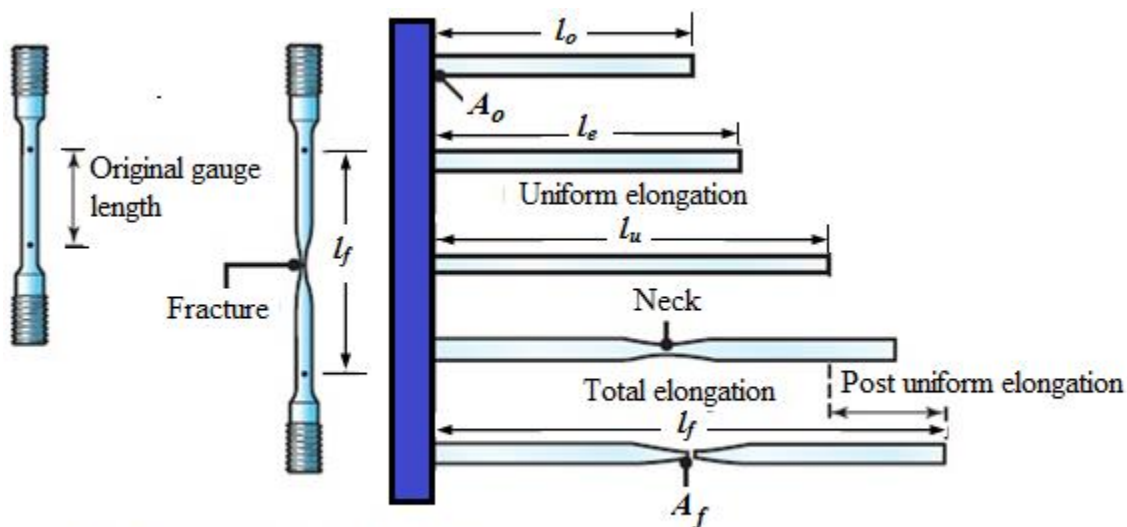


Fig. Typical diagram of tensile testing material specimen

### Stress-strain curve for ductile material:

A representative/typical stress-strain ( $\sigma$ - $\epsilon$ ) diagram for ductile material (mild steel) is displayed in the figure with salient points on the curve.

Stress is proportional to the strain up to the point A, i.e., the stress-strain variation is linear. This point represents **limit of proportionality**. Hooke's law holds good up to this point. Slope of stress-strain line from O to A gives the modulus of elasticity also known as Young's modulus.

## Simple stresses and strains

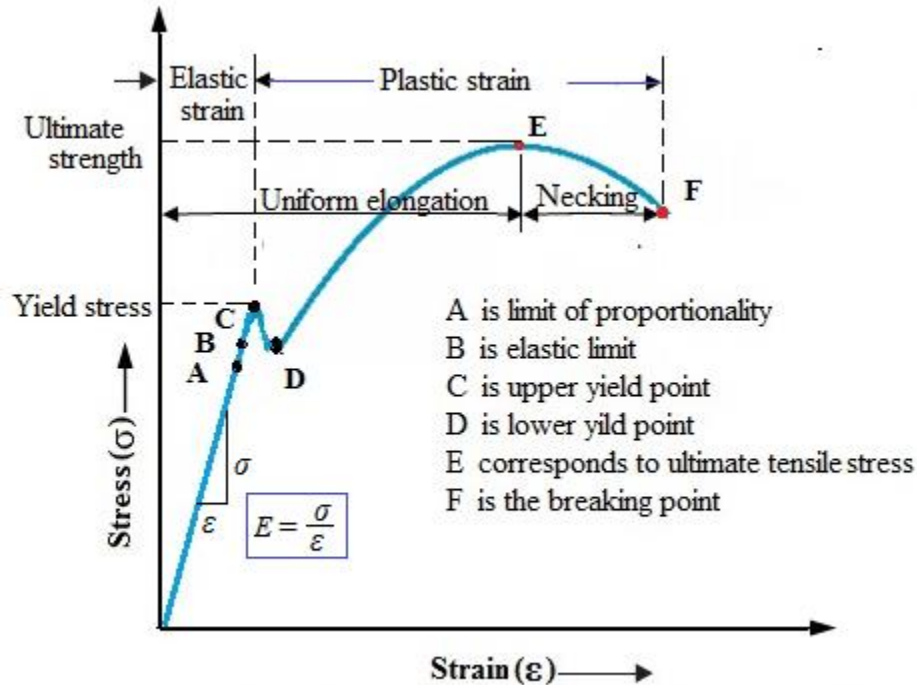


Fig. Stress-strain diagram for ductile material

Beyond point *A* and up to point *B*, material remains elastic *i.e.*, the material returns to its original condition of the forces acting on it is removed. The stress corresponding to represents the ***stress at elastic limit***. If the specimen is stressed beyond point *B*, permanent set takes place and we enter plastic deformation region. In the plastic deformation region, the strain does not get fully removed even with the removal of the force causing it.

If the force is increased further, point '*C*' is reached where the test specimen elongates even when the stress is not increased. This point is called yield point. In fact, there are two yield points *C* and *D* which are called upper and lower yield points respectively. The stress corresponding to the yield point is known as ***yield stress or yield strength***.

With further straining, the effect of a phenomenon called strain hardening or work hardening takes place.\* The material becomes stronger and harder and its load bearing capacity increases. The test specimen is therefore able to bear more stress. On progressively increasing the force acting on the specimen, point *E* is reached. This point is the highest point in the stress-strain curve and represents the point of maximum stress. It is, therefore, called ***ultimate tensile strength*** (UTS) of the material. It is equal to the maximum load applied divided by the original cross-sectional area ( $A_0$ ) of the test specimen.

## Simple stresses and strains

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After UTS point  $E$ , a sharp reduction in cross-sectional area of the test specimen takes place and a “**neck**” is formed in the centre of the specimen. Ultimately the test specimen breaks in two pieces as the neck becomes thinner and thinner. The point  $F$  represents the breaking point and the corresponding stress is known as **breaking stress or fractured stress**. The actual breaking stress is much higher than the UTS, if the reduced cross-sectional area of the test specimen is taken into account.

As plastic deformation increases, the cross-sectional area of the specimen decreases. However for calculation of the stress in the stress-strain graph, the original cross-sectional area is considered. It is for this reason that the breaking point,  $F$  seems to occur at a lower stress level than the UTS point  $E$ .

The measure of the strength of a material is the ultimate tensile strength ( $\sigma$  at point  $E$ ). However, from the design point of view, the yield point is more important as the designed structure should withstand forces without yielding. Usually yield stress ( $\sigma$  at point  $D$ ) is two-thirds of the UTS and this is referred to as **yield-strength** of the material.

### Stress-strain curve for brittle material:

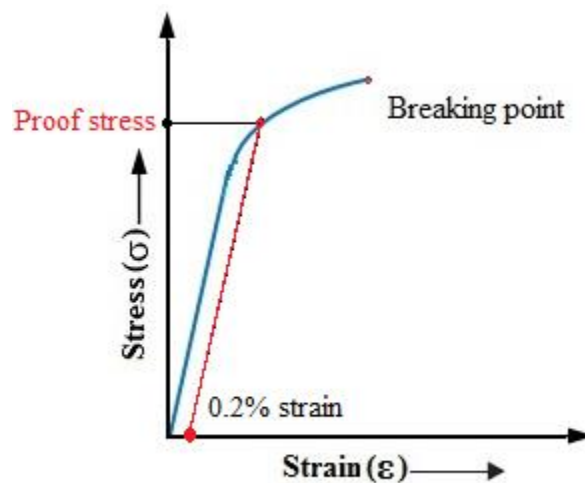


Fig. Stress-strain curve for brittle material

A stress-strain curve for brittle material (cast iron) is obtained by subjecting a test bar of such material in a tensile testing machine. The tensile load is gradually increased and the extension of

## Simple stresses and strains

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the test piece is recorded. The stress-strain curve for a brittle material is significantly different from that for a ductile material. A typical stress-strain curve for a brittle material is shown in Fig.

This curve displays no yield point, and the test specimen breaks suddenly without any appreciable necking or extension. In the absence of a yield point, concept of “*proof-stress*” has been evolved for measuring yield strength of a brittle material. For example, 0.2% proof-stress indicates the stress at which the test specimen ‘suffers’ a permanent elongation equal to 0.2% of initial gauge length.

**Percentage of elongation:** Increase in length of tensile test sample expressed in percentage of original length of the specimen is known as percentage of elongation or percentage of increase in length. It has considerable significance in engineering because it indicates the ductility of the material.

The ability of material to deform appreciably without rupture is known as ductility. It is a measure of the amount of plastic deformation that a material goes through before it fails. It refers to plastic deformation under tensile loads. Ductility enables the material to be drawn into wires. Highly ductile metals can exhibit significant strain before fracturing, whereas brittle materials frequently display very little strain. Ductility increases with temperature.

There are two common measures of ductility.

The first is the **total elongation** of the specimen, given by

$$\text{Percentage elongation (\%)} = \frac{(l_f - l_0)}{l_0} \times 100$$

where  $l_f$  and  $l_0$  are the original and final length of the test specimen.

The second measure of ductility is the reduction of area, given by

$$\text{Reduction of area (\%)} = \frac{(A_f - A_0)}{A_0} \times 100$$

where,  $A_f$  and  $A_0$  are, respectively, the original and final (fracture) cross-sectional area of the test specimen.

## Simple stresses and strains

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Reduction of area and elongation are generally interrelated. Thus, the ductility of a piece of chalk is zero, because it does not stretch at all or reduce in cross section; by contrast, a ductile specimen, such as putty or chewing gum, stretches and necks considerably before it fails.

Metals with more than 15% elongation at fracture are considered as **ductile**. Metals with 5 to 15% elongation are considered of **intermediate ductility**. However, the metals with less than 5% elongation, i.e., strain of 0.05 are considered as **brittle** ones. Brittle materials include ceramic, glass and some alloys. Cast iron is also classified as brittle.

**Working stress:** The maximum stress to which a structural member is ever allowed to be subjected to is called **working stress**. It should be below the elastic limit.

**Ultimate stress:** The maximum stress to which the material of the test piece is subjected to during the test is known as **ultimate stress** or **ultimate strength**. It is obtained by the maximum load to which the test piece is subjected to divided by the original cross-sectional area.

**Factor of safety:** the ratio of the ultimate stress to the working stress is called the **factor of safety**. The value of factor of safety in engineering design varies from 3 (for accurately known dead load) to 12 (for shock loads of indefinite magnitude).

$$\text{Factor of safety} = \frac{\text{Ultimate stress}}{\text{Working stress}}$$

In case of ductile material, since excessive deformation creates a problem in the performance of the structural member, working stress is taken as a factor of yield stress or that of proof stress (if yield stress doesn't exist) in place of ultimate stress.

Factor of safety for steel is 1.85, for concrete it is 3.

**True stress:** it is the ratio of the load to the actual cross-sectional area of the test piece.

**Engineering stress:** It is the ratio of the load to the original cross-sectional area of the test piece. It is also known as **nominal stress**.

# Simple stresses and strains

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## Numerical Examples

1. An elastic rod 25 mm diameter, 200 mm long extends by 0.25 mm under a tensile load of 40 kN. Find the intensity of stress, strain and elastic modulus for the material of the rod.

Solution. Given, Diameter of rod,  $d = 25 \text{ mm}$ , Length,  $l = 200 \text{ mm}$

Increase in length,  $\delta l = 0.25 \text{ mm}$ , Load,  $P = 40 \text{ kN} = 40,000 \text{ N}$

$$\text{Area, } A = \frac{\pi}{4} d^2 = \frac{\pi}{4} \times 25^2 = 490.87 \text{ mm}^2$$

$$\text{Intensity of stress, } \sigma = \frac{\text{Load}}{\text{Area}} = \frac{P}{A} = \frac{40000 \text{ N}}{490.87 \text{ mm}^2} = 81.49 \text{ N/mm}^2$$

$$\text{Strain, } \epsilon = \frac{\delta l}{l} = \frac{0.25}{200} = 0.00125$$

$$\text{Elastic modulus, } E = \frac{\sigma}{\epsilon} = \frac{81.49}{0.00125} = 65,192 \text{ N/mm}^2$$

2. A steel rod of 25 mm diameter and 2 m long is subjected to an axial pull of 45 kN. Find the (a) intensity of stress, (b) strain and (c) elongation. Take  $E = 2 \times 10^5 \text{ N/mm}^2$ .

Solution. Given, Diameter of rod,  $d = 25 \text{ mm}$ , Length,  $l = 2 \text{ m} = 2000 \text{ mm}$

Load,  $P = 45 \text{ kN} = 45,000 \text{ N}$

$E = 2 \times 10^5 \text{ N/mm}^2$

$$\text{Area, } A = \frac{\pi}{4} d^2 = \frac{\pi}{4} \times 25^2 = 490.87 \text{ mm}^2$$

$$\text{Intensity of stress, } \sigma = \frac{\text{Load}}{\text{Area}} = \frac{P}{A} = \frac{45,000 \text{ N}}{490.87 \text{ mm}^2} = 91.67 \text{ N/mm}^2$$

$$\text{Strain, } \epsilon = \frac{\sigma}{E} = \frac{91.67}{2 \times 10^5} = 4.583 \times 10^{-4} = 0.0004583$$

$$\begin{aligned} \text{Elongation} &= \text{Strain} \times \text{Original length} = \epsilon \times l = 4.583 \times 10^{-4} \times 2000 \\ &= 0.916 \text{ mm} \end{aligned}$$

3. A load of 4000 N has to be raised at the end of a steel wire. If the unit stress in the wire must not exceed  $80 \text{ N/mm}^2$ , what is the minimum diameter of the rod required? What will be the extension of 3.5 m length of the wire? Take  $E = 2 \times 10^5 \text{ N/mm}^2$ .

Solution. Given, Length,  $l = 3.5 \text{ m} = 3500 \text{ mm}$ , Permissible stress =  $80 \text{ N/mm}^2$

## Simple stresses and strains

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$$\text{Load to be raised, } P = 4000 \text{ N, } \quad E = 2 \times 10^5 \text{ N/mm}^2$$

Let the minimum required diameter of rod =  $d$

$$\text{Cross-sectional Area, } A = \frac{\pi}{4} d^2$$

$$\text{Stress in the wire, } \sigma = \frac{P}{A} = \frac{4,000}{\left(\frac{\pi}{4}\right) d^2}$$

For the wire to sustain the load,  $\sigma \leq$  Permissible stress

$$\Rightarrow \frac{4,000}{\left(\frac{\pi}{4}\right) d^2} \leq 80$$

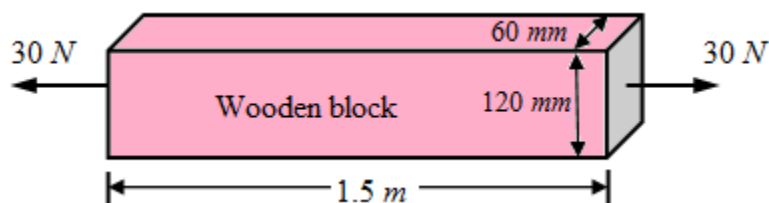
$$\Rightarrow \frac{\pi}{4} d^2 \geq \frac{4000}{80}$$

$$\Rightarrow d \geq \sqrt{\frac{4000}{\pi}}$$

$$\Rightarrow d \geq 7.97 \text{ mm}$$

$$\text{Elongation, } \delta l = \frac{\sigma \cdot l}{E} = \frac{80 \times 3500}{2 \times 10^5} = 1.44 \text{ mm}$$

4. A wooden tie in the figure is 60 mm wide, 120 mm deep and 1.5 m long. It is subjected to an axial pull of 30 N. The stretch of the member is found to be 0.625 mm. Find the Young's modulus of the material.



Solution. Given, Length,  $l = 1.5 \text{ m} = 1500 \text{ mm}$ , Width = 60 mm

Depth = 120 mm, Pull,  $P = 30 \text{ N}$

Increase in length,  $\delta l = 0.625 \text{ mm}$

## Simple stresses and strains

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$$\begin{aligned}\text{Area of cross-section} &= \text{Width} \times \text{Depth} = 60 \times 120 \\ &= 7200 \text{ mm}^2\end{aligned}$$

$$\text{Young's modulus, } E = \frac{PL}{A \cdot \delta l} = \frac{30 \times 1500}{7200 \times 0.625} = 10 \text{ N/mm}^2$$

5. A hollow steel column of external diameter 250 mm has to support an axial load of 2000 kN. If the ultimate stress for the steel column is 480 N/mm<sup>2</sup>, find the internal diameter of the column allowing a load factor of 4.

Solution. Given, External diameter,  $D = 250 \text{ mm}$ , Axial load,  $P = 2000 \text{ kN}$

Ultimate stress,  $\sigma_u = 480 \text{ N/mm}^2$ , Load factor = 4

Let the internal diameter of rod =  $d$

$$\text{Load factor} = \frac{\text{Ultimate stress}}{\text{Safe stress}}$$

$$\Rightarrow \text{Safe stress} = \frac{480}{4} = 120 \text{ N/mm}^2$$

$$\text{Safe stress} = \frac{P}{A} = \frac{2000 \times 1000}{\frac{\pi}{4} \times (250^2 - d^2)}$$

$$\Rightarrow 120 = \frac{8 \times 10^6}{\pi(250^2 - d^2)}$$

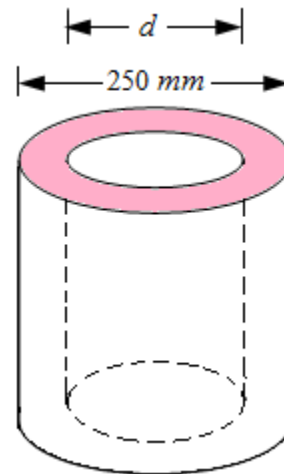
$$\Rightarrow (250^2 - d^2) = \frac{8 \times 10^6}{120\pi}$$

$$\Rightarrow 250^2 - d^2 = 21220.659$$

$$\Rightarrow d^2 = 250^2 - 21220.659$$

$$\Rightarrow d^2 = 41279.341$$

$$\Rightarrow d = 203.173 \text{ mm}$$



6. The following data refers to the tensile test conducted on a mild steel bar.

- i. Diameter of steel bar = 3 mm
- ii. Gauge length = 200 mm
- iii. Extension at a load of 100 kN = 0.139 mm
- iv. Load at elastic limit = 230 kN



## Simple stresses and strains

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- v. Maximum load = 360 kN
- vi. Total extension = 56 mm
- vii. Diameter of rod at failure = 22.25 mm

Calculate (a) Young's modulus, (b) the stress at elastic limit, (c) the percentage of elongation and (d) the percentage decrease in area

Solution. Gauge length,  $l_o = 200 \text{ mm}$ , Diameter of the bar,  $d_o = 30 \text{ mm}$

a) Young's modulus.

$$\text{Cross-sectional area, } A = \frac{\pi d_o^2}{4} = \frac{\pi \times 30^2}{4} = 706.86 \text{ mm}^2$$

$$\text{Stress at load } (P = 100 \text{ kN}), \sigma = \frac{P}{A} = \frac{100 \times 1000}{706.86} = 141.47 \text{ N/mm}^2$$

$$\text{Strain at load } (P = 100 \text{ kN}), \epsilon = \frac{\text{Extension}}{\text{Original length}} = \frac{0.139}{200} = 0.000695$$

$$\text{Young's modulus, } E = \frac{\sigma}{\epsilon} = \frac{141.47}{0.000695} = 2.035 \times 10^5 \text{ N/mm}^2$$

b) Stress at elastic limit.

Load at elastic limit,  $P_e = 230 \text{ kN}$

$$\text{Stress at elastic limit, } \sigma_e = \frac{P_e}{A} = \frac{230 \times 1000}{706.86} = 325.383 \text{ N/mm}^2$$

c) Percentage of elongation.

$$\text{Percentage of elongation} = \frac{\text{Total increase in length}}{\text{Gauge length}} \times 100 = \frac{56}{200} \times 100 = 28\%$$

d) Percentage decrease in area.

Diameter of rod at failure,  $d_f = 22.5 \text{ mm}$

$$\begin{aligned} \text{Decrease in area} &= \frac{\pi}{4} (d_o^2 - d_f^2) = \frac{\pi}{4} (30^2 - 22.5^2) \\ &= 309.250 \text{ mm}^2 \end{aligned}$$

$$\begin{aligned} \text{Percentage decrease in area} &= \frac{\text{Decrease in area}}{\text{Gauge area}} \times 100 = \frac{309.25}{706.86} \times 100 \\ &= 43.74\% \end{aligned}$$

## Simple stresses and strains

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7. A steel rod of 28 mm diameter and 300 mm long is subjected to axial forces alternating between a maximum compression of 16 kN and a maximum tension of 7 kN. Find the difference between the greatest and least lengths of the rod. Take  $E = 210 \text{ GPa}$ .

**Solution.** Given: Diameter of rod,  $d = 28 \text{ mm}$ ; Length of the rod,  $l = 300 \text{ mm}$

Axial compression,  $P_c = 16 \text{ kN} = 16000 \text{ N}$

Axial tension,  $P_t = 7 \text{ kN} = 7000 \text{ N}$

$E = 210 \text{ GPa} = 210 \times 10^3 \text{ N/mm}^2$

Area of cross – section,  $A = \frac{\pi}{4} d^2 = \frac{\pi}{4} \times 28^2 = 615.75 \text{ mm}^2$

- a) When the rod is subjected to axial pull (tension)

Increase in length,  $\delta l_1 = \frac{P_t l}{AE} = \frac{7000 \times 300}{615.75 \times 210 \times 10^3} = 0.016(+)$

Greatest length after elongation =  $300 + 0.016 = 300.016 \text{ mm}$

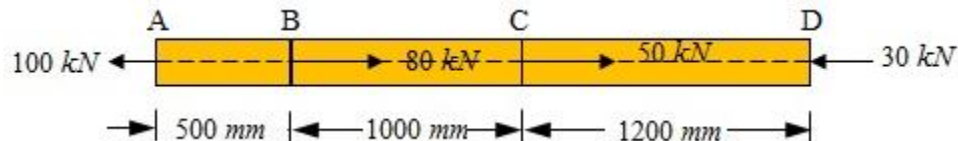
- b) When the rod is subjected to axial push (compression)

Increase in length,  $\delta l_2 = \frac{P_c l}{AE} = \frac{16000 \times 300}{615.75 \times 210 \times 10^3} = 0.037(-)$

Least length after compression =  $300 - 0.037 = 299.963 \text{ mm}$

Difference between the greatest and least length =  $300.016 - 299.963$   
 $= 0.053 \text{ mm}$

8. A straight bar of brass having cross-sectional area of  $500 \text{ mm}^2$  is subjected to axial forces as shown in the figure.



Find the total elongation of the bar. Take  $E = 80 \text{ GPa}$ .

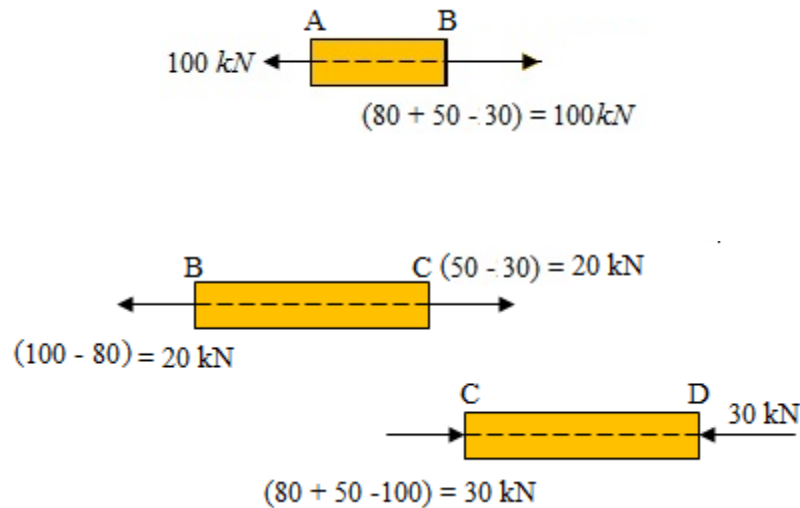
**Solution.** Given: Cross-sectional area,  $A = 500 \text{ mm}^2$ ,  $E = 80 \text{ GPa} = 80 \times 10^3 \text{ N/mm}^2$

For the sake of simplicity, the bar may be considered to be comprised of three portions, AB, BC and CD. The elongation or contractions of each of the three portions are computed separately.

## Simple stresses and strains

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Equilibrium of each portion of the bar is considered for computation of the elongation or contraction.



Portion AB: AB is subjected to a tensile force of 100 kN

$$\text{Elongation of } AB, \delta_{AB} = \frac{P_1 l_1}{AE} = \frac{100 \times 1000 \times 500}{500 \times 80 \times 1000} = 1.25 \text{ mm}(+)$$

Portion BC: BC is subjected to a tensile force of 20 kN

$$\text{Elongation of } BC, \delta_{BC} = \frac{P_2 l_2}{AE} = \frac{20 \times 1000 \times 1000}{500 \times 80 \times 1000} = 0.5 \text{ mm}(+)$$

Portion CD: CD is subjected to a tensile force of 20 kN

$$\text{Contraction of } CD, \delta_{CD} = \frac{P_3 l_3}{AE} = \frac{30 \times 1000 \times 1200}{500 \times 80 \times 1000} = 0.9 \text{ mm}(-)$$

$$\text{Total elongation } , \delta = \delta_{AB} + \delta_{BC} + \delta_{CD} = 1.25 + 0.5 - 0.9 = 0.85 \text{ mm}$$

### Complex stresses

In real life, except for a few simple cases, the structural components and machine parts are not as simple, that they would be subjected only to one dimensional (uniaxial force) stress. Instead, components of machineries and structural elements of large and complex civil engineering structures are most likely to be subjected to complex three dimensional stress systems due to the system of forces acting on them. In such situations, the analysis and failure of the structural elements involve analysis of complex stresses.

Even when subjected to uniaxial stress, instead of failure at the plane, normal to the force, the component may fail due to yielding at a different plane due to induced shear stress exceeding the permissible shear stress of the material of the member.

Hence, it is not always the case that the plane normal or parallel to the force would experience the maximum stress. The element under direct tension or compression will experience shear stress and that under shear stress will experience normal stress albeit at different planes.

To obtain a complete picture of the stresses in a bar, we must consider the stresses acting on an “inclined” (as opposed to a “normal”) section through the bar.

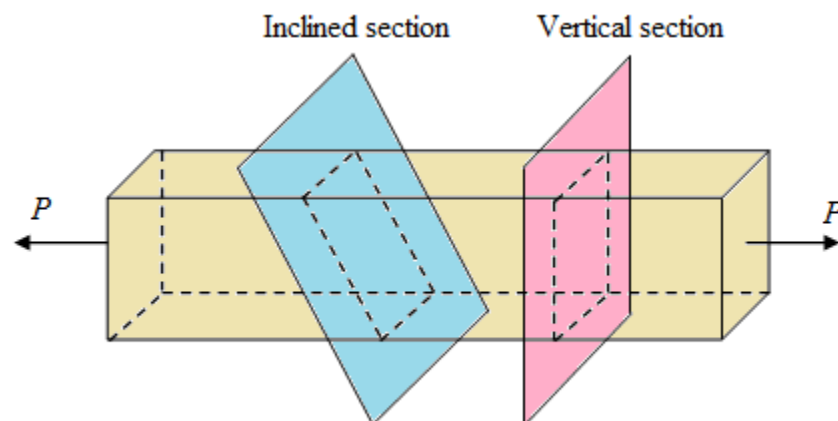


Fig. 1

Because the stresses are the same throughout the entire bar, the stresses on the sections are uniformly distributed.

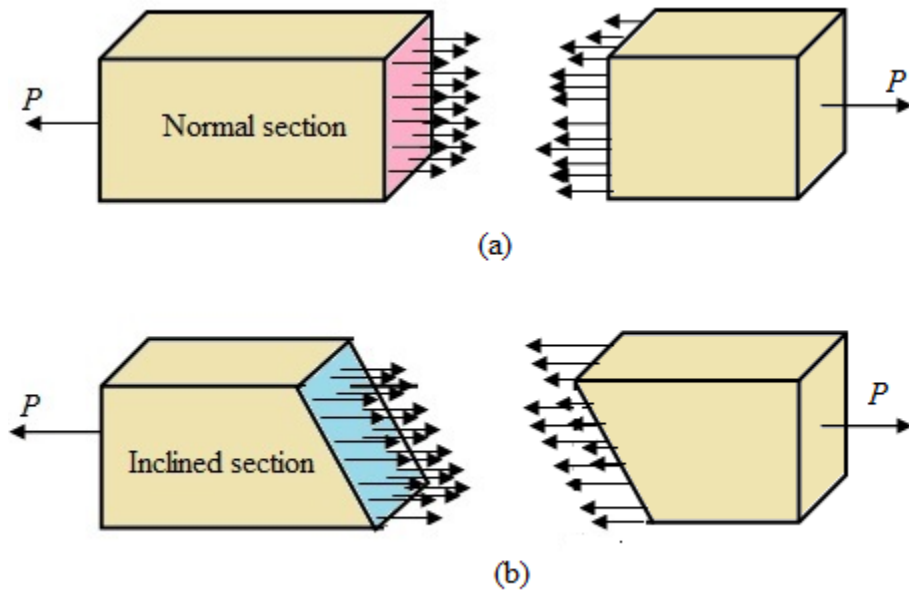


Fig. 2

## Stresses in an oblique plane due to uniaxial normal stress:

In the previous section, we were computing the stresses on a plane when the force was either normal to the plane or tangential to it. In other words, the plane of interest was either normal to the force or parallel to it.

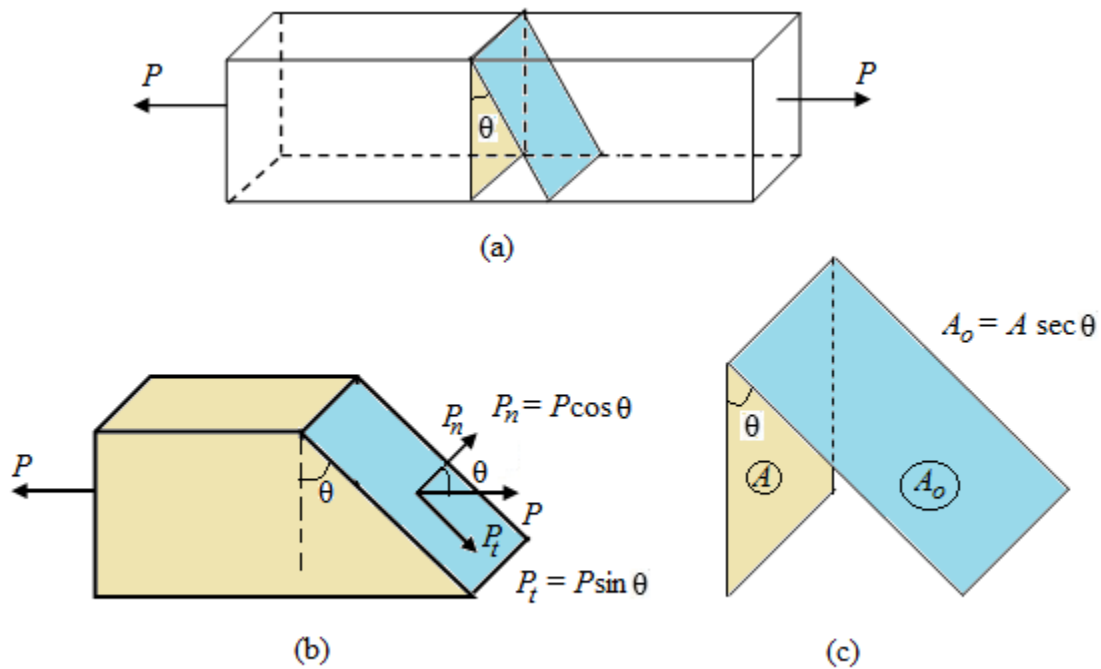


Fig. 3

## Complex stresses

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In this section, the stresses on an oblique plane, i.e. plane making some angle ( $\theta$ ) with the vertical plane, will be dealt with. The normal stress on the vertical plane (perpendicular to the force) is  $\sigma = P/A$ , where  $P$  is the force and  $A$  is the area of cross-section of the plane.

In figure (b), however, there are two components of the force, one normal to the oblique plane ( $P_n$ ) and the other tangential to the plane ( $P_t$ ). Hence, the oblique plane will experience two stresses, one normal stress ( $\sigma_n$ ) and the other tangential stress or shear stress ( $\tau_n$ ). The stress components can be calculated by dividing the respective forces with the area of the oblique plane.

Normal force perpendicular to the oblique plane,  $P_n = P \cos \theta$

Shear force tangential to the oblique plane,  $P_t = P \sin \theta$

Area of the oblique plane,  $A_o = \frac{A}{\cos \theta} = A \sec \theta$

Normal stress, 
$$\sigma_\theta = \frac{P_n}{A_o} = \frac{P \cos \theta}{A \sec \theta} = \frac{P \cos \theta \cdot \cos \theta}{A}$$
$$= \sigma \cos^2 \theta \quad (1)$$

Tangential or shear stress, 
$$\tau_\theta = \frac{P_t}{A_o} = \frac{P \sin \theta}{A \sec \theta} = \frac{P \sin \theta \cos \theta}{A}$$
$$= \frac{\sigma}{2} \sin 2\theta \quad (2)$$

Thus, an oblique plane in a member under axial force will have two components of stresses on it, one normal to the plane while the other is tangential.

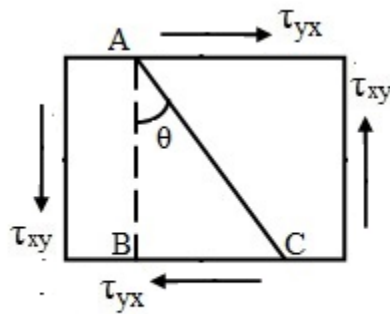
The stresses on the inclined plane, therefore, are not simply the resolutions of  $\sigma$ , perpendicular and tangential to that plane. The direct stress,  $\sigma_\theta$  has a maximum value of  $\sigma$ , when  $\theta = 0^\circ$  whilst the shear stress  $\tau$ , has a maximum value of  $\sigma/2$ , when  $\theta = 45^\circ$ .

## Complex stresses

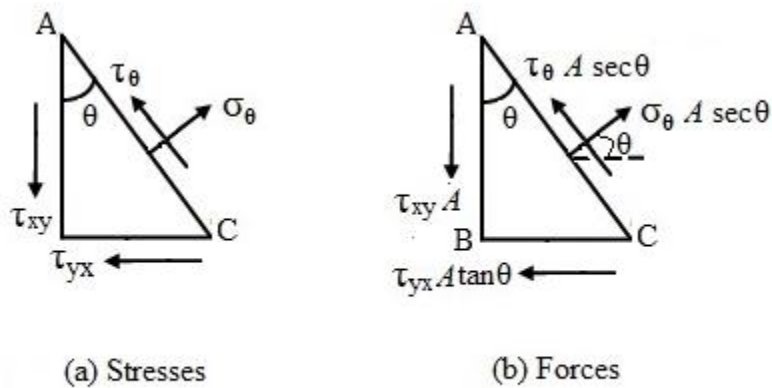
Thus, a material whose yield stress in shear is less than half that in tension or compression will yield initially in shear under the action of direct tensile or compressive forces.

### Stresses in an oblique plane under pure shear stress system:

Consider a rectangular stressed element shown in Fig. 4 to which shear stresses ( $\tau_{xy}$ ) have been applied on the vertical sides so as to produce counterclockwise rotation. *Complementary shear stresses* of equal magnitude ( $\tau_{yx}$ ) but of opposite sense are then set up on the horizontal sides in order to prevent rotation of the element and to keep it in equilibrium.



**Fig. 4 Pure shear system**



**Fig. 5 Free body diagram of triangular prism**

Consider the equilibrium of the triangular prism element in Fig. 5.

Resolving the forces along the direction normal to the plane AC, we have

$$\sigma_{\theta} \cdot A \sec \theta = \tau_{xy} A \sin \theta + \tau_{yx} A \tan \theta \cos \theta$$

$$\sigma_{\theta} = \tau_{xy} \sin \theta \cos \theta + \tau_{yx} \sin \theta \cos \theta$$

## Complex stresses

$$\begin{aligned}
 &= 2\tau_{xy} \sin \theta \cos \theta \\
 &= \tau_{xy} \sin 2\theta
 \end{aligned} \tag{3}$$

The maximum value of  $\sigma$  is  $\tau_{xy}$  when  $\theta = 45^\circ$ .

Resolving the forces along the plane AC, we have

$$\tau_\theta \cdot A \sec \theta + \tau_{yx} A \tan \theta \sin \theta = \tau_{xy} A \cos \theta$$

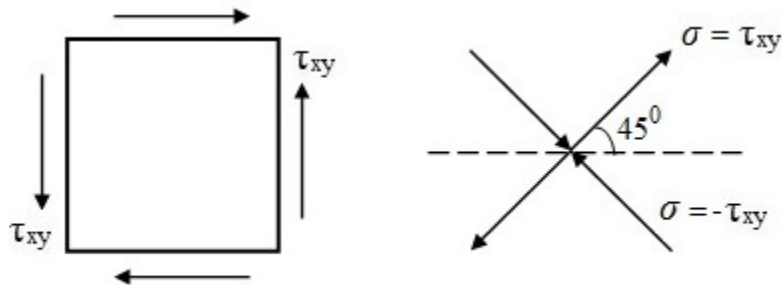
$$\tau_\theta \cdot \sec \theta = \tau_{xy} \cos \theta - \tau_{yx} \tan \theta \sin \theta$$

$$\tau_\theta = \tau_{xy} \cos^2 \theta - \tau_{yx} \sin^2 \theta$$

$$= \tau_{xy} (\cos^2 \theta - \sin^2 \theta)$$

$$= \tau_{xy} \cos 2\theta \tag{4}$$

The maximum value of  $\tau_\theta$ , is  $\tau_{xy}$  when  $\theta = 0^\circ$  or  $90^\circ$  and it has a value of zero when  $\theta = 45^\circ$ , i.e. on the planes of maximum direct stress.



**Fig. 6**

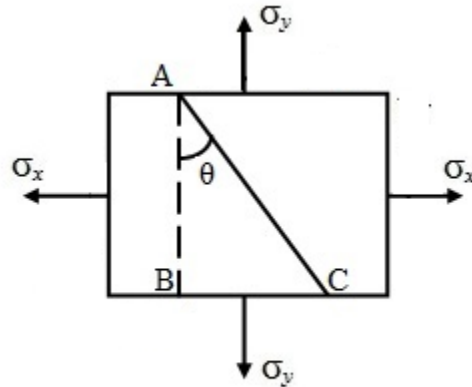
Further consideration of eqn. (3) shows that the system of pure shear stresses produces an equivalent direct stress system as shown in Fig., one set compressive and one tensile, each at  $45^\circ$  to the original shear directions, and equal in magnitude to the applied shear.

### Stresses in an oblique plane under biaxial normal stress system:

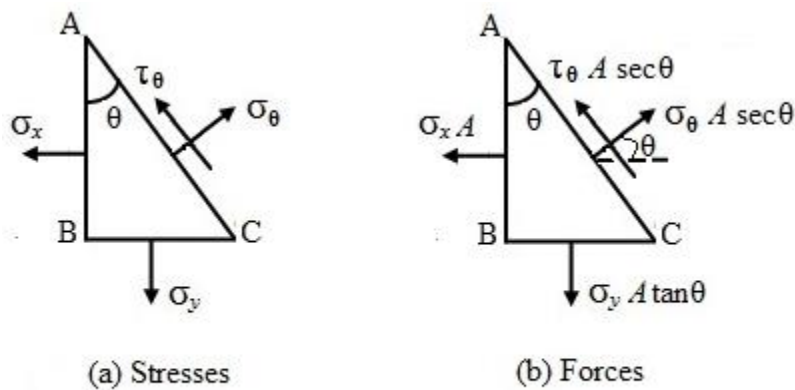
Consider the rectangular element as shown in Fig. 7 subjected to a system of two direct stresses, both tensile, at right angles,  $\sigma_x$  and  $\sigma_y$ .

For equilibrium of the portion  $ABC$ , resolving the forces perpendicular to  $AC$ ,





**Fig.7 Two -dimensional normal stress system**



**Fig.8 Free body diagram of triangular prism**

Consider the equilibrium of the triangular prism element in Fig. 8.

Resolving the forces along the direction normal to the plane AC, we have

$$\begin{aligned}
 \sigma_{\theta} \cdot A \sec \theta &= \sigma_x A \cdot \cos \theta + \sigma_y A \tan \theta \cdot \sin \theta \\
 \sigma_{\theta} &= \sigma_x \cos^2 \theta + \sigma_y \sin^2 \theta \\
 &= \frac{\sigma_x}{2} (1 + \cos 2\theta) + \frac{\sigma_y}{2} (1 - \cos 2\theta) \\
 &= \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta \tag{5}
 \end{aligned}$$

Resolving the forces along the plane AC, we have

$$\tau_{\theta} \cdot A \sec \theta + \sigma_x A \sin \theta = \sigma_y A \tan \theta \cdot \cos \theta$$

## Complex stresses

$$\tau_{\theta} \cdot \sec \theta = -\sigma_x \sin \theta + \sigma_y \tan \theta \cdot \cos \theta$$

$$\tau_{\theta} = -\sigma_x \sin \theta \cos \theta + \sigma_y \sin \theta \cdot \cos \theta$$

$$= -(\sigma_x - \sigma_y) \sin \theta \cos \theta$$

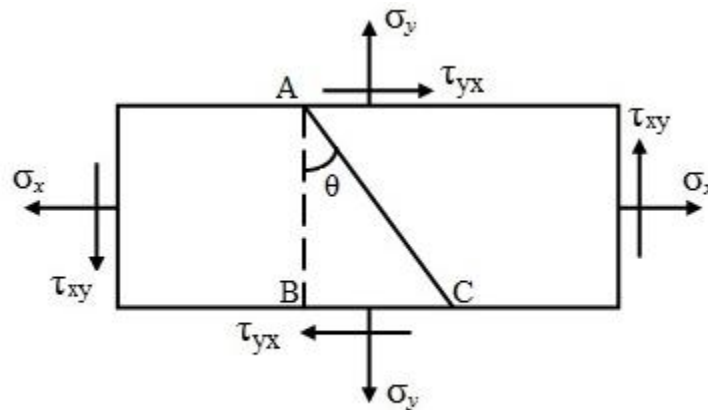
$$= -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta \quad (6)$$

The maximum direct stress will equal  $\sigma_x$  or  $\sigma_y$ , whichever is the greater, when  $\theta = 0$  or  $90^\circ$ . The maximum shear stress occurs in *the plane* when  $\theta = 45^\circ$ .

$$\tau_{\max} = -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta$$

### Stresses in an oblique plane under general two-dimensional stress system:

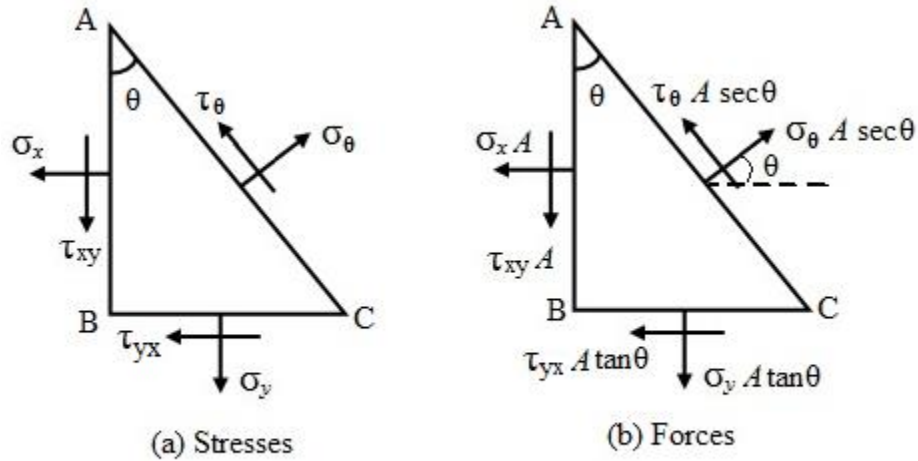
Consider a rectangular prism of uniform cross-sectional area under bi-axial/ two-dimensional stress as shown in the Fig 9.



**Fig. 9 Rectangular block under two-dimensional stress**

### Sign convention:

1. Angle of obliquity is measured in counter-clockwise direction with respect to vertical plane; the reference plane is taken as positive.
2. Tensile stress is taken as positive and compressive stress as negative.
3. Shear stress on vertical reference plane producing anticlockwise rotation is taken as positive.



**Fig.10 Free body diagram of triangular prism**

Consider the equilibrium of the triangular prism element in Fig. 10.

Resolving the forces along the direction normal to the plane AC, we have

$$\begin{aligned}
 \sigma_{\theta} \cdot A \sec \theta &= \sigma_x A \cdot \cos \theta + \tau_{xy} A \sin \theta + \sigma_y A \tan \theta \cdot \sin \theta + \tau_{yx} A \tan \theta \cos \theta \\
 \sigma_{\theta} &= \sigma_x \cos^2 \theta + \sigma_y \sin^2 \theta + \tau_{xy} \sin \theta \cos \theta + \tau_{yx} \sin \theta \cos \theta \\
 &= \sigma_x \cos^2 \theta + \sigma_y \sin^2 \theta + 2\tau_{xy} \sin \theta \cos \theta \\
 &= \frac{\sigma_x}{2} (1 + \cos 2\theta) + \frac{\sigma_y}{2} (1 - \cos 2\theta) + \tau_{xy} \sin 2\theta \\
 &= \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta \quad (7)
 \end{aligned}$$

Resolving the forces along the plane AC, we have

$$\begin{aligned}
 \tau_{\theta} \cdot A \sec \theta + \sigma_x A \sin \theta + \tau_{yx} A \tan \theta \sin \theta &= \sigma_y A \tan \theta \cdot \cos \theta + \tau_{xy} A \cos \theta \\
 \tau_{\theta} \cdot \sec \theta &= -\sigma_x \sin \theta - \tau_{yx} \tan \theta \sin \theta + \sigma_y \tan \theta \cdot \cos \theta + \tau_{xy} \cos \theta \\
 \tau_{\theta} &= -\sigma_x \sin \theta \cos \theta - \tau_{yx} \sin^2 \theta + \sigma_y \sin \theta \cdot \cos \theta + \tau_{xy} \cos^2 \theta \\
 &= -(\sigma_x - \sigma_y) \sin \theta \cos \theta + \tau_{xy} (\cos^2 \theta - \sin^2 \theta)
 \end{aligned}$$

## Complex stresses

---

$$= -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta + \tau_{xy} \cos 2\theta \quad (8)$$

The maximum and minimum normal stresses ( $\sigma_1$  and  $\sigma_2$ ) are known as the **principal stresses**. The *maximum* and *minimum* normal stresses ( $\sigma_1$  and  $\sigma_2$ ) which occur on any plane in the member can now be determined as follows:

For  $\sigma_\theta$  to be a maximum or minimum,  $\frac{d\sigma_\theta}{d\theta} = 0$

We have 
$$\sigma_\theta = \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta$$

$$\frac{d\sigma_\theta}{d\theta} = -(\sigma_x - \sigma_y) \sin 2\theta + 2\tau_{xy} \cos 2\theta = 0$$

Since we are dealing with maximum and minimum normal stresses (principal stress), let us denote this angle as  $\theta_p$ . The above equation can be written as,

$$\tan 2\theta_p = \frac{2\tau_{xy}}{(\sigma_x - \sigma_y)} \quad (9)$$

There are two values of  $2\theta_p$  in the range 0-360°, with values differing by 180°. There are two values of  $\theta_p$  in the range 0-180°, with values differing by 90°. So, the planes on which the maximum and minimum normal stresses act are mutually perpendicular and are known as **principal planes**.

We can now solve for the principal stresses by substituting for  $\theta_p$  in the normal stress equation for  $\sigma_\theta$ .

From triangle I and II of Fig. 11,

$$\sin 2\theta_p = \pm \frac{2\tau_{xy}}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\cos 2\theta_p = \pm \frac{(\sigma_x - \sigma_y)}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

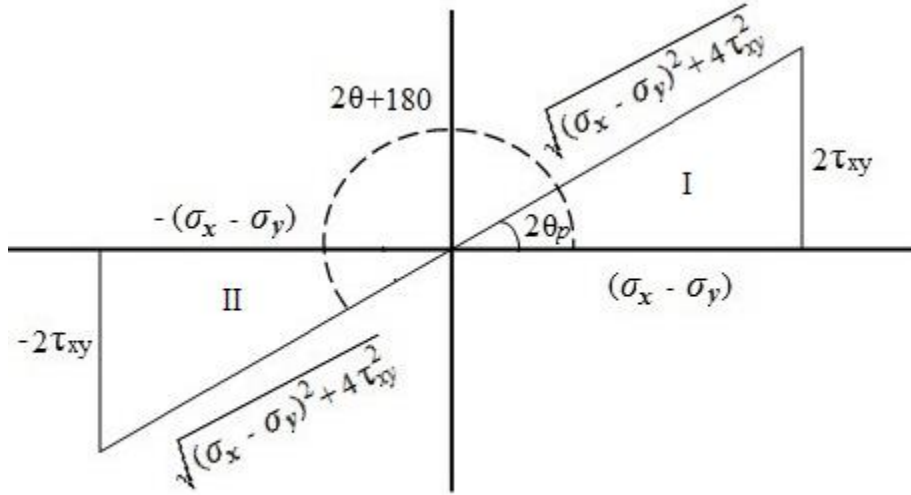


Fig. 11

Substituting the values of  $\theta_p$  in (1), the maximum and minimum normal stresses are given by

$$\sigma_{1,2} = \frac{\sigma_x + \sigma_y}{2} \pm \frac{(\sigma_x - \sigma_y)}{2} \frac{(\sigma_x - \sigma_y)}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}} \pm \tau_{xy} \frac{2\tau_{xy}}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\sigma_{1,2} = \frac{\sigma_x + \sigma_y}{2} \pm \frac{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\sigma_{1,2} = \frac{1}{2}(\sigma_x + \sigma_y) \pm \frac{1}{2}\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2} \quad (10)$$

We have derived the maximum and minimum values of the normal stresses denoted as  $\sigma_1$  (maximum) and  $\sigma_2$  (minimum).

To find out which principal stress is associated with which principal angle, we could use the equations for  $\sin \theta_p$  and  $\cos \theta_p$  or for  $\sigma_\theta$ .

Similarly substituting the value of  $\theta_p$  in eqn. (8),

$$\tau_{\theta_p} = \mp \frac{(\sigma_x - \sigma_y)}{2} \frac{2\tau_{xy}}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}} \pm \tau_{xy} \frac{(\sigma_x - \sigma_y)}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\tau_{\theta_p} = 0$$

Hence, the shear stresses are zero on the principal planes.

## Complex stresses

---

Now let us find out the maximum value of the shear stress. Differentiate eqn. (8) with respect to  $\theta$  and equate it to zero. We get,

$$\frac{d\tau_{\theta}}{d\theta} = -\frac{(\sigma_x - \sigma_y)}{2} \cdot 2 \cos 2\theta - 2\tau_{xy} \sin 2\theta = 0$$
$$\tau_{xy} \sin 2\theta = -\frac{(\sigma_x - \sigma_y)}{2} \cdot \cos 2\theta$$

Since we are dealing with shear, let us denote this angle as  $\theta_s$ . The above equation can be written as,

$$\tan 2\theta_s = -\frac{(\sigma_x - \sigma_y)}{2\tau_{xy}} \quad (11)$$

There are two values of  $2\theta_s$  in the range 0-360°, with values differing by 180°. There are two values of  $\theta_s$  in the range 0-180°, with values differing by 90°. So, the planes on which the maximum shear stresses act are mutually perpendicular.

Because shear stresses on perpendicular planes have equal magnitudes, the maximum positive and negative shear stresses differ only in sign.

Comparing the equation (4) for  $\theta_s$  with that the equation (5) for  $\theta_p$ , it is observed that both are reciprocal of each other. So we can write,

$$\tan 2\theta_s = -\frac{1}{\tan 2\theta_p} = -\cot 2\theta_p$$

$$\tan 2\theta_s = \tan(90 + 2\theta_p)$$

$$\theta_s = 45 + \theta_p$$

So, the planes of maximum shear stress ( $\theta_s$ ) occur at 45° to the principal planes ( $\theta_p$ ). We have therefore derived maximum & minimum values of principal stresses, their angles, maximum values of shear stress and its orientation with respect to principal planes.

Further, from triangle I and II of Fig. 12,

## Complex stresses

$$\sin 2\theta_p = \mp \frac{(\sigma_x - \sigma_y)}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\cos 2\theta_p = \pm \frac{2\tau_{xy}}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

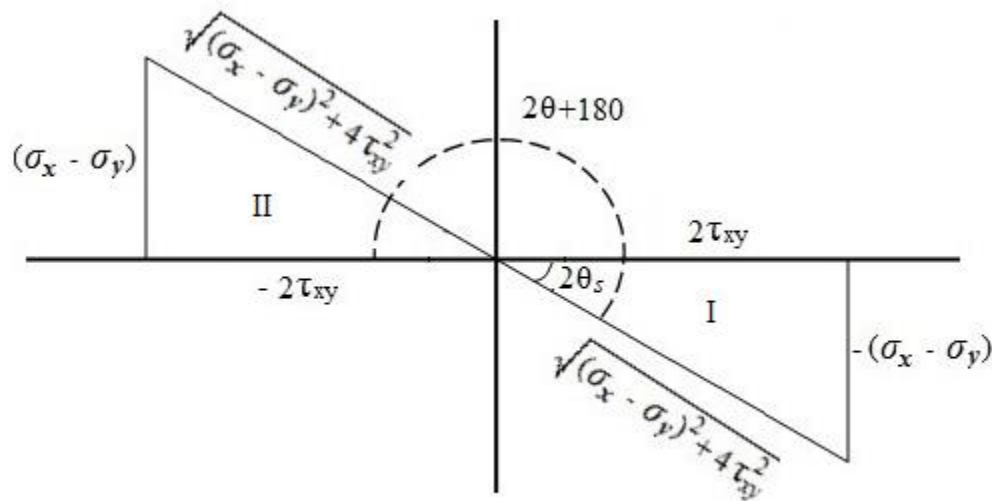


Fig. 12

Substituting the values of  $\theta_s$  in (2), the maximum and minimum shear stresses are given by

$$\tau_{\max} = \pm \frac{(\sigma_x - \sigma_y)}{2} \cdot \frac{(\sigma_x - \sigma_y)}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}} \pm \tau_{xy} \frac{2\tau_{xy}}{\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}}$$

$$\tau_{\max} = \pm \frac{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}{2\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}} = \pm \frac{1}{2} \sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}$$

$$\tau_{\max} = \pm \frac{1}{2} \sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}$$

In a structural member under biaxial complex stress system, there exist two mutually perpendicular planes on which the normal stress is either maximum or minimum. These planes are known as *principal planes*. The shear stress on these planes is zero.

A little observation will show that,  $\tau_{\max} = \pm \frac{1}{2}(\sigma_1 - \sigma_2)$ , where  $\sigma_1$  and  $\sigma_2$  are principal stresses.

## Complex stresses

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Further, it can be seen that the sum of the principal stresses is same as the sum of the normal stresses in any two mutually perpendicular directions.

$$\sigma_1 + \sigma_2 = \sigma_x + \sigma_y$$

### Transformation of stress coordinates:

There exists only one intrinsic/fundamental state of stress at a point in a stressed body akin to the position of a point on a plane. As the coordinates of a point  $(x, y)$  on a plane changes with the orientation of coordinate axes of reference, so does the stress coordinates  $(\sigma, \tau)$  of a point in a stressed body (with respect to the plane of reference) with the orientation of its plane of consideration.

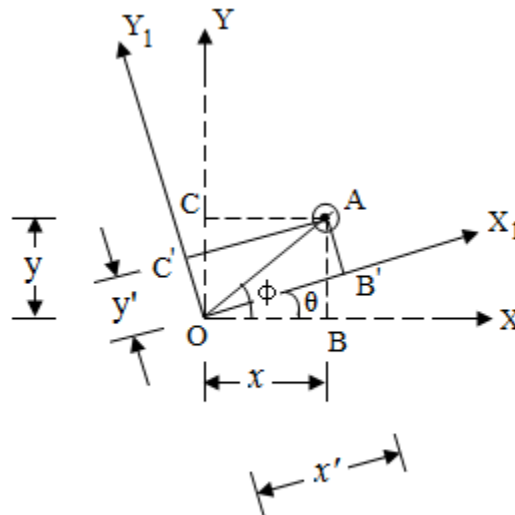


Fig. 13

'A' is a point on a plane with coordinate axes  $OX$  and  $OY$  as shown in Fig. 13. The coordinates of A is  $(x, y)$ . If the orientation of axes of reference  $OX$  and  $OY$  is changed by an angle of  $\theta$  in counterclockwise direction, then the position is not going to be changed. Instead its coordinates  $(x_1, y_1)$  in respect of changed coordinate axes will change.

$d$  = distance of A from the origin  $O$  of the plane

$\phi$  = angular distance of A from the origin  $O$  of the plane

$$x = d \cos \phi, y = d \sin \phi$$



## Complex stresses

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$$x_1 = d \cos(\phi - \theta), y_1 = d \sin(\phi - \theta)$$

It is evident that only the coordinates of the point and not its inherent position depends on the orientation of the axes of reference. Similarly, regardless of the orientation of the element used to portray the state of stress, the intrinsic state of stress remains the same. In other words the stress coordinates  $(\sigma, \tau)$  and not the inherent state of stress of a point change.

### Member subjected to principal stresses:

### Mohr's circle:

Mohr's circle is a graphical representation of stress transformation equations. The equations of stress transformation describe a circle if normal stress and shear stress are represented as abscissa and ordinate respectively. Each point on the circumference of Mohr's circle represents a plane through the centre of the circle and the coordinates  $(\sigma, \tau)$  of the point represents the normal stress  $(\sigma)$  and shear stress  $(\tau)$  on the given plane.

Mohr's circle can be drawn from a given state of stress at a point in a structural member. Consider a stress element representing the state of stress at a point as shown in the Fig. 14.

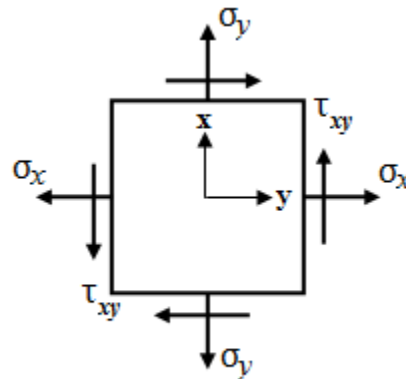


Fig. 14

Stress transformation equations for normal and tangential components on a plane are given by

$$\text{Normal stress on the plane, } \sigma_{\theta} = \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta \quad (12)$$

$$\text{Shear stress on the plane, } \tau_{\theta} = -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta + \tau_{xy} \cos 2\theta \quad (13)$$

## Complex stresses

---

Rearranging the equation (12), we have

$$\sigma_{\theta} - \frac{\sigma_x + \sigma_y}{2} = \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta \quad (14)$$

Squaring both sides of equation (13) and (14) and adding them together, we have

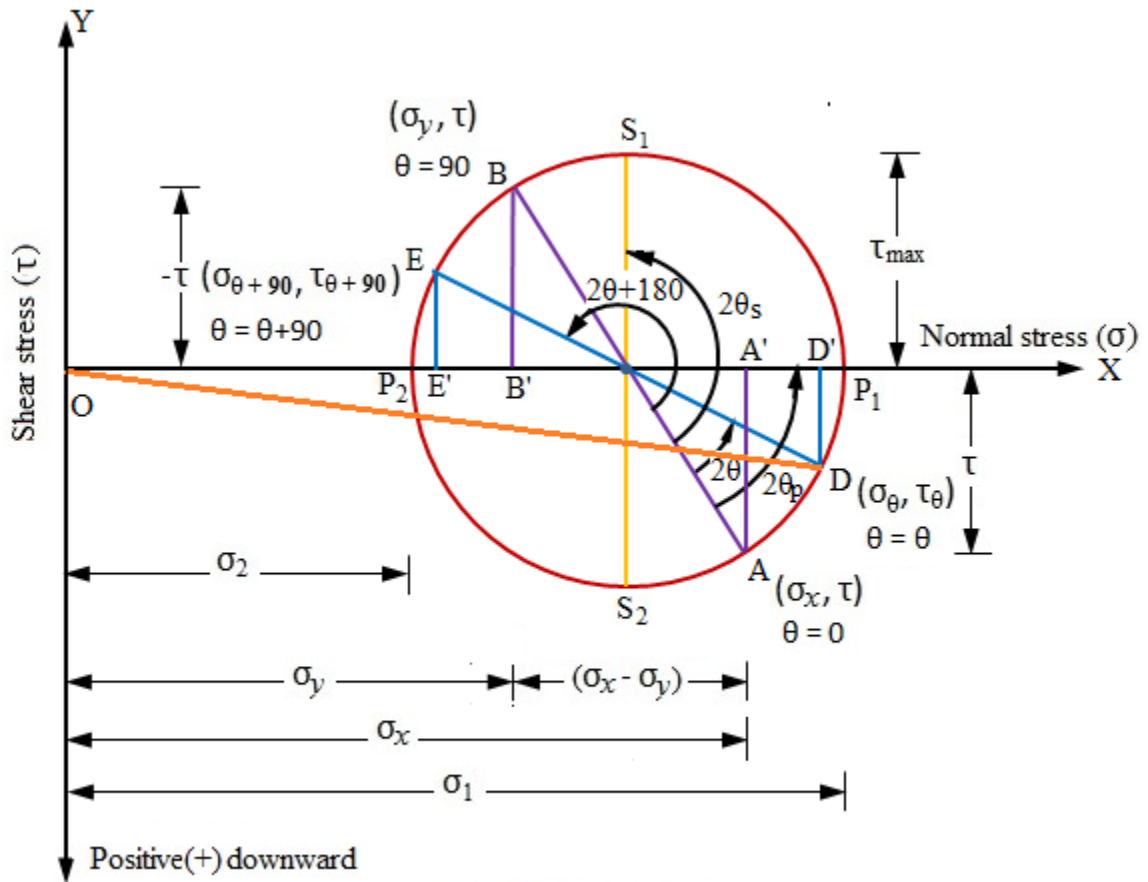
$$\left( \sigma_{\theta} - \frac{\sigma_x + \sigma_y}{2} \right)^2 + \tau_{\theta}^2 = \left( \frac{\sigma_x - \sigma_y}{2} \right)^2 + \tau_{xy}^2$$
$$\left( \sigma_{\theta} - \frac{\sigma_x + \sigma_y}{2} \right)^2 + \tau_{\theta}^2 = \left( \sqrt{\left( \frac{\sigma_x - \sigma_y}{2} \right)^2 + \tau_{xy}^2} \right)^2$$

This is the equation of a circle with centre  $\left( \frac{\sigma_x + \sigma_y}{2}, 0 \right)$  and radius  $R = \sqrt{\left( \frac{\sigma_x - \sigma_y}{2} \right)^2 + \tau_{xy}^2}$  and

this circle is known as Mohr's circle named after the German Civil Engineer Otto Mohr (1835-1918). It provides a simple and clear picture of an otherwise complicated analysis.

### Procedure for drawing Mohr's circle:

1. Draw coordinates axes in Cartesian coordinate system with  $O$  as origin, normal stress ( $\sigma$ ) as abscissa (positive to the right) and shear stress ( $\tau$ ) as ordinate (positive downward).
2. Locate the centre  $C$  of the circle at the point having coordinates  $\left( \frac{\sigma_x + \sigma_y}{2}, 0 \right)$ .
3. Locate point  $A$ , representing the state of stress on the vertical plane, i.e., face  $x$  of the element by plotting its coordinates  $\sigma_x$  and  $\tau$ . Point  $A$  on the circle corresponds to  $\theta = 0^\circ$  and represents the vertical plane.
4. Locate point  $B$ , representing the state of stress on the horizontal plane, i.e., face  $y$  of the element by plotting its coordinates  $\sigma_y$  and  $-\tau$ . Point  $A$  on the circle corresponds to  $\theta = 90^\circ$  and represents the horizontal plane.
5. Join  $AB$  so as to intersect the normal stress axis at  $C$ .
6. With the point  $C$  as the centre and  $CA (= CB)$  as radius, draw Mohr's circle through points  $A$  and  $B$ . This is the required Mohr's circle which has radius  $R$ .



**Fig. 15 Mohr's circle**

Every point on the circumference of the circle then represents a state of stress on some plane through  $C$ .

The stress state on an inclined plane with an angle  $\theta$  is represented at point  $D$  on the Mohr's circle, which is measured an angle  $2\theta$  counter-clockwise from point  $A$  to show the coordinate at  $D$ .

Consider any point  $D$  on the circumference of the circle, such that  $CD$  makes an angle  $2\theta$  with  $CA$ , and drop a perpendicular from  $D$  to meet the  $\sigma$  axis at  $D'$ .

Coordinates of  $D$ :

$$OD' = OC + CD' = \frac{1}{2}(\sigma_x + \sigma_y) + R \cos(2\theta_p - 2\theta)$$

## Complex stresses

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$$= \frac{1}{2}(\sigma_x + \sigma_y) + R \cos 2\theta_p \cos 2\theta + R \sin 2\theta_p \sin 2\theta$$

But  $R \cos 2\theta_p = \frac{1}{2}(\sigma_x - \sigma_y)$  and  $R \sin 2\theta_p = \tau_{xy}$

Therefore,  $OD' = \frac{1}{2}(\sigma_x + \sigma_y) + \frac{1}{2}(\sigma_x - \sigma_y)\cos 2\theta + \tau_{xy} \sin 2\theta$

On inspection this is seen to be eqn. (12) for the normal stress  $\sigma_\theta$  on the plane inclined at  $\theta$  to the vertical plane  $AB$ .

Similarly,  $DD' = R \sin(2\theta_p - 2\theta)$

$$DD' = R \sin 2\theta_p \cos 2\theta - R \cos 2\theta_p \sin 2\theta$$

$$DD' = -\frac{1}{2}(\sigma_x - \sigma_y)\sin 2\theta + \tau_{xy} \cos 2\theta$$

Again, on inspection this is seen to be eqn. (13) for the shear stress  $\tau_\theta$  on the plane inclined at  $\theta$  to the vertical plane  $AB$ .

Thus the coordinates of  $Q$  are the normal and shear stresses on a plane inclined at  $\theta$  to  $AB$  in the original stress system.

### Characteristics of Mohr's circle

1. The direct stress is maximum when  $D$  is at  $P_1$ , i.e.  $OP_1$  is the length representing the maximum principal stress  $\sigma_1$  and  $2\theta_p$  gives the angle of the plane  $\theta_p$ , from  $AB$ . Similarly,  $OP_2$  is the other principal stress.
2. The maximum shear stress is given by the highest point on the circle and is represented by the radius of the circle. This follows since shear stresses and complementary shear stresses have the same value; *therefore the centre of the circle will always lie on the  $\sigma$ -axis midway between  $\sigma_x$  and  $\sigma_y$ .*

## Complex stresses

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3. From the above point the direct stress on the plane of maximum shear must be midway between  $\sigma_x$  and  $\sigma_y$ , i.e.  $\frac{1}{2}(\sigma_x + \sigma_y)$ .
4. The shear stress on the principal planes is zero.
5. Since the resultant of two stresses at  $90^\circ$  can be found from the parallelogram of vectors as the diagonal, as shown in Fig. **13.10**, the resultant stress on the plane at  $\theta$  to  $AB$  is given by  $OD$  on Mohr's circle.
6. Comparing the Mohr's circle and the stress element, it is observed points  $S_1$  and  $S_2$  representing the points of maximum and minimum shear stresses, are located on the circle at  $90^\circ$  from points  $P_1$  and  $P_2$  i.e. the planes of maximum and minimum shear stress are at  $45^\circ$  to the principal planes, and

The graphical method of solution of complex stress problems using Mohr's circle is a very powerful technique since all the information relating to any plane within the stressed element is contained in the single construction. It thus provides a convenient and rapid means of solution which is less prone to arithmetical errors and is highly recommended.

## Complex stresses

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### Numerical

1. A material has permissible stresses in tension, compression and shear as  $30 \text{ N/mm}^2$ ,  $90 \text{ N/mm}^2$  and  $25 \text{ N/mm}^2$  respectively. If specimens of diameter  $20 \text{ mm}$  are tested in tension and compression, identify the failure surfaces and failure load.

#### Solution:

*Case I (Test in tension):* If subjected to full tensile strength,

$$\text{Maximum tensile stress, } \sigma_t = 30 \text{ N/mm}^2$$

$$\text{Corresponding maximum stress in shear, } \tau_{\max} = \frac{\sigma_t}{2} = \frac{30}{2} = 15 \text{ N/mm}^2 < 25 \text{ N/mm}^2$$

Hence, the failure will occur due to tension. The maximum tensile stress is in the axial direction. Hence failure will occur on the plane of axial tensile stress, i.e., at right angle to the stress.

Corresponding tensile force,  $P_t = \text{Area of specimen} \times \text{tensile strength}$

$$\begin{aligned} &= A \times \sigma_t = \frac{\pi}{4} \times 20^2 \times 30 \\ &= 9224.778 \text{ N} \end{aligned}$$

*Case II (Test in compression):*

$$\text{Maximum compressive stress, } \sigma_c = 90 \text{ N/mm}^2$$

$$\text{Corresponding maximum stress in shear, } \tau_{\max} = \frac{\sigma_c}{2} = \frac{90}{2} = 45 \text{ N/mm}^2 > 25 \text{ N/mm}^2$$

Hence, the failure will occur due to shear. The failure will occur on the plane at  $45^\circ$  to the plane of axial stress.

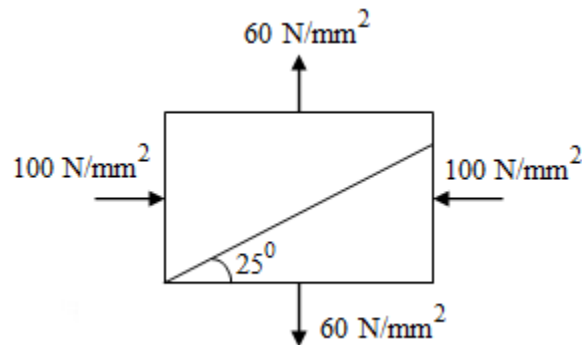
Corresponding compressive stress causing shear failure,  $\sigma_{cf} = 2 \times 25 = 50 \text{ N/mm}^2$

Corresponding compressive force,  $P_c = \text{Area of specimen} \times \text{compressive stress}$

$$\begin{aligned} &= A \times \sigma_{cf} = \frac{\pi}{4} \times 20^2 \times 50 \\ &= 15707.93 \text{ N} \end{aligned}$$

## Complex stresses

2. Normal stresses acting at a point in a strained material are  $100 \text{ N/mm}^2$  compressive and  $60 \text{ N/mm}^2$  tensile as shown in the figure. Find the stresses on the given oblique plane.

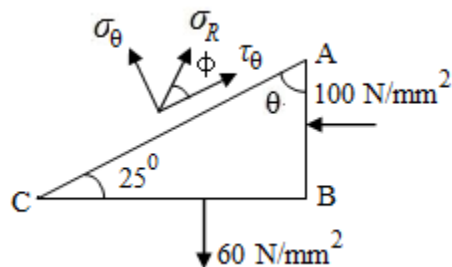
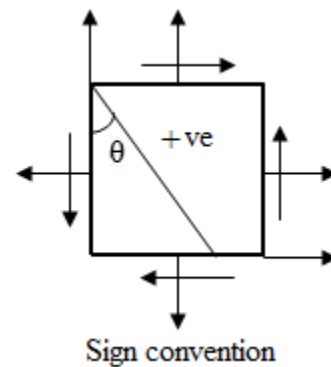


**Solution:** Sign convention:

Define the stresses in terms of the sign convention:

$$\sigma_x = -100 \text{ MPa}, \sigma_y = 60 \text{ MPa}$$

Angle of orientation of plane of plane AC,  $\theta = -65^\circ$   
(clockwise)



$$\begin{aligned} \text{Normal stress on the oblique plane, } \sigma_\theta &= \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta \\ &= \frac{-100 + 60}{2} + \frac{-100 - 60}{2} \cos 2(-65^\circ) \\ &= -20 - 80 \cos(-130^\circ) \\ &= 31.423 \text{ N/mm}^2 \text{ (tensile)} \end{aligned}$$

$$\text{Shear stress on the oblique plane, } \tau_\theta = -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta$$

## Complex stresses

$$\begin{aligned}
 &= -\frac{(-100 - 60)}{2} \sin 2(-65^\circ) \\
 &= 80 \sin(-130^\circ) \\
 &= -61.284 \text{ N/mm}^2 \text{ (Clockwise)}
 \end{aligned}$$

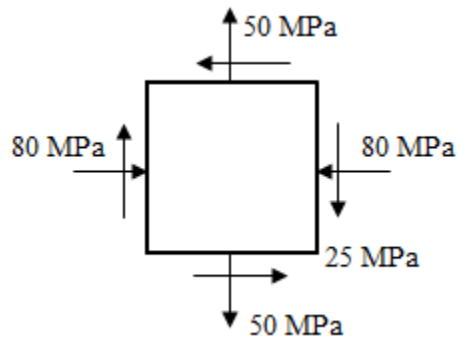
Resultant stress on the plane,  $\sigma_R = \sqrt{\sigma_\theta^2 + \tau_\theta^2} = \sqrt{31.432^2 + (-61.284)^2}$

$$= 68.87 \text{ N/mm}^2$$

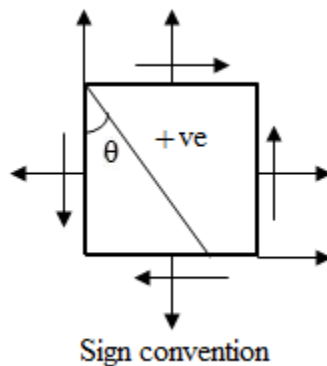
Angle of obliquity,

$$\begin{aligned}
 \phi &= \tan^{-1}\left(\frac{\sigma_\theta}{\tau_\theta}\right) = \tan^{-1}\left(\frac{31.423}{61.284}\right) \\
 &= 27.14^\circ
 \end{aligned}$$

3. The state of plane stress at a point is represented by the stress element below. Determine the stresses acting on plane oriented  $30^\circ$  clockwise with respect to the vertical plane.



**Solution:** Sign convention:





## Complex stresses

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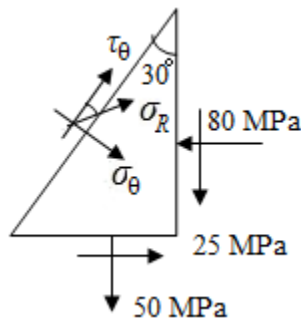
Define the stresses in terms of the sign convention:

$$\sigma_x = -80 \text{ MPa}, \sigma_y = 50 \text{ MPa} \text{ and } \tau_{xy} = -25 \text{ MPa}$$

Angle of orientation of plane,  $\theta = -30^\circ$

Normal stress on the given plane,  $\sigma_\theta = \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta$

$$\begin{aligned} &= \frac{-80 + 50}{2} + \frac{-80 - 50}{2} \cos 2(-30) + (-25) \sin 2(-30) \\ &= \frac{-30}{2} + \frac{-130}{2} \cos(-60) - 25 \sin(-60) \\ &= -15 - 65 \cos(-60) - 25 \sin(-60) \\ &= -15 - 65 \cos(-60) - 25 \sin(-60) \\ &= -25.9 \text{ MPa (Compression)} \end{aligned}$$



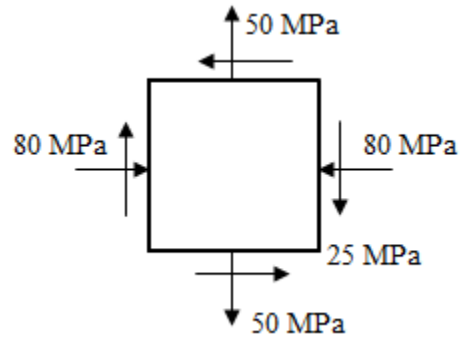
Shear stress on the oblique plane,  $\tau_\theta = -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta + \tau_{xy} \cos 2\theta$

$$\begin{aligned} &= -\frac{(-80 - 50)}{2} \sin 2(-30) + (-25) \cos 2(-30) \\ &= 65 \times \sin(-60) - 25 \cos(-60) \\ &= -68.80 \text{ MPa (Clockwise)} \end{aligned}$$

## Complex stresses

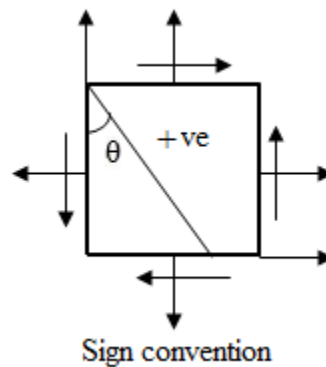
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4. Example: The state of plane stress at a point is represented by the stress element below. Determine the principal stresses and draw the corresponding stress element.



### Solution:

Sign convention:



Define the stresses in terms of the sign convention:

$$\sigma_x = -80 \text{ MPa}, \sigma_y = 50 \text{ MPa} \text{ and } \tau_{xy} = -25 \text{ MPa}$$

Angle of orientation of plane,  $\theta = -30^\circ$

$$\begin{aligned} \text{Principal stresses, } \sigma_{1,2} &= \frac{1}{2}(\sigma_x + \sigma_y) \pm \frac{1}{2}\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2} \\ &= \frac{1}{2}(-80 + 50) \pm \frac{1}{2}\sqrt{(-80 - 50)^2 + 4(-25)^2} \\ &= -15 \pm \frac{1}{2}\sqrt{(-130)^2 + 4 \times 625} \end{aligned}$$

## Complex stresses

---

$$= -15 \pm 69.64$$

$$\sigma_1 = 54.64 \text{ MPa and } \sigma_2 = -84.64 \text{ MPa}$$

$$\text{Angle of principal plane, } \tan 2\theta_p = \frac{2\tau_{xy}}{(\sigma_x - \sigma_y)} = \frac{2(-25)}{(-80 - 50)}$$

$$\tan 2\theta_p = 0.3846$$

$$2\theta_p = 21.0^\circ \text{ and } 21.0^\circ + 180^\circ$$

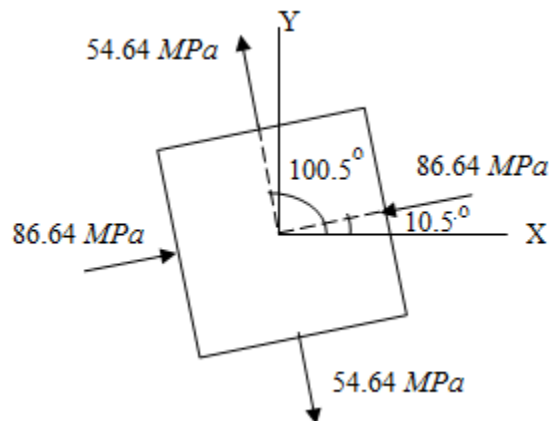
$$\theta_p = 10.5^\circ \text{ and } 100.5^\circ$$

Check for which angle goes with which principal stress. Put  $\theta = 10.5^\circ$  in the stress equation,

$$\begin{aligned}\sigma_\theta &= \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta \\ &= \frac{-80 + 50}{2} + \frac{-80 - 50}{2} \cos 2(10.5^\circ) + (-25) \sin 2(10.5^\circ) \\ &= -84.6 \text{ MPa}\end{aligned}$$

$$\sigma_1 = 54.64 \text{ MPa with } \theta_{p1} = 100.5^\circ$$

$$\sigma_2 = -84.64 \text{ MPa with } \theta_p = 10.5^\circ$$



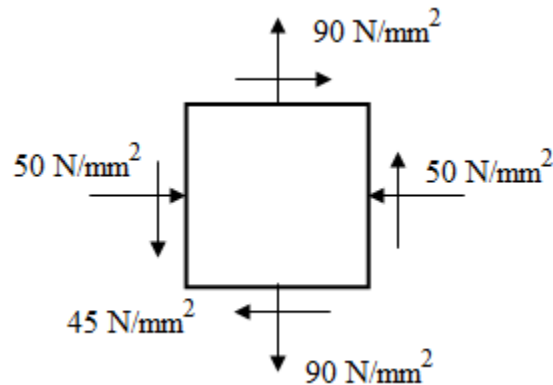
5. A point in a stressed structural member is subjected to a tensile stress of  $90 \text{ N/mm}^2$  on a horizontal plane and compressive stress of  $50 \text{ N/mm}^2$  on the vertical plane. There is a shear

## Complex stresses

---

stress of  $45 \text{ N/mm}^2$  such that when on vertical plane, it tends to rotate the member in counterclockwise direction. Determine the maximum shear stress and also the resultant stress on the planes of maximum shear stresses.

**Solution:**



Define the stresses in terms of the sign convention:

$$\sigma_x = -50 \text{ MPa}, \sigma_y = 90 \text{ MPa} \text{ and } \tau_{xy} = 45 \text{ MPa}$$

Angle of orientation of plane,  $\theta = -30^\circ$

$$\begin{aligned} \text{Maximum shear stress, } \tau_{\max} &= \frac{1}{2} \sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2} \\ &= \frac{1}{2} \sqrt{(-50 - 90)^2 + 45^2} = 73.53 \text{ N/mm}^2 \text{ (counterclockwise)} \end{aligned}$$

Inclination of plane of maximum shear to the vertical plane,

$$\tan 2\theta_s = -\frac{(\sigma_x - \sigma_y)}{2\tau_{xy}} = -\frac{(-50 - 90)}{2 \times 45}$$

$$\tan 2\theta_s = -1.555$$

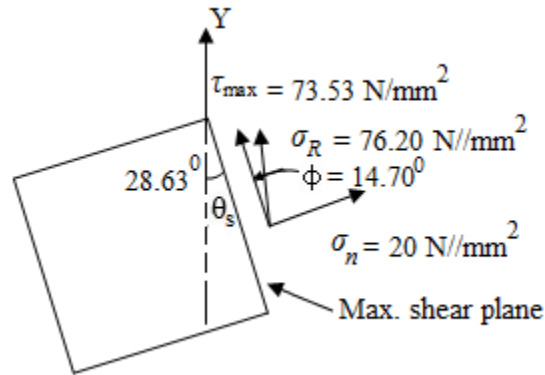
$$2\theta_s = \tan^{-1}(1.555) = 57.26^\circ \text{ and } 237.26^\circ$$

$$\theta_s = 28.63^\circ \text{ and } 118.63^\circ \text{ (counterclockwise)}$$

Normal stress on plane of maximum shear stress,

## Complex stresses

$$\begin{aligned}\sigma_n &= \frac{-50 + 90}{2} + \frac{-50 - 90}{2} \cos 2(28.63^\circ) + 45 \sin 2(28.63^\circ) \\ &= 20 \text{ N/mm}^2\end{aligned}$$



$$\begin{aligned}\text{Resultant stress on the plane, } \sigma_R &= \sqrt{\sigma_\theta^2 + \tau_\theta^2} = \sqrt{20^2 + 73.53^2} \\ &= 76.20 \text{ N/mm}^2 \\ \phi &= \tan^{-1} \left( \frac{\sigma_n}{\tau_{\max}} \right) = \tan^{-1} \left( \frac{20}{76.20} \right) \\ &= 14.70^\circ\end{aligned}$$

### Summary

1. The normal stress  $\sigma$  and shear stress  $\tau$  on oblique planes resulting from direct loading are

$$\sigma_\theta = \sigma \cos^2 \theta$$

$$\tau_\theta = \frac{\sigma}{2} \sin 2\theta$$

2. The stresses on oblique planes owing to a complex stress system are:

$$\text{Normal stress, } \sigma_\theta = \frac{\sigma_x + \sigma_y}{2} + \frac{\sigma_x - \sigma_y}{2} \cos 2\theta + \tau_{xy} \sin 2\theta$$

$$\text{Normal stress, } \tau_\theta = -\frac{(\sigma_x - \sigma_y)}{2} \sin 2\theta + \tau_{xy} \cos 2\theta$$

## Complex stresses

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3. The *principal stresses* (i.e. the maximum and minimum direct stresses) are then

$$\sigma_{1,2} = \frac{1}{2}(\sigma_x + \sigma_y) \pm \frac{1}{2}\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}$$

and these occur on planes at an angle  $\theta_p$  to the vertical plane on which  $\sigma_x$ , acts, given by either

$$\tan 2\theta_p = \frac{2\tau_{xy}}{(\sigma_x - \sigma_y)}$$

where  $\sigma_p = \sigma_1$  or  $\sigma_2$ , the planes being termed *principal planes*. The principal planes are always at  $90^\circ$  to each other, and the *planes of maximum shear* are then located at  $45^\circ$  to them.

4. The *maximum shear stress* is

$$\tau_{\max} = \pm \frac{1}{2}\sqrt{(\sigma_x - \sigma_y)^2 + 4\tau_{xy}^2}$$

$$\tau_{\max} = \pm \frac{1}{2}(\sigma_1 - \sigma_2)$$

5. Normal stress on plane of maximum shear =  $\frac{1}{2}(\sigma_x + \sigma_y)$

6. Shear stress on plane of maximum direct stress (principal plane) = 0

7. In problems where the principal stress in the third dimension  $\sigma_3$  either is known or can be assumed to be zero, the true maximum shear stress is then

$$\frac{1}{2}(\text{greatest principal stress} - \text{least principal stress})$$

# Bending Stress in Beams

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## Bending Stress

When a beam is loaded with external loads, all the sections of the beam will experience bending moments and shear forces. The shear forces and bending moments at various sections of the beam can be evaluated as discussed in the earlier chapter. In this chapter, the bending and bending stress distribution across a section will be dealt with.

Some practical applications of bending stress shall also be dealt with. These are

1. Moment carrying capacity of a section
2. Evaluation of extreme normal stresses due to bending
3. Design of beam for bending
4. Evaluation of load bearing capacity of the beam

The major stresses induced due to bending are normal stresses of tension and compression. But the state of stress within the beam includes *shear stresses due to the shear force* in addition to the major *normal stresses due to bending* although the former are generally of *smaller order* when compared to the latter.

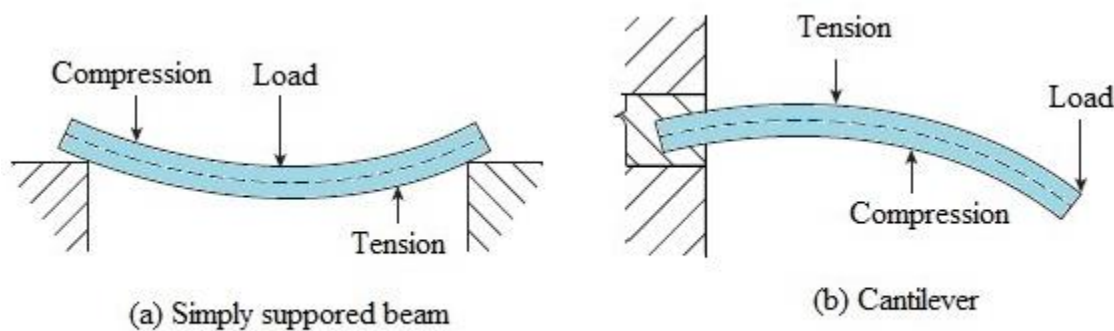


Fig. 1

## Simple Bending or Pure Bending

A beam or a part of it is said to be in a state of pure bending when it bends under the action of uniform/constant bending moment, without any shear force.

Alternatively, a portion of a beam is said to be in a state of simple bending or pure bending when the shear force over that portion is zero. In that case there is no chance of shear stress in the beam. But, the stress that will propagate in the beam as a result will be known as normal stress.

# Bending Stress in Beams

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However, in practice, when a beam is subjected to transverse loads, the bending moment at a section is accompanied by shear force. But, it is generally observed that the shear force is zero where the bending moment is maximum. Therefore, the condition of pure bending or simple bending is deemed to be satisfied at that section.

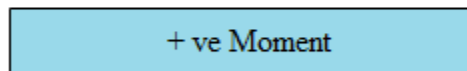
Examples of pure bending are –

1. Bending of simple supported beam due to end coupling (Uniform pure bending)
2. Bending of cantilever beam with end moment (Uniform pure bending)
3. Bending of the portion between two equal point loads in a simple supported beam with two-point loading (Non-uniform pure bending)

The *four point bending* of the simply supported beam

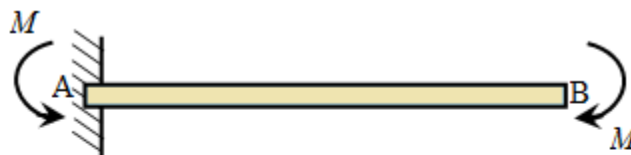


(a) Simple supported beam with end coupling

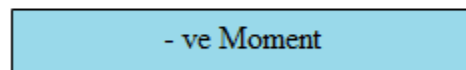


(b) Bending moment diagram

Fig. 2



(a) Cantilever subjected to moment at its end



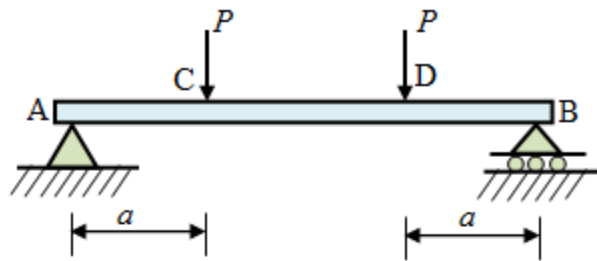
(b) Bending moment diagram

Fig. 3

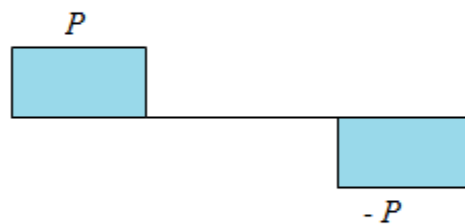


## Bending Stress in Beams

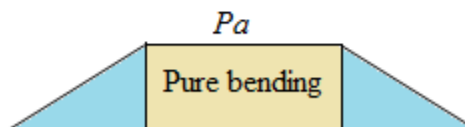
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(a) Simply supported beam with two-point loading



(b) Shear force diagram



(c) Bending moment diagram

Fig. 4

### Theory of Simple Bending

The theory which deals with the determination of stresses at a section of a beam due to pure bending is called *theory of simple bending*. In this chapter, bending of *straight homogeneous* beams of *uniform cross sectional area* with *vertical axis of symmetry* shall be considered. The application of this theory can be extended to beams with two or more different materials as well as curved beams.

Several cross-sections of beams satisfying the above conditions are shown in the Fig. 5.

A beam of rectangular cross-section with typical loading condition is shown in the Fig. 6. Also shown in the Fig. 7 is the three-dimensional beam with longitudinal vertical plane of symmetry, with the cross-section symmetric about this plane. It is assumed that the loading and supports are

## Bending Stress in Beams

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also symmetric about this plane. With these conditions, the beam has no tendency to twist and will undergo bending only.

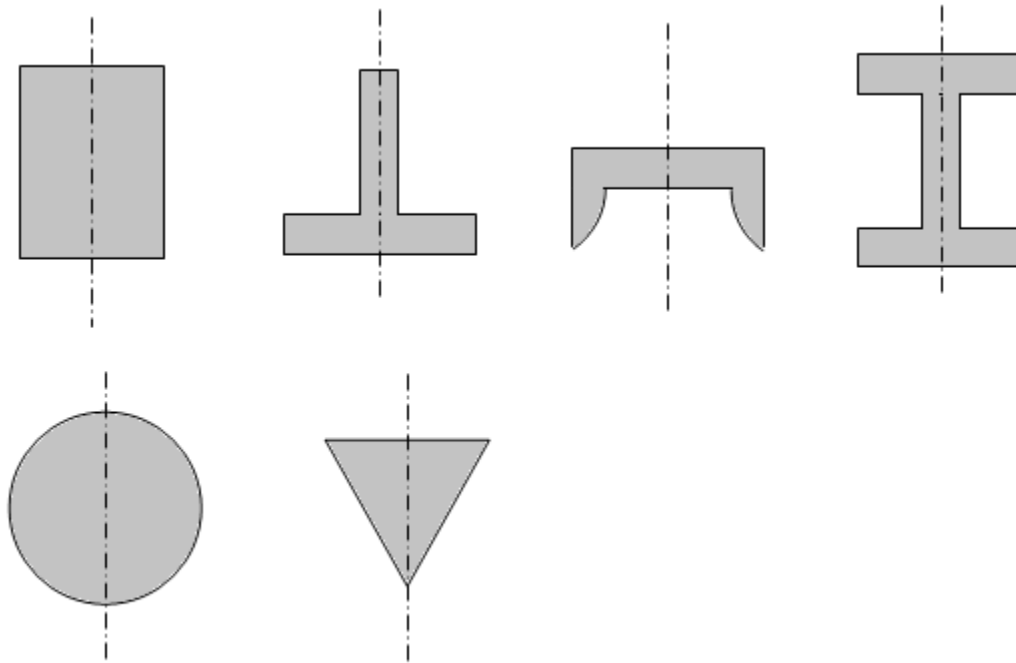


Fig.5 Beam cross-sections with vertical axis of symmetry

A beam subjected to sagging moment is shown in the Fig. 8. The beam is imagined to be consisting of a number of longitudinal fibres; one such fibre is shown in colour. It is obvious that the fibres near the upper side of the beam are compressed; hence an element in the upper part is under compression. The fibres at the bottom side of the beam get stretched and, hence, the elements on the lower side are subjected to tension. Somewhere in between, there will be a plane where the fibres are subjected to neither tension nor compression. Such a plane is termed as *neutral surface or neutral plane*.

In the conventional coordinate system attached to the beam in Fig. 8,  $x$  axis is the longitudinal axis of the beam, the  $y$  axis is in the transverse direction and the longitudinal plane of symmetry is in the  $x$ -  $y$  plane, also called the *plane of bending*.

### Neutral Surface

The longitudinal surface of a beam under bending which experiences neither tension nor compression is known as neutral surface. There is only one neutral surface in a beam.

# Bending Stress in Beams

## Neutral Axis

The line of intersection of transverse section of beam with the neutral surface is known as neutral axis. In other words, the line of intersection of the longitudinal plane of symmetry and the neutral surface is known as neutral axis. Neutral axis experiences no extension or contraction.

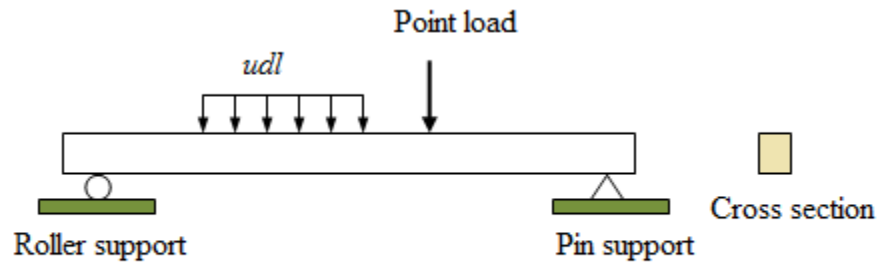


Fig.6 Loaded beam

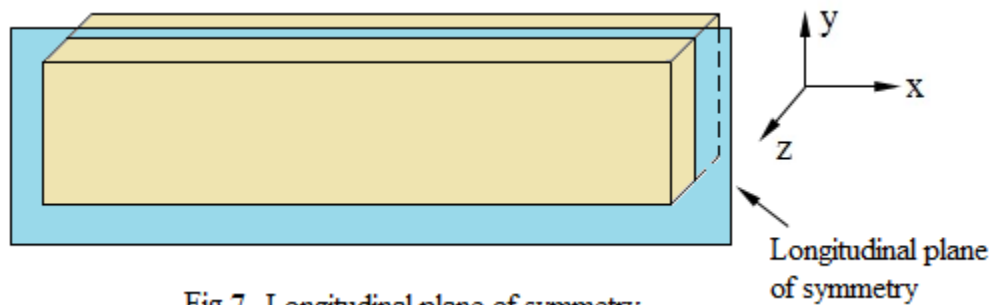


Fig.7 Longitudinal plane of symmetry

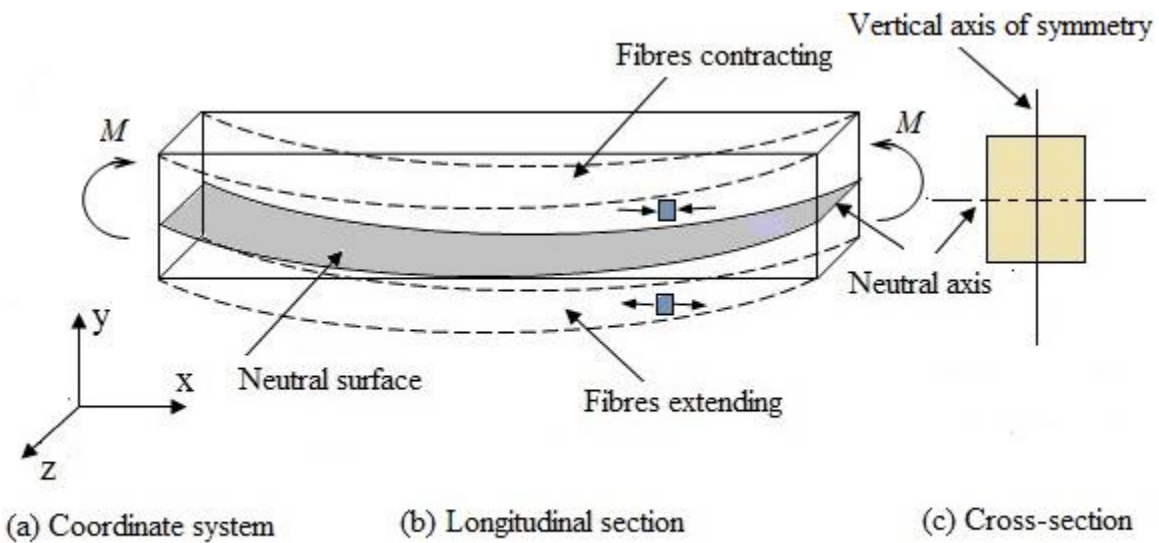


Fig.8 Pure moment (sagging)

# Bending Stress in Beams

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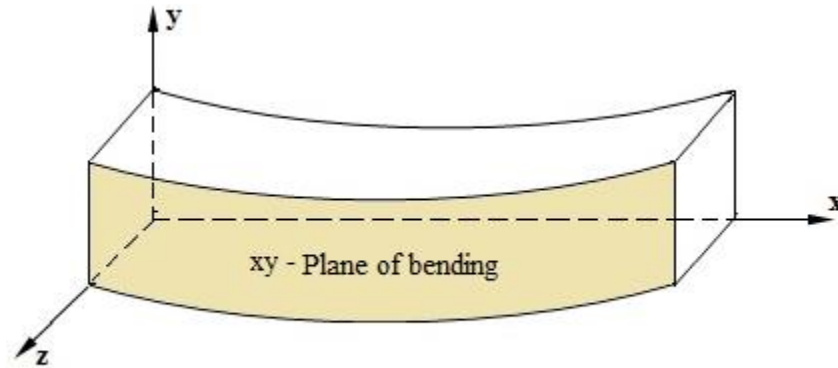


Fig.9 Plane of bending

## Axis of beam

The intersection of the longitudinal plane of symmetry and the neutral surface is called the **axis of the beam**. In other words, the line through the centroid of all the cross-sections of the beam is known as axis of the beam.

## Assumptions for theory of pure bending:

The assumptions made in the theory of simple bending are as follows:

1. The material of the beam is perfectly homogeneous (i.e. of the same kind throughout) and isotropic (i.e. of same elastic properties in all directions).
2. The material is stressed within elastic limit and obeys Hooke's law.
3. The value of modulus of elasticity for the material is same in tension and compression.
4. The beam is subjected to pure bending and therefore bends in the form of an arc of a circle.
5. The transverse sections, which are plane and normal to the longitudinal axis before bending, remain plane and normal to the longitudinal axis of the beam after bending.
6. The radius of curvature of the bent axis of the beam is large compared to the dimensions of the section of beam.
7. Each layer of the beam is free to expand or contract independently.
8. The cross-sectional area is symmetric about an axis perpendicular to the neutral axis.

## Explanation of the assumptions

## Bending Stress in Beams

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According to assumption No. 5, plane section  $ABCD$  before bending as shown in Fig. 10 remains plane after bending as shown by  $A'B'C'D'$ . This assumption, also known as Bernoulli's assumption, is perfectly valid for beams with pure bending. If there is any shear along with the bending, the shear deformation distorts the plane and  $A'B'$  will not remain plane. However, for beams with smaller depth ( $d < 1/10$ th span) shear deformation is small and this assumption is not much affected. In case of deep beams, with shear forces, this assumption fails.

Assumption No. 6, the radius of curvature is large compared to depth is valid if deflections are less than  $1/10$ th to  $1/5$ th of depth of beam. Therefore, the theory derived with this assumption may be called *small deflection theory*.

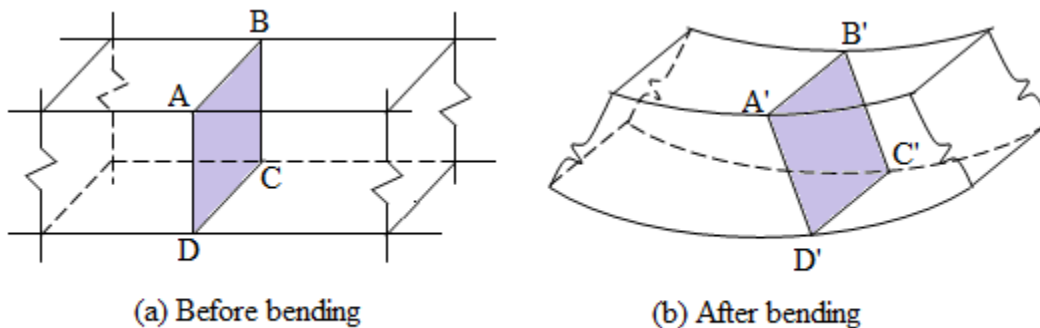


Fig. 10

### Relationship between Bending Stress and Radius of Curvature

Consider a part of beam  $ABCD$  of length  $dx$  subjected to pure bending of bending moment  $M$  as shown in the Fig. 11. As the beam is subjected to pure bending, it bends into a circular arc.

The topmost layer  $AB$  is contracted to  $A'B'$ . The layer  $PQ$  below it is compressed to a lesser degree than it. The bottom most layer  $CD$  is elongated to  $C'D'$ . All other layers are subjected to different degrees of elongation or contraction degrees depending upon their position. However, one layer  $MN$  has not suffered any change in its length. This layer is called the *neutral layer* or *neutral surface*.

Let  $d\theta$  be the angle formed by the planes  $A'C'$  and  $B'D'$  and  $R$  be the radius of the neutral layer. Consider a fibre  $PQ$  at a distance of  $y$  from the neutral layer.

Original length of the fibre,  $PQ = dx = Rd\theta$

After deformation, the length of the fibre is compressed to  $P'Q'$ .

## Bending Stress in Beams

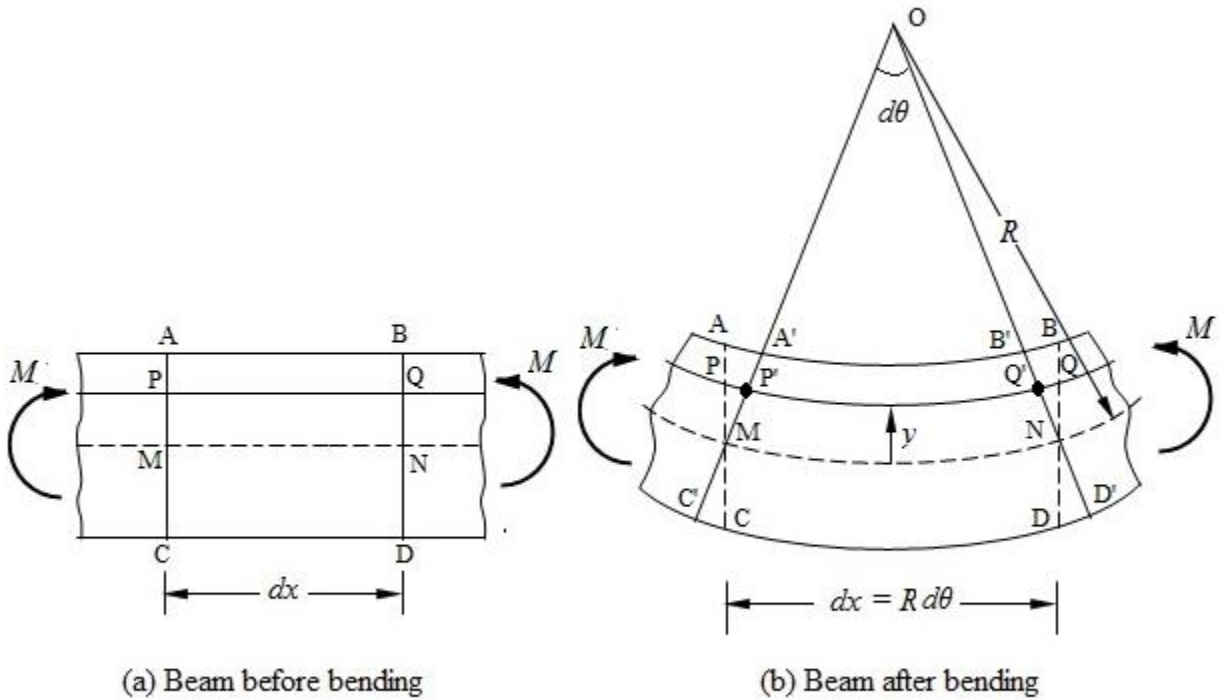


Fig. 11

$$\begin{aligned}
 \text{Decrease in length of the fibre } PQ &= PQ - P'Q' \\
 &= R d\theta - (R - y) d\theta \\
 &= y d\theta
 \end{aligned}$$

Let the projection of  $C'A'$  and  $D'B'$  meet at  $O$ .

$$\text{Strain in the fibre } PQ, \quad \epsilon = \frac{\text{Decrease in length}}{\text{Original length}}$$

$$\therefore \epsilon = \frac{y d\theta}{R d\theta} = \frac{y}{R}$$

Let  $\sigma$  be the stress in the fibre  $PQ$ .

Then,  $\epsilon = \frac{\sigma}{E}$ , where  $E$  is the Modulus of elasticity of the material.

$$\therefore \epsilon = \frac{\sigma}{E} = \frac{y}{R}$$

$$\therefore \sigma = \frac{E}{R} \times y$$

## Bending Stress in Beams

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Hence, the stress intensity in any fibre is proportional to the distance of the fibre from the neutral layer.

### Position of Neutral Axis

Consider a beam of arbitrary cross-section as shown in the Fig. 12. Consider an elemental area  $\delta a$  at a distance  $y$  from the neutral axis. Let the bending stress on the element be  $\sigma$ .

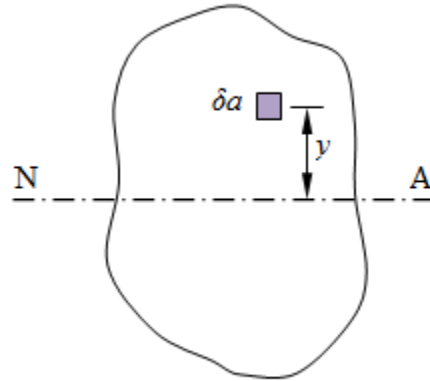


Fig. 12

Force on the elemental area =  $\sigma \delta a$

Force over the entire cross-section of the beam =  $\sum \sigma \delta a$

We also know,  $\sigma = \frac{E}{R} \times y$

Substituting the value of  $\sigma$ , we get

Force over the entire cross-section of the beam =  $\sum \frac{E}{R} y \delta a = \frac{E}{R} \sum y \delta a$

Since there is no axial force on the beam, from equilibrium consideration, the above axial force should be zero.

Hence, 
$$\frac{E}{R} \sum y \delta a = 0$$

Since,  $\frac{E}{R}$  is constant for a given section, we have  $\sum y \delta a = 0$

We know, 
$$A \bar{y} = \sum y \delta a$$

Where,  $A$  is the area of cross-section of the beam.

So, 
$$A \bar{y} = 0$$

## Bending Stress in Beams

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$$\text{or } \bar{y} = 0$$

$\bar{y}$  is the distance of the centroid from the neutral axis. Hence, the neutral axis of the section coincides with the centroid of the section. Thus, to locate the neutral axis of a section, the centroid of the section should be determined. The line passing through the centroid, parallel to the plane of bending is the neutral axis of the beam section.

### Relationship between Moment and Radius of Curvature

Consider an elemental area  $\delta a$  from the neutral axis of a beam section as shown in the Fig. 13.

The stress on the elemental area,  $\sigma = \frac{E}{R} y$

Force on the elemental are  $\sigma \delta a = \frac{E}{R} y \delta a$

Moment of resistance offered by this elemental area about the neutral axis

$$= \left( \frac{E}{R} y \delta a \right) y = \frac{E}{R} y^2 \delta a$$

Total moment of resistance,  $M$  offered by the cross-sectional area of beam,

$$M = \sum \frac{E}{R} y^2 \delta a$$

$$M = \frac{E}{R} \sum y^2 \delta a$$

But,  $\sum y^2 \delta a$  is the moment of inertia  $I$  of the beam section about the neutral axis.

$$\therefore M = \frac{E}{R} I$$

$$\frac{M}{I} = \frac{E}{R}$$

We have earlier seen that,  $\frac{\sigma}{y} = \frac{E}{R}$

Combining the two equations, we get

$$\frac{M}{I} = \frac{\sigma}{y} = \frac{E}{R}, \text{ which is known as the } \textit{bending equation}.$$



# Bending Stress in Beams

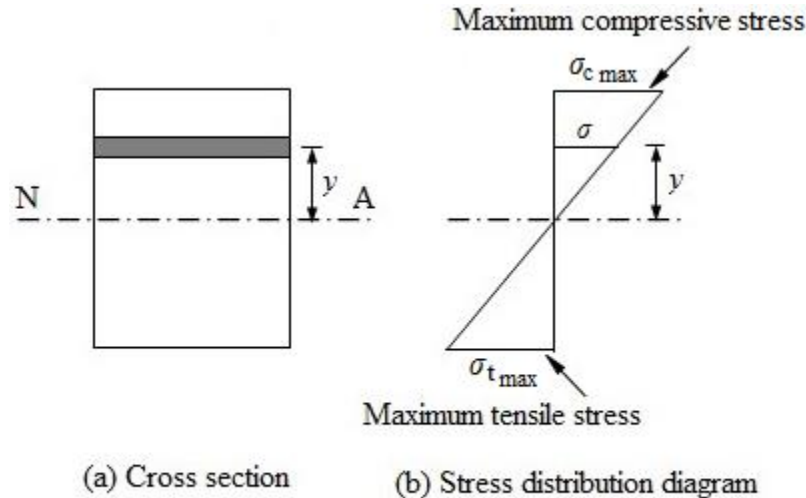


Fig. 13

Where,

$M$  = bending moment at a section,

$I$  = moment of inertia of the beam section,

$\sigma$  = stress at any layer of the beam,

$y$  = distance of the layer from the neutral axis,

$E$  = Young's Modulus and

$R$  = radius of curvature.

$M$  and  $I$  are constants for a particular beam section. Hence,  $\sigma$  varies proportionally to the distance  $y$ . So, maximum stress occurs at extreme fibres. The stress distribution will be triangular as shown in the Fig. 13.

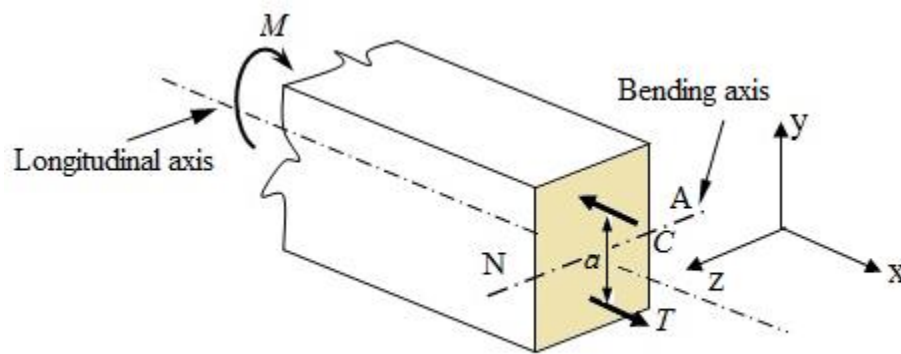


Fig. 14

The formula for flexural stress derived as above applies only to cases where the material behaves elastically. The important concepts used in deriving the flexural formula may be summed up as follows.

## Bending Stress in Beams

---

1. Strains in different layers of beam vary linearly with their distances from the neutral axis.
2. Properties of materials are used to relate strain and stress.
3. Equilibrium conditions are used to locate the neutral axis and to determine the internal stresses.

The internal bending moment developed by the induced flexural stresses due to bending at a section is known as ***moment of resistance*** of the section. For equilibrium of the section, the moment of resistance of a section should be equal to or greater than the applied external moment.

### Flexural rigidity:

From equation of flexure, we have

$$\frac{M}{I} = \frac{E}{R}$$

$$EI = MR$$

$EI$  is known as flexural rigidity. ***Flexural rigidity*** is the measure of flexural strength of a beam section. Higher is the flexural rigidity better is the flexural strength. It depends upon the material as well as the geometric property of the section. Elastic modulus,  $E$  reflects the material character and moment of inertia,  $I$  reflects the geometric characteristic

### Economical section

In a beam of rectangular or circular section, the fibres near neutral axis are under-stressed compared to those at the top and bottom. As a result, a large portion of the beam cross-section remains under-stressed and under utilized for resisting flexure or bending.

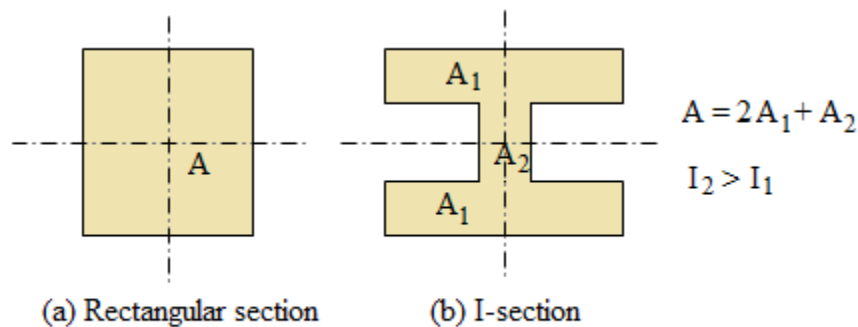


Fig. 15

## Bending Stress in Beams

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The expression  $M = \frac{\sigma}{y} I$  indicates that moment of resistance of a section can be greatly increased by increasing the moment of inertia by rearranging or redistributing the area while keeping the cross-sectional area and the depth of the beam unchanged. This can be achieved by changing the geometry of the section so as to spread the area farther from the neutral axis.

In order to increase the moment of resistance to bending of a beam section, it is advisable to use sections which have large area away from the neutral axis. Hence, I-section and T-sections are preferable to rectangular section.

Sections of different geometry, (i) rectangular section and (ii) I-section of equal cross-sectional area and same depth are shown in the Fig. 15.

### **Moment carrying capacity of a section:**

From equation of flexure, we have

$$\frac{\sigma}{y} = \frac{M}{I}$$

$$\sigma = \frac{M}{I} y$$

It is obvious that bending stress is maximum on the extreme fibre at the top and bottom of the beam where  $y$  is maximum. In design of beam, the extreme fibre stress should not be allowed to exceed the allowable or permissible stress of the material. If  $\sigma_{allow}$  is the allowable stress for bending, then for safe design

$$\sigma_{max} \leq \sigma_{allow}$$

$$\frac{M}{I} y_{max} \leq \sigma_{allow}$$

If  $M$  is taken as the maximum moment carrying capacity of the section,

$$\frac{M}{I} y_{max} \leq \sigma_{allow}$$

# Bending Stress in Beams

---

$$M \leq \frac{I}{y_{\max}} \sigma_{allow}$$

The moment of inertia  $I$  and the extreme fibre distance  $y_{\max}$  are the geometrical properties of the section. The ratio of the moment of inertia and the extreme fibre distance ( $I/y_{\max}$ ) for a given cross-section of beam is constant and is known as **section modulus** ( $Z$ ). Thus the moment carrying capacity of a beam is given by

$$M = \sigma_{allow} Z$$

If  $\sigma_{allow}$  in tension and compression are same, doubly symmetric section is selected. Doubly symmetric section means a section which is symmetric about the vertical as well as neutral axis. If  $\sigma_{allow}$  in tension and compression are different, un-symmetric cross-section is selected such that the distance to the extreme fibers are nearly the same ratio as the respective allowable stresses. In the latter case, the moment carrying capacity in tension and compression are found separately and the smaller one is taken as the moment carrying capacity of the section.

## Section Modulus of Sections of Standard Geometry

### 1. Rectangular section

Let us consider a rectangular section of width  $b$  and depth  $d$  as shown in the Fig. The neutral axis coincides with the centroidal axis of the beam.

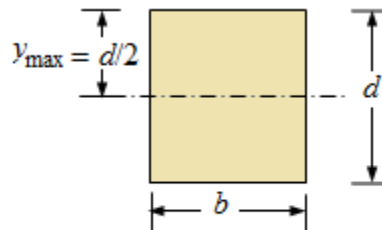


Fig. 16

Moment of inertia about the neutral axis,  $I = \frac{bd^3}{12}$

Distance of outermost fibre from the neutral axis,  $y_{\max} = \frac{d}{2}$

Section modulus,  $Z = \frac{I}{y_{\max}} = \frac{bd^3}{12} \times \frac{2}{d}$

## Bending Stress in Beams

---

$$= \frac{1}{6}bd^2$$

Let  $\sigma$  is the maximum bending stress developed at the outermost layer.

$$\text{Moment of resistance, } M = \sigma Z = \frac{1}{6}\sigma bd^2$$

### 2. Hollow Rectangular section

Let us consider a hollow rectangular section of size  $B \times D$  with a symmetrical opening  $b \times d$  as shown in the Fig. 17.

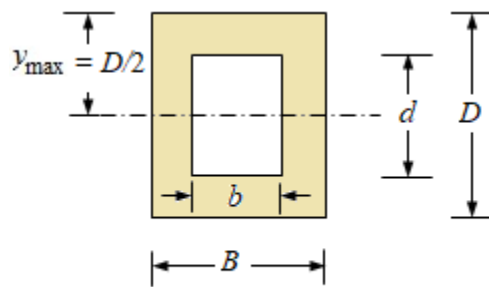


Fig. 17

$$\text{Moment of inertia about the neutral axis, } I = \frac{BD^3}{12} - \frac{bd^3}{12}$$

$$\text{Distance of outermost fibre from the neutral axis, } y_{\max} = \frac{D}{2}$$

$$\text{Section modulus, } Z = \frac{I}{y_{\max}} = \frac{BD^3 - bd^3}{12} \times \frac{2}{D}$$

$$= \frac{1}{6} \frac{(BD^3 - bd^3)}{D}$$

Let  $\sigma$  is the maximum bending stress developed at the outermost layer.

$$\text{Moment of resistance, } M = \sigma Z = \frac{1}{6}\sigma \frac{(BD^3 - bd^3)}{D}$$

### 3. Circular section

Let us consider a circular section of diameter  $d$  as shown in the Fig. 18.

## Bending Stress in Beams

---



Fig. 18

Moment of inertia about the neutral axis,  $I = \frac{\pi d^4}{64}$

Distance of outermost fibre from the neutral axis,  $y_{\max} = \frac{d}{2}$

Section modulus,

$$Z = \frac{I}{y_{\max}} = \frac{\pi d^4}{64} \times \frac{2}{d}$$
$$= \frac{\pi d^3}{32}$$

Let  $\sigma$  is the maximum bending stress developed at the outermost layer.

Moment of resistance,  $M = \sigma Z = \sigma \frac{\pi d^3}{32}$

### 4. Hollow Circular section

Let us consider a hollow circular section of external and internal diameter  $D$  and  $d$  respectively as shown in the Fig. 19.

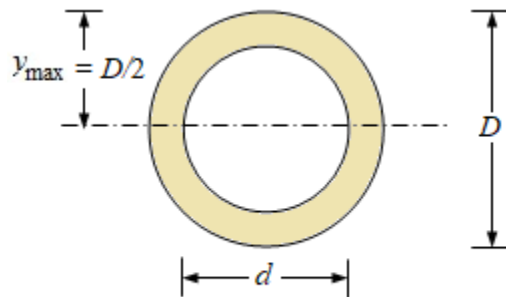


Fig. 19

Moment of inertia about the neutral axis,  $I = \frac{\pi}{64} (D^4 - d^4)$

## Bending Stress in Beams

---

Distance of outermost fibre from the neutral axis,  $y_{\max} = \frac{D}{2}$

Section modulus,

$$Z = \frac{I}{y_{\max}} = \frac{\pi}{64}(D^4 - d^4) \times \frac{2}{D}$$
$$= \frac{\pi}{32D}(D^4 - d^4)$$

Let  $\sigma$  is the maximum bending stress developed at the outermost layer.

Moment of resistance,  $M = \sigma Z = \frac{\sigma\pi}{32D}(D^4 - d^4)$

### Design of beam for bending

Design of beam involves the determination of the size (cross-section) of the beam for given loading condition. The maximum bending moment of the beam is determined from the loading condition. Given the bending moment and permissible bending stress of the material of the beam, the section modulus of the beam is determined from the expression of bending stress. Once the section modulus is known, width and depth can easily determined assuming the depth to width ratio.

### Beam of uniform strength

In practice, a beam of uniform cross section is designed for moment of resistance same as the maximum bending moment that the beam is supposed to carry. Hence, the material in all sections except the section of maximum bending moment remains under-stressed and underutilized. Although practical, such a beam is uneconomical. Ideally, a beam of varying cross-section should be designed so that all sections attain the maximum permissible stress simultaneously. A beam in which permissible stress at all sections is reached simultaneously under a given loading, is called a *beam of uniform strength*.

A beam of uniform strength can be obtained in different ways

- a) By varying the width of beam and keeping the depth constant
- b) By varying the depth of beam and keeping the width constant
- c) By varying both width and depth

### By varying the width of beam and keeping the depth constant

## Bending Stress in Beams

---

Derive the formula for cross section of a rectangular beam of uniform strength for a cantilever beam of length  $L$  carrying concentrated load at free end by keeping the depth constant.

Consider a cantilever beam of length  $L$  and uniform depth  $d$  carrying a concentrated load  $W$  at its free end as shown in the Fig. 20. Let the width varies from a minimum at its free end to a maximum of  $b$  near the fixed end.

It is obvious that the bending moment varies from minimum zero at the free end to maximum at  $WL$  at the fixed support.

Bending moment at any section at a distance of  $x$  from the free end,

$$M = Wx$$

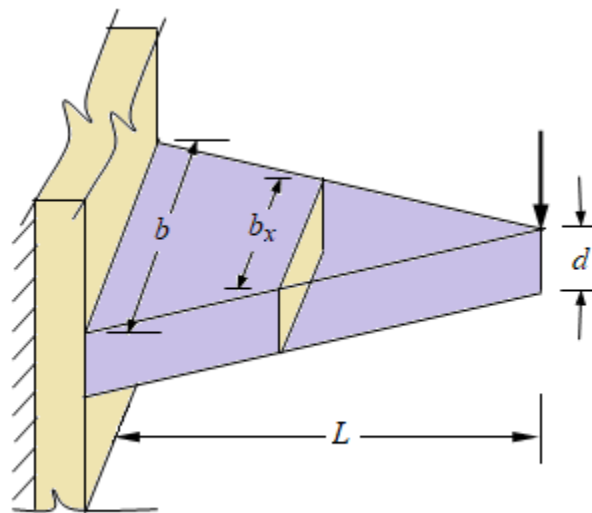


Fig. 20

From expression of flexure, we have

$$M = \sigma Z$$

$$Wx = \sigma Z$$

Where  $\sigma$  is the maximum stress at every section of the beam.

If  $b_x$  width at any section XX, then  $Z = \frac{b_x d^2}{6}$



## Bending Stress in Beams

---

$$\therefore \sigma = \frac{Wx}{\frac{b_x d^2}{6}} = \frac{6Wx}{b_x d^2}$$

Similarly, maximum stress at support,  $\sigma = \frac{6WL}{bd^2}$

Equating equation () and (), we have

$$\frac{6Wx}{b_x d^2} = \frac{6WL}{bd^2}$$

$$b_x = b \left( \frac{x}{L} \right)$$

At free end, i.e.,  $x = 0$ , the width of beam  $b_0 = 0$

At the fixed end, i.e.,  $x = L$ , the width  $b_L = b \left( \frac{L}{L} \right) = b$

### By varying the depth of beam and keeping the width constant

Consider a cantilever beam of length  $L$  and uniform width  $b$  carrying a concentrated load  $W$  at its free end as shown in the Fig. 20. Let the depth varies from a minimum at its free end to a maximum of  $d$  near the fixed end.

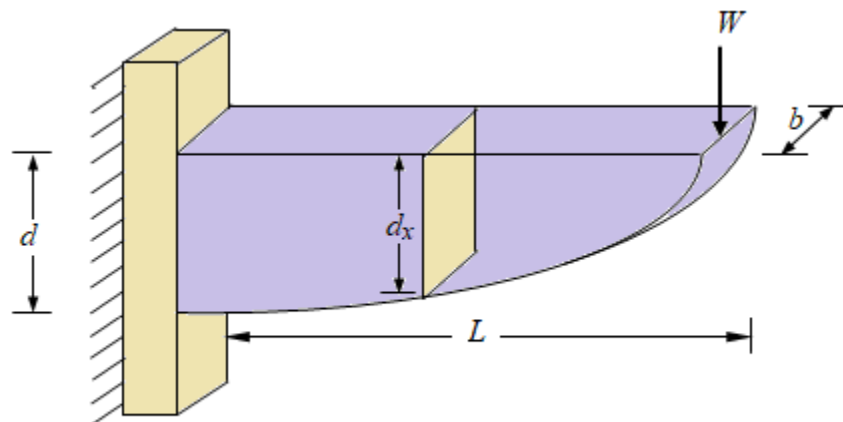


Fig. 21

## Bending Stress in Beams

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It is obvious that the bending moment varies from minimum zero at the free end to maximum at  $WL$  at the fixed support.

Bending moment at any section at a distance of  $x$  from the free end,

$$M = Wx$$

From expression of flexure, we have

$$M = \sigma Z$$

$$Wx = \sigma Z$$

Where  $\sigma$  is the maximum stress at every section of the beam.

If  $b_x$  width at any section XX, then  $Z = \frac{bd_x^2}{6}$

$$\therefore \sigma = \frac{Wx}{\frac{bd_x^2}{6}} = \frac{6Wx}{bd_x^2}$$

Similarly, maximum stress at support,  $\sigma = \frac{6WL}{bd^2}$

Equating equation () and (), we have

$$\frac{6Wx}{bd_x^2} = \frac{6WL}{bd^2}$$

$$d_x = d \sqrt{\left(\frac{x}{L}\right)}$$

At free end, i.e.,  $x = 0$ , the depth of beam,  $d_0 = 0$

At the fixed end, i.e.,  $x = L$ , the depth,  $d_L = d \sqrt{\left(\frac{L}{L}\right)} = d$

# Bending Stress in Beams

## Numerical

1. A rectangular beam of breadth  $100\text{ mm}$  and depth  $200\text{ mm}$  is simply supported over a span of  $4\text{ m}$ . The beam is loaded with an uniformly distributed load of  $5\text{ kN/m}$  over the entire span. Find the maximum bending stresses.

### Solution:

Breadth of the beam,  $b = 100\text{ mm}$

Depth of beam,  $d = 200\text{ mm}$

Moment of inertia,  $I = \frac{1}{12}bd^3 = \frac{1}{12} \times 100 \times (200)^3 = 66.67 \times 10^6\text{ mm}^4$

Span of beam,  $l = 4\text{ m}$

Uniformly distributed load,  $w = 5\text{ kN/m}$

Maximum bending moment at centre of beam,  $M = \frac{wl^2}{8} = \frac{5 \times 4^2}{8}$

$$= 10\text{ kN.M} = 10^7\text{ N.mm}$$

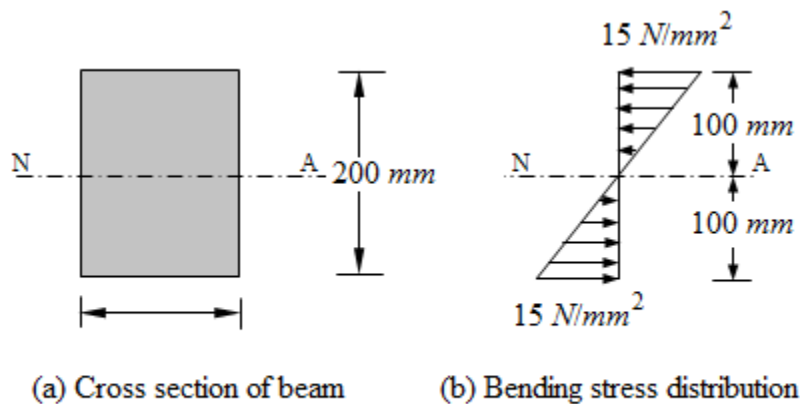


Fig. 22

Neutral axis passes through the centroid of section.

The distance of top and bottom fibre from the neutral axis,  $y = 100\text{ mm}$

Thus, maximum bending stress,  $\sigma = \frac{M}{I} y = \frac{10^7}{66.67 \times 10^6} \times 100$

## Bending Stress in Beams

---

$$= 15 \text{ N/mm}^2$$

2. A beam of I-section shown in Fig. 23 is simply supported over a span of 10 m. It carries a uniform load of 4 kN/m over the entire span. Evaluate the maximum bending stresses.

**Solution:**

$$\text{Moment of inertia, } I = \frac{1}{12}(BD^3 - bd^3) = \frac{1}{12}(300 \times 660^3 - 280 \times 600^3)$$

$$= 21.474 \times 10^8 \text{ mm}^4$$

Span of the beam,  $l = 10 \text{ m}$

Uniformly distributed load,  $w = 4 \text{ kN/m}$

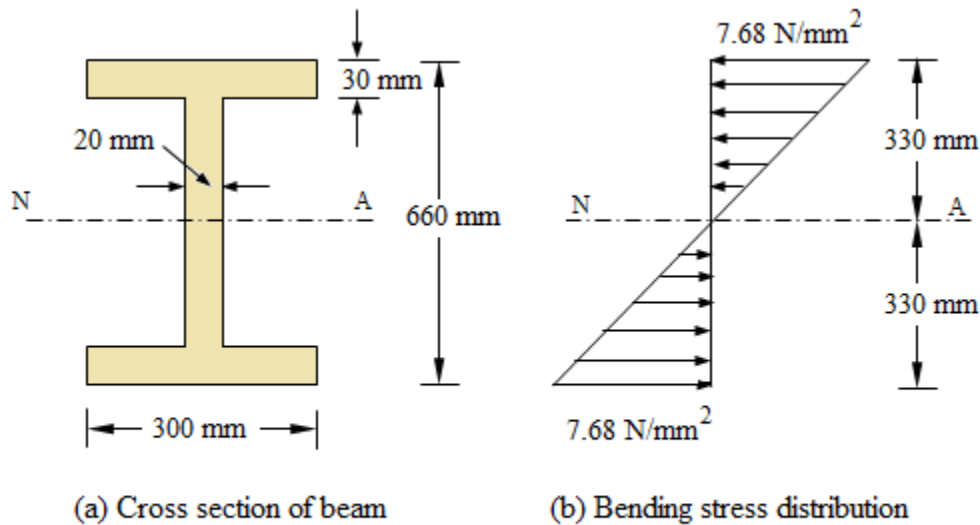


Fig. 23

$$\text{Maximum bending moment at centre of beam, } M = \frac{4 \times 10^2}{8} = 50 \text{ kN.m}$$

$$= 5 \times 10^7 \text{ N.mm}$$

Neutral axis passes through the centroid of I-section.

The distance of top and bottom fibre from the neutral axis,  $y = 330 \text{ mm}$

## Bending Stress in Beams

---

Thus, maximum bending stress,  $\sigma = \frac{M}{I} y = \frac{5 \times 10^7}{21.474 \times 10^8} \times 330 = 7.68 \text{ N/mm}^2$

The bending stress at top and bottom fibres =  $7.68 \times 10^8 \text{ N/mm}^2$

3. A beam of an I-section shown in Fig. 24 is simply supported over a span of 4 m. Find the uniformly distributed load the beam can carry if the bending stress is not to exceed 100 N/mm<sup>2</sup>.

**Solution:**

$$\begin{aligned} \text{Moment of inertia, } I &= \frac{1}{12} (BD^3 - bd^3) = \frac{1}{12} (200 \times 300^3 - 180 \times 260^3) \\ &= 180.36 \times 10^6 \text{ mm}^4 \end{aligned}$$

Maximum bending stress,  $\sigma_{\max} = 100 \text{ N/mm}^2$

Span of beam,  $l = 4 \text{ m}$

Extreme fibre distance,  $y_{\max} = 150 \text{ mm}$

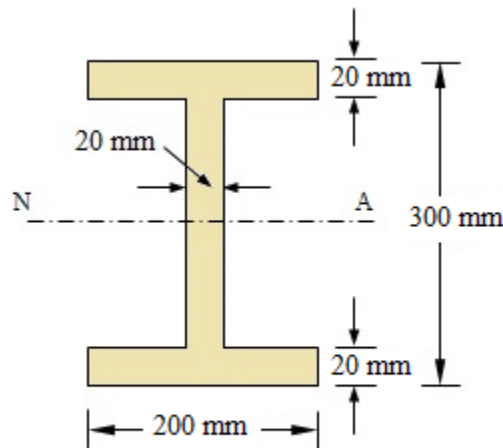


Fig. 24

$$\text{Section modulus, } Z = \frac{I}{y_{\max}} = \frac{180.36 \times 10^6}{150} = 1242400 \text{ mm}^3$$

Maximum bending moment,  $M = \sigma_{\max} Z = 100 \times 1242400$

$$= 124240000 \text{ N.mm}$$

## Bending Stress in Beams

---

$$= 124.24 \text{ kN.m}$$

But 
$$M = \frac{wl^2}{8}$$

$$124.24 = \frac{w \times (4)^2}{8}$$

$$w = \frac{124.24 \times 8}{16} = 64.12 \text{ kN/m}$$

The maximum uniformly distributed load the beam can carry = 64.12 kN/m.

4. A timber beam of rectangular section carries a load of 2 kN at mid-span. The beam is simply supported over a span of 3.6 m. If the depth of section is to be twice the breadth, and the bending stress is not to exceed 9 N/mm<sup>2</sup>, determine the cross-sectional dimensions.

### Solution:

Span of the beam,  $l = 3.6 \text{ m}$

Uniformly distributed load,  $w = 2 \text{ kN}$

Allowable bending stress,  $\sigma_{\text{allow}} = 9 \text{ N/mm}^2$

Maximum bending moment at centre of beam,  $M = \frac{WL}{4} = \frac{2 \times 3.6}{4} = 1.8 \text{ kN.m}$

$$= 1.8 \times 10^6 \text{ N.mm}$$

From the flexural relationship, we have 
$$Z = \frac{M}{\sigma_{\text{allow}}}$$

$$\frac{1}{6}bd^2 = \frac{1.8 \times 10^6}{9}$$

$$bd^2 = \frac{1.8 \times 10^6}{9} \times 6 = 1.2 \times 10^6$$

Depth of section is to be twice the breadth, i.e.,  $d = 2b$

So, we have 
$$b(2b)^2 = 1.2 \times 10^6$$

## Bending Stress in Beams

---

$$b^3 = \frac{1.2 \times 10^6}{4} = 0.3 \times 10^6$$

$$b = 64.94 \text{ mm}$$

$$d = 2 \times 64.943 = 129.886 \text{ mm}$$

Therefore, width of beam = 65 mm, and depth of beam = 130 mm

5. A rectangular beam of width 200 mm and depth 300 mm is simply supported over a span of 5 m. Find the safe uniformly distributed load that the beam can carry per metre length if the allowable bending stress in the beam is 100 N/mm<sup>2</sup>.

### Solution:

Span of beam,  $l = 5 \text{ m}$

Width Breadth of the beam,  $b = 100 \text{ mm}$

Depth of beam,  $d = 200 \text{ mm}$

Allowable bending stress,  $\sigma_{\text{allow}} = 100 \text{ N/mm}^2$

Section modulus,  $Z = \frac{1}{6}bd^2 = \frac{1}{2} \times 200 \times 300^2 = 3 \times 10^6 \text{ mm}^3$

Moment of resistance of the beam,  $M = \sigma_{\text{allow}}Z = 100 \times 3 \times 10^6$   
 $= 300 \times 10^6 \text{ N.mm} = 300 \text{ kN.m}$

Maximum bending moment at the centre of the beam,

$$M = \frac{wl^2}{8}$$

$$300 = \frac{w \times (5)^2}{8}$$

$$\therefore w = \frac{300 \times 8}{25} = 96 \text{ kN.m}$$

So, the load that the beam can carry is 96 kN/m.

## Bending Stress in Beams

---

6. A rectangular beam of size  $60\text{ mm} \times 100\text{ mm}$  has a central rectangular hole of size  $15\text{ mm} \times 20\text{ mm}$ . The beam is subjected to bending and the maximum bending stress is limited to  $100\text{ N/mm}^2$ . Find the moment of resistance of the hollow beam section.

### Solution:

External dimension of hollow rectangular beam:  $B = 60\text{ mm}$ ,  $D = 100\text{ mm}$

Size of the central hole:  $b = 15\text{ mm}$ ,  $d = 20\text{ mm}$

$$\begin{aligned}\text{Moment of inertia of the hollow beam section, } I &= \frac{1}{12}(BD^3 - bd^3) = \frac{1}{12}(60 \times 100^3 - 15 \times 20^3) \\ &= 4.999 \times 10^6\text{ mm}^4\end{aligned}$$

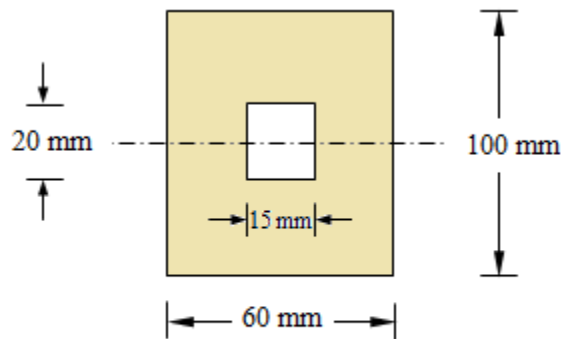


Fig.25

$$\text{Extreme fibre distance, } y_{\max} = \frac{100}{2} = 50\text{ mm}$$

$$\text{Section modulus, } Z = \frac{I}{y_{\max}} = \frac{4.999 \times 10^6}{50} = 9.98 \times 10^4\text{ mm}^3$$

$$\text{Allowable bending stress, } \sigma_{\text{allow}} = 100\text{ N/mm}^2$$

$$\text{Moment of resistance, } M = \sigma_{\text{allow}} Z = 100 \times 9.98 \times 10^4$$

$$= 9.98 \times 10^6\text{ N.mm}$$

$$= 9.98\text{ kN.mm}$$



## Bending Stress in Beams

---

7. Find the ratio of the dimensions of the strongest rectangular beam that can be cut from a circular log of wood of diameter  $D$ .

**Solution:**

Let  $b$  be the width and  $d$  the depth of the strongest rectangular beam section as shown in the Fig. 26.

From the geometry, we have  $b^2 + d^2 = D^2$

$$d^2 = D^2 - b^2$$

Section modulus of the rectangular section,

$$\begin{aligned} Z &= \frac{1}{6}bd^2 = \frac{1}{6}b(D^2 - b^2) \\ &= \frac{1}{6}(bD^2 - b^3) \end{aligned}$$

Strongest section in bending should have largest section modulus.

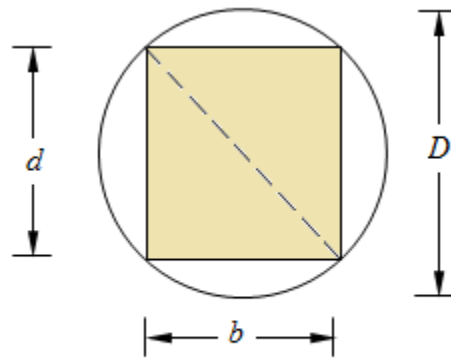


Fig. 26

Hence,

$$\frac{dZ}{db} = \frac{1}{6}(D^2 - 3b^2) = 0$$

$$3b^2 = D^2$$

$$b = \frac{D}{\sqrt{3}}$$

## Bending Stress in Beams

---

And

$$d = \sqrt{D^2 - b^2} = \sqrt{D^2 - \frac{D^2}{3}}$$
$$= \sqrt{\frac{2D^2}{3}} = \left(\sqrt{\frac{2}{3}}\right)D$$

8. Two sections of same material; one of solid circular section and the other hollow circular section of internal diameter half the external diameter, have the same flexural strength. Which one of them is economical?

**Solution:**

Let  $D$  = Diameter of solid circular section

$D_1$  = Outer diameter of hollow circular section

Inside diameter of hollow circular section,  $D_2 = 0.5 D_1$

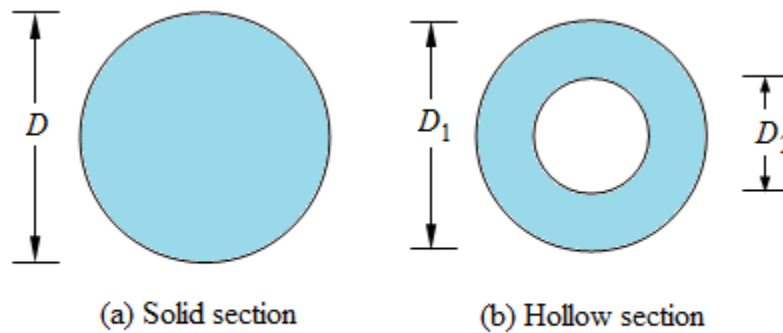


Fig. 27

Section modulus of solid section,  $Z_1 = \frac{\pi}{32} D^3$

Section modulus of hollow section,  $Z_2 = \frac{\pi}{32D_1} (D_1^4 - D_2^4) = \frac{\pi}{32D_1} \{D_1^4 - (0.5D_1)^4\}$

$$= \frac{\pi}{32} \times 0.9375 D_1^3$$

Since both sections have same flexural strength, their section modulus should be equal.

Hence,

$$\frac{\pi}{32} D^3 = \frac{\pi}{32} \times 0.9375 D_1^3$$

## Bending Stress in Beams

$$D^3 = 0.9375D_1^3$$

$$D = 0.98D_1$$

$$\begin{aligned} \frac{\text{Cross-sectional area of solid section}}{\text{Cross-sectional area of hollow section}} &= \frac{A_s}{A_h} = \frac{\frac{\pi}{4}D^2}{\frac{\pi}{4}(D_1^2 - D_2^2)} = \left\{ \frac{D^2}{D_1^2 - (0.5D_1)^2} \right\} \\ &= \frac{D^2}{0.75D_1^2} = \frac{1}{0.75} \times \left( \frac{D}{D_1} \right)^2 \\ &= \frac{1}{0.75} \times (0.98)^2 = 1.28 \end{aligned}$$

Since the sectional area of hollow section is less than that of solid section, for a given length of the beam, the weight of hollow section will be less. Hence hollow section is economical.

9. A cantilever of 2 m length and square section 200 mm x 200 mm, just fails in bending when a point load of 12 kN is placed at its free end. A beam of rectangular cross section of same material, 150 mm wide and 300 mm deep, is simply supported over a span of 3 m. Calculate the maximum concentrated load that the beam can carry at its centre without failure.

### Solution:

The two beams with loading conditions are shown in the Fig.

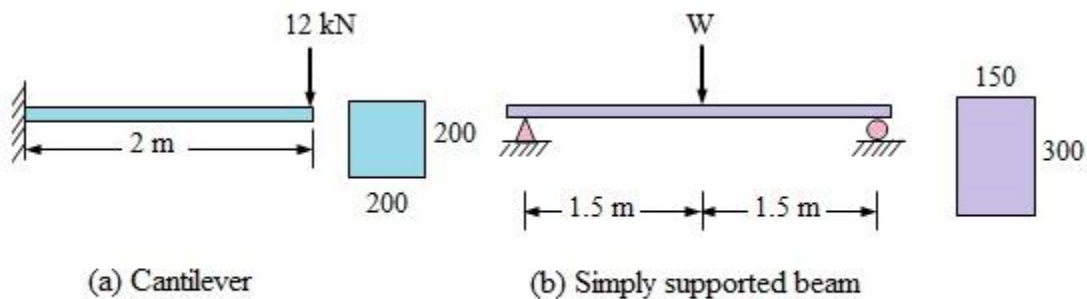


Fig. 28

$$\begin{aligned} \text{Maximum bending moment in cantilever beam, } M_c &= 12 \times 2 = 24 \text{ kN.m} \\ &= 24 \times 10^6 \text{ N.mm} \end{aligned}$$

## Bending Stress in Beams

---

Let  $\sigma_{allow}$  is the stress at which the beam fails,  $M_c = \sigma_{allow} Z = \frac{1}{6} b d^2 \sigma_{allow}$

$$\frac{1}{6} \times 200 \times 200^2 \times \sigma_{allow} = 24 \times 10^6$$

$$\sigma_{allow} = 18 \text{ N/mm}^2$$

Let  $W \text{ kN}$  be the maximum central concentrated that the beam can carry without failure.

Maximum bending moment at the mid span,  $M_s = \frac{WL}{4} = \frac{W \times 3}{4} = 0.75W \text{ kN.m}$

$$= 0.75 \times 10^6 W \text{ N.mm}$$

Moment of resistance of simply supported beam section,

$$M_R = \sigma_{allow} Z = 18 \times \frac{1}{6} \times 150 \times 300^2$$

$$= 40.5 \times 10^6 \text{ N.mm}$$

Equating maximum bending moment ( $M_s$ ) to moment of resistance ( $M_R$ ), we have

$$0.75 \times 10^6 W = 40.5 \times 10^6$$

$$W = 54 \text{ kN}$$

10. For a given sectional area, compare the moments of resistance of circular and square section.

### **Solution:**

Let the diameter of the circular section be  $d$ .

Area of circular section,  $A = \frac{\pi}{4} d^2$

Section modulus,  $Z_c = \frac{\pi}{32} d^3$

Let the square section has side of  $a$ .

Since both circular and square section have the same area,

$$a^2 = \frac{\pi}{4} d^2$$

$$a = \frac{\sqrt{\pi}}{2} d$$

## Bending Stress in Beams

Section modulus of square section,  $Z_s = \frac{a^3}{6} = \frac{\pi\sqrt{\pi}}{48} d^3$

Ratio of Section modulus of square section and circular section,

$$\frac{Z_s}{Z_c} = \frac{\frac{\pi\sqrt{\pi}}{48} d^3}{\frac{\pi}{32} d^3} = 1.18$$

Hence, flexural strength of square section is 1.18 times more than that of circular section of equal area.

11. Compare the moments of resistance of a square section of given material when the beam section is placed such that (i) two sides are parallel and (ii) one diagonal vertical.

**Solution:**

Square section with two sides horizontal is shown in the Fig. 29(a).

Section modulus of square section with two sides horizontal,  $Z_1 = \frac{a^3}{6}$

Let  $\sigma$  is the permissible flexural stress.

Moment of resistance,  $M_1 = \sigma Z_1 = \frac{\sigma a^3}{6}$

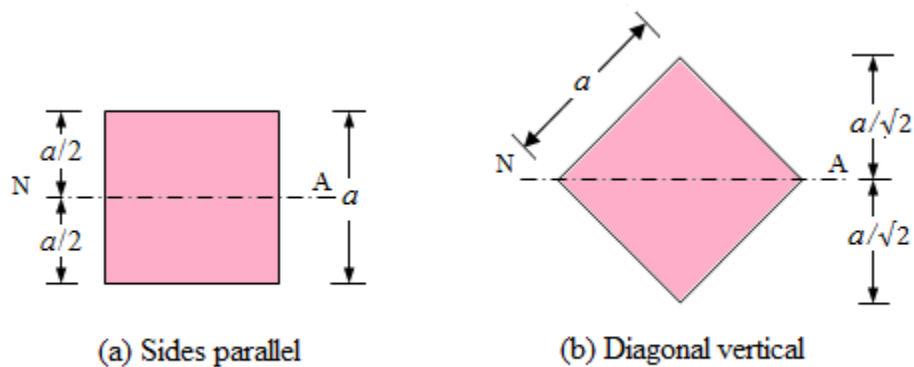


Fig. 29

Square section with one diagonal vertical is shown in the Fig. 29(b).

Moment of inertia about the neutral axis, i.e., the diagonal of the square section = Twice the moment of inertia of triangle of base  $\sqrt{2}a$  and height  $a/\sqrt{2}$ .

## Bending Stress in Beams

$$I_2 = 2 \times \frac{\sqrt{2}a \left( \frac{a}{\sqrt{2}} \right)^3}{12} = \frac{a^4}{12}$$

Extreme fibre distance,

$$y_{\max} = \frac{\sqrt{2}a}{2} = \frac{a}{\sqrt{2}}$$

Section modulus of square section with one diagonal vertical,

$$Z_2 = \frac{I_2}{y_{\max}} = \frac{\frac{a^4}{12}}{\frac{a}{\sqrt{2}}} = \frac{\sqrt{2}a^3}{12}$$

Moment of resistance,

$$M_2 = \sigma Z_2 = \frac{\sqrt{2} \sigma a^3}{12}$$

Ration of the moments of resistance of section in two different positions,

$$\frac{M_1}{M_2} = \frac{\frac{\sigma a^3}{6}}{\frac{\sqrt{2} \sigma a^3}{12}} = \sqrt{2} = 1.414$$

12. Three beams of same material with circular, square and rectangular cross sections have the same length and are subjected to same maximum bending moment. The depth of the rectangular section is twice the width. Compare their weights.

**Solution:**

Fig. 30 shows three different sections, circular, square, and rectangular of beam.

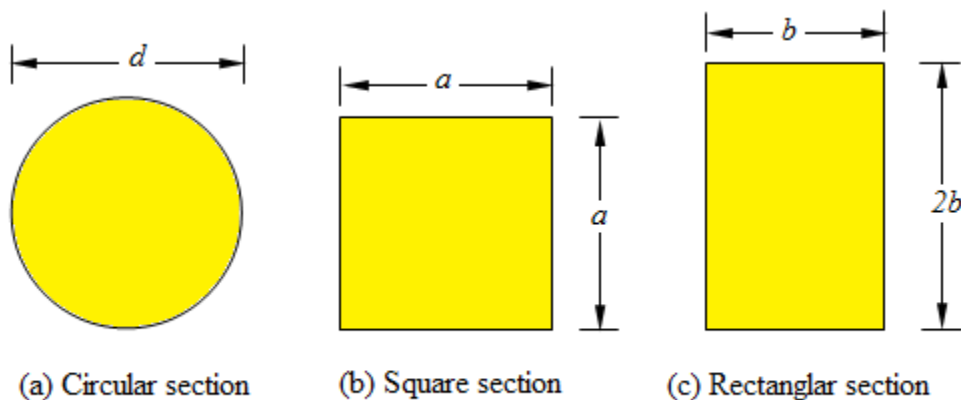


Fig. 30

## Bending Stress in Beams

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Let Diameter of circular section =  $d$ ,

Side of square section =  $a$ , and

Width and depth of rectangular section are  $b$  and  $2b$  respectively.

As beams of three different cross sections of equal allowable stress are subjected same maximum bending moment, they must have same strength. Hence, all sections should have equal section moduli.

$$\text{Section modulus of circular section, } Z_c = \frac{\pi d^3}{32}$$

$$\text{Section modulus of circular section, } Z_s = \frac{a^3}{6}$$

$$\text{Section modulus of circular section, } Z_R = \frac{b(2b)^2}{6} = \frac{2}{3}b^3$$

We have 
$$\frac{\pi d^3}{32} = \frac{a^3}{6} = \frac{2}{3}b^3$$

$$\therefore d = 1.193a \text{ and } b = 0.6299a$$

$$\begin{aligned} \frac{\text{Weight of circular beam}}{\text{Weight of square beam}} &= \frac{\text{Area of circular section}}{\text{Area of square section}} = \frac{\frac{\pi d^2}{4}}{a^2} = \frac{\pi}{4} \left( \frac{d}{a} \right)^2 \\ &= \frac{\pi}{4} (1.193)^3 = 1.118 \end{aligned}$$

$$\begin{aligned} \frac{\text{Weight of rectangular beam}}{\text{Weight of square beam}} &= \frac{\text{Area of rectangular section}}{\text{Area of square section}} = \frac{2b^2}{a^2} = 2 \left( \frac{b}{a} \right)^2 \\ &= 2(0.6299)^2 = 0.7936 \end{aligned}$$

13. A beam of symmetric  $I$ -section has flange size 100 mm x 15 mm, overall depth 250 mm. Thickness of web is 8 mm. Compare the flexural strength of this section with that of a beam of rectangular section of same material and area. The width of rectangular section is two-third of its depth.

### Solution:

The  $I$ -section and the rectangular section of equal area are shown in the Fig. 31.

## Bending Stress in Beams

---

$$\text{Area of } I\text{-section, } A_I = (2 \times 100 \times 15) + (220 \times 8) = 4760 \text{ mm}^2$$

$$\text{Moment of inertia of } I\text{-section, } I_I = \frac{100 \times 250^3}{12} - \frac{92 \times 220^3}{12} = 4.8574 \text{ mm}^4$$

$$\begin{aligned} \text{Section modulus of } I\text{-section, } Z_I &= \frac{I}{y_{\max}} = \frac{4.8574 \times 10^7}{125} \\ &= 388592 \text{ mm}^3 \end{aligned}$$

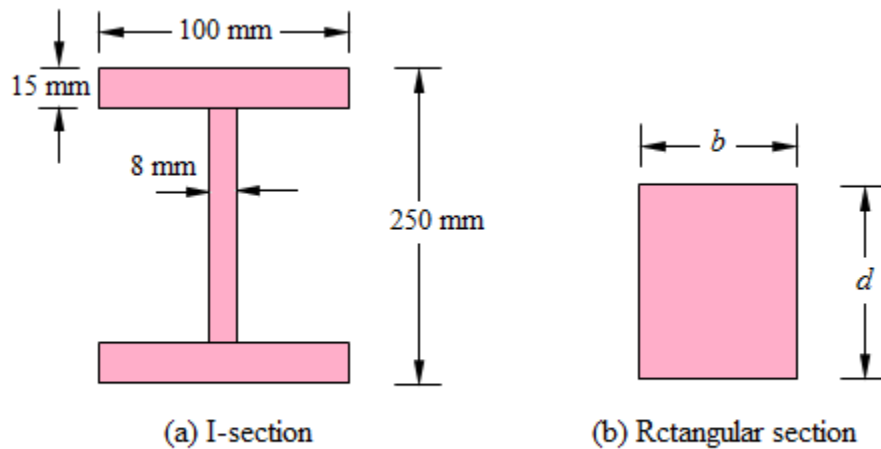


Fig. 31

Let the depth of the rectangular section =  $d$  mm

$$\text{Width of the rectangular section, } b = \frac{2}{3}d$$

$$\text{Area of the rectangular section, } A_R = \frac{2}{3}d \times d = \frac{2}{3}d^2$$

$$\text{Since the area of two sections are equal, } \frac{2}{3}d^2 = 4760$$

$$d = 84.50 \text{ mm}$$

and

$$b = \frac{2}{3} \times 84.50 = 56.33 \text{ mm}$$

$$\begin{aligned} \text{Section modulus of rectangular section, } Z_R &= \frac{bd^2}{6} = \frac{56.33 \times (84.50)^2}{6} \\ &= 67035 \text{ mm}^3 \end{aligned}$$



## Bending Stress in Beams

$$\frac{\text{Flexural strength of I-section}}{\text{Flexural strength of rectangular section}} = \frac{Z_I}{Z_R} = \frac{388592}{67035} = 5.80$$

14. A cast iron beam of an I-section with top flange 80 mm x 40 mm, bottom flange 160 mm x 40 mm and web 120 mm x 20 mm. If the tensile stress is not to exceed 30 N/mm<sup>2</sup> and compressive stress 90 N/mm<sup>2</sup>, what is the maximum uniformly distributed load the beam can carry over a simply supported span of 6 m, if the bottom flange is in tension?

### Solution:

The cross section of the beam is as shown in the Fig. 32.

Let  $\bar{y}$  is the distance of the centroid (neutral axis) from the bottom fibre (tension fibre).

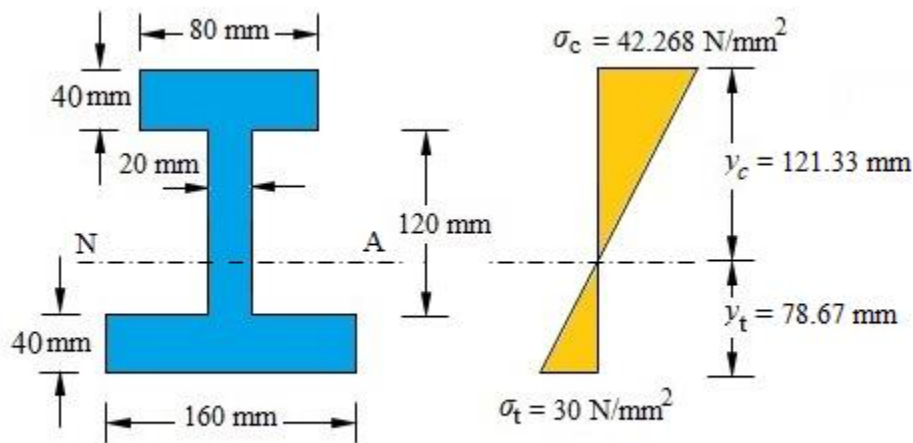


Fig. 32

$$\begin{aligned}\bar{y} &= \frac{\sum a_i y_i}{A} = \frac{160 \times 40 \times 20 + 20 \times 120 \times 100 + 80 \times 40 \times 180}{80 \times 40 + 20 \times 120 + 80 \times 40 \times 120} \\ &= \frac{944000}{12000} = 78.67 \text{ mm}\end{aligned}$$

Moment of inertia,

$$\begin{aligned}I &= \frac{1}{12} \times 160 \times 40^3 + 160 \times 40 \times (78.67 - 20)^2 + \frac{1}{12} \times 20 \times 120^3 + 20 \times 120 \times (100 - 78.67)^2 \\ &\quad + \frac{1}{12} \times 80 \times 40^3 + 80 \times 40 \times (180 - 78.67)^2 \\ &= 60138670 \text{ mm}^4\end{aligned}$$

## Bending Stress in Beams

---

Tension occurs at the bottom and compression at the top.

Bottom extreme fibre distance (large flange, tension flange),  $y_t = 78.67 \text{ mm}$

Top extreme fibre distance (compression flange),  $y_c = 200 - 78.67 = 121.33 \text{ mm}$

Moment of resistance from tensile strength consideration,

$$\begin{aligned} &= \sigma_{allow} \frac{I}{y_t} = 30 \times \frac{60138670}{78.67} = 22933266.81 \text{ N.mm} \\ &= 22.933 \text{ kN.m} \end{aligned}$$

Moment of resistance from compressive strength consideration,

$$\begin{aligned} &= \sigma_{allow} \frac{I}{y_c} = 90 \times \frac{60138670}{121.33} = 44609579.65 \text{ N.mm} \\ &= 44.609 \text{ kN.m} \end{aligned}$$

Hence, actual moment resistance is smaller of the above two, i.e., 22.993 kN

$$\text{Maximum bending moment, } = \frac{wl^2}{8} = \frac{w \times 6^2}{8} = 4.5w$$

Equating the maximum bending moment with the moment of resistance, we have

$$\begin{aligned} 4.5w &= 22.933 \\ w &= 5.096 \text{ kN/m} \end{aligned}$$

**Alternatively,**

Suppose the maximum stress in compression at the top is  $90 \text{ N/mm}^2$ .

Corresponding maximum stress in tension at the bottom,

$$\begin{aligned} \sigma_t &= \frac{y_t}{y_c} \times \sigma_c = \frac{78.67}{121.33} \times 90 \\ &= 58.355 > 30 \text{ N/mm}^2 \quad (\text{Not possible}) \end{aligned}$$

But the permissible tensile stress is only  $30 \text{ N/mm}^2$ . Hence, let the maximum tensile stress be allowed to reach  $30 \text{ N/mm}^2$ .

## Bending Stress in Beams

---

Corresponding maximum compressive stress at the top,

$$\begin{aligned}\sigma_c &= \frac{y_c}{y_t} \times \sigma_t = \frac{121.33}{78.67} \times 30 \\ &= 42.268 \text{ N/mm}^2 < 90 \text{ N/mm}^2 \text{ (OK)}\end{aligned}$$

Hence, the beam will fail in tension at the bottom flange.

Moment of resistance from tensile strength consideration,

$$\begin{aligned}&= \sigma_{allow} \frac{I}{y_t} = 30 \times \frac{60138670}{78.67} = 22933266.81 \text{ N.mm} \\ &= 22.933 \text{ kN.m}\end{aligned}$$

$$\text{Maximum bending moment, } = \frac{wl^2}{8} = \frac{w \times 6^2}{8} = 4.5w$$

Equating the maximum bending moment with the moment of resistance, we have

$$4.5w = 22.933$$

$$w = 5.096 \text{ kN/m}$$

15. Two wooden planks 60 mm x 160 mm each are connected together to form a cross section of a beam as shown in the Fig. If a sagging bending moment of 3500 N.m is applied about the horizontal axis, find the stresses at the extreme fibre of the cross-section. Also calculate the total tensile force on the cross-section.

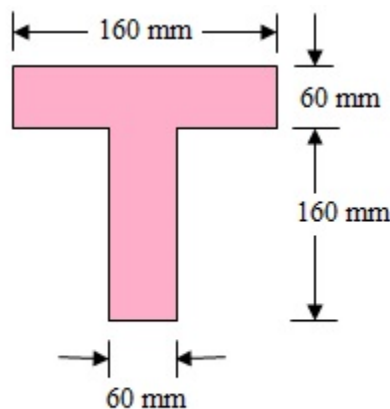


Fig.

## Bending Stress in Beams

### Solution:

Let us locate the centroid and hence the neutral axis, and find moment of inertia of the section. Consider the bottom of T-section as the reference axis for location of centroid. The T-section consists of two components, web and flange.

The relevant calculations are shown in the table.

Distance of the centroidal axis GG from the bottom edge,

$$\bar{y} = \frac{\sum ay}{\sum a} = \frac{2610000}{19200} = 135.94 \text{ mm}$$

Moment of inertia at the bottom edge,  $I_b = \sum I_{self} + \sum ay^2$

$$= 23.36 \times 10^6 + 408 \times 10^6 = 431.36 \times 10^6 \text{ mm}^4$$

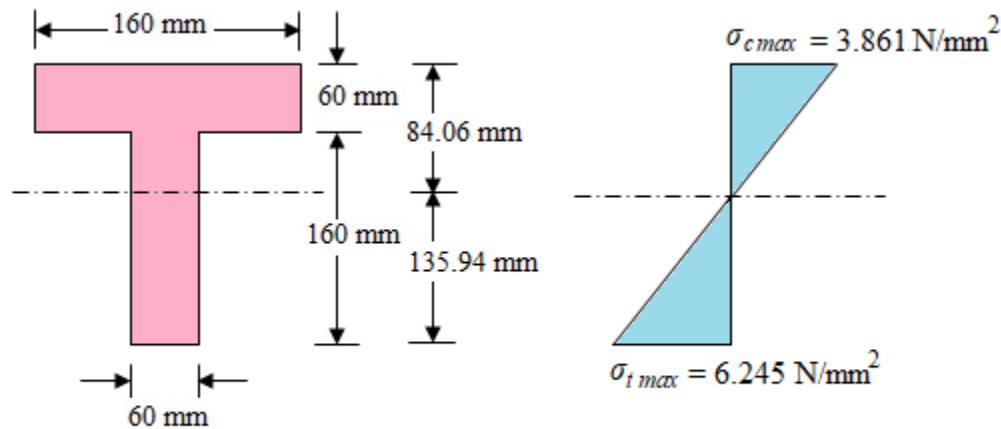


Fig.

But,

$$I_b = I_G + (\sum a)\bar{y}^2$$

$$I_G = I_b - (\sum a)\bar{y}^2 = 431.36 \times 10^6 - 19200 \times 135.94^2$$

$$= 76190074.88 \text{ mm}^4$$

Let the maximum tensile and compressive stresses at extreme fibres be  $\sigma_{max}$  and  $\sigma_{cmax}$  respectively.

## Bending Stress in Beams

Components	Area $a$ ( $mm^2$ )	Centroidal distance from the bottom edge, $y$ ( $mm$ )	$ay$ ( $mm^3$ )	$ay^2$ ( $mm^4$ )	$I_{Self}$ ( $mm^4$ )
Web	9600	80	786000	$61.44 \times 10^6$	$\frac{60 \times 160^3}{12} = 20.48 \times 10^6$
Flange	9600	190	1824000	$346.56 \times 10^6$	$\frac{160 \times 60^3}{12} = 2.88 \times 10^6$
Total	19200		2610000	$408 \times 10^6$	$23.36 \times 10^6$

We have,

$$\sigma_{t_{max}} = \frac{M}{I} y_t = \frac{3500 \times 1000}{76190074.88} \times 135.94$$

$$= 6.245 \text{ N/mm}^2$$

$$\sigma_{c_{max}} = \frac{M}{I} y_c = \frac{3500 \times 1000}{76190074.88} \times 84.06$$

$$= 3.861 \text{ N/mm}^2$$

Total tensile force = Average tensile stress x area of tensile zone

$$= \frac{6.245}{2} \times (135.94 \times 60) = 25468.359 \text{ N}$$

16. A water main of 1000 mm internal diameter and 10 mm thickness is running full. If the bending stress is not to exceed  $56 \text{ N/mm}^2$ , find the greatest span on which the pipe may be freely supported. Steel and water weigh  $76800 \text{ N/m}^3$  and  $10000 \text{ N/m}^3$  respectively.

### Solution:

Internal diameter of the pipe,  $d = 1000 \text{ mm} = 1 \text{ m}$

External diameter of pipe,  $D = 1000 + 2 \times 10 = 1020 \text{ mm} = 1.02 \text{ m}$

Consider 1 m length of the water main.

$$\text{Area of the pipe section, } A = \frac{\pi}{4} (D^2 - d^2) = \frac{\pi}{4} (1.02^2 - 1^2)$$

$$= 0.03173 \text{ m}^2$$

## Bending Stress in Beams

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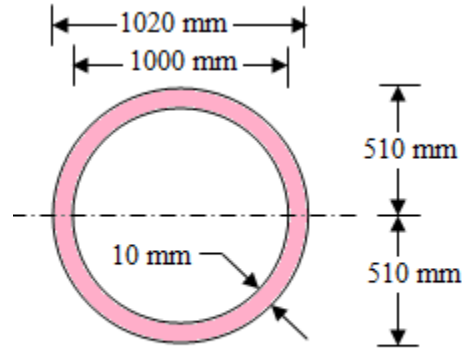


Fig.

$$\text{Area of the water section, } A = \frac{\pi}{4} d^2 = \frac{\pi}{4} \times 1^2$$

$$= 0.7854 \text{ mm}^2$$

$$\text{Weight of one metre length of pipe} = 0.03173 \times 1 \times 76800 = 2493.978 \text{ N}$$

$$\text{Weight of water in one metre length of the pipe} = 0.7854 \times 1 \times 10000 = 7854 \text{ N}$$

$$\text{Total load on the pipe per metre run} = 2493.978 + 7854 = 10347.978 \text{ N}$$

Let the maximum span of the pipe  $l$  m.

$$\text{Maximum bending moment, } M = \frac{wl^2}{8} = \frac{10347.978l^2}{8} = 1293.497l^2 \text{ N.m}$$

$$= 1293.497 \times 1000l^2 \text{ N.mm}$$

Moment of inertia of the pipe section about the neutral axis,

$$I = \frac{\pi}{64} (D^4 - d^4) = \frac{\pi}{64} (1020^4 - 1000^4)$$

$$= 4046.379 \times 10^6 \text{ mm}^4$$

We know,

$$\frac{M}{I} = \frac{\sigma}{y}$$

$$\frac{1293.497 \times 1000l^2}{4046.379 \times 10^6} = \frac{56}{510}$$

## Bending Stress in Beams

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$$l^2 = \frac{56 \times 4046.379 \times 10^6}{510 \times 1293.497 \times 1000} = 343.494$$

$$l = 18.533m$$

### Shear Force and Bending Moment

**TYPES OF FORCES:** Basically, structural members experience two types of forces.

**External Forces:** Actions of other bodies on the structure under consideration are known as external forces.

**Internal Forces:** Forces and couples exerted on a member or portion of the structure by the rest of the structure. Internal forces always occur in equal but opposite pairs.

#### TYPES OF LOAD

The following are the important types of load which act on a beam.

1. Concentrated or point load,
  2. Uniformly distributed load, and
  3. Uniformly varying load
1. **Concentrated or Point Load:** Load acting at a point or over very limited area compared to the length of the beam is known as concentrated load or point load.

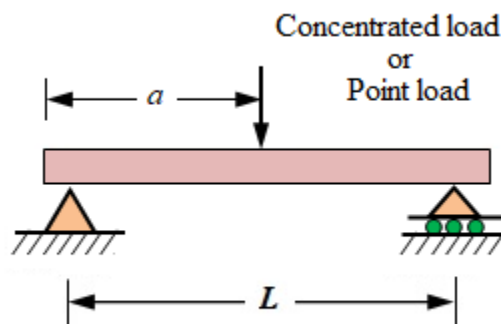


Fig. 1

2. **Uniformly Distributed Load:** Load that is spread over a beam with uniform rate of loading, (' $w$ ' per unit run) is known as uniformly distributed load or *UDL*. Uniformly distributed load is also known as rectangular load.



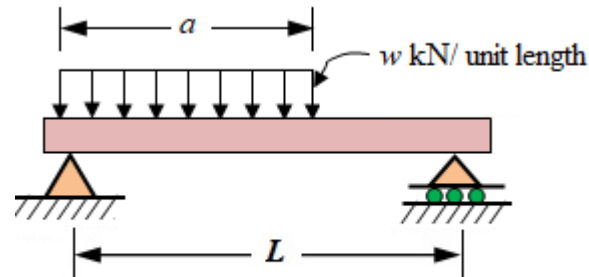


Fig. 2

3. **Uniformly Varying Load:** Load that is spread over a beam with the rate of loading uniformly from one point to the other along the beam is known as uniformly varying load. Uniformly varying distributed load is also known as triangular load.

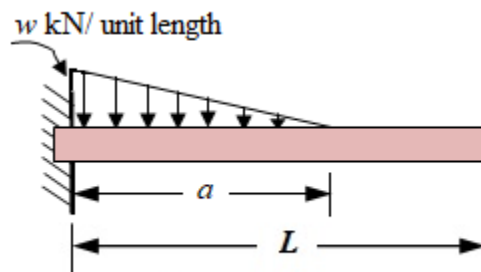


Fig. 3

4. **Parabolic Load:** If the variation of load distribution follows the equation of parabola, it is known as parabolic distributed load or simply parabolic load.

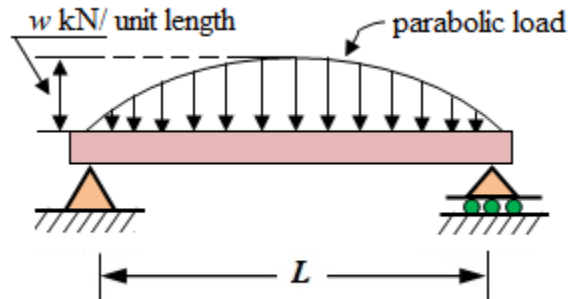


Fig. 4

### TYPES OF SUPPORTS

1. Simple support
2. Roller Support

## *Shear Force and Bending Moment*

---

3. Pin (or) Hinge Support
4. Fixed support

### **Simple Supports**

Simple support is just a support on which structural member rests. It is idealized to be a frictionless surface support. It only resists vertical movement of support. A simple support is free to rotate and translate along the surface upon which it rests. The resulting reaction force is always a single force perpendicular to the plane of support.

The horizontal or lateral movement allowed is up to a limited extent and after that the structure loses its support. For example, if a plank is laid across gap to provide a bridge, it is assumed that the plank will remain in its place. It will do so until a foot kicks it or moves it. At that moment the plank will move because the simple connection cannot develop any resistance to the lateral load.

This type of support is not commonly used in structural purposes. However, Simple supports are often found in zones of frequent seismic activity.

### **Roller Supports**

Roller supports are free to rotate and translate along the surface upon which they rest. The surface can be horizontal, vertical, or sloped at any angle. They cannot resist parallel or horizontal forces and moment. They only resist perpendicular forces. Hence, the resulting reaction force is always a single force that is perpendicular to the plane of support.

This type of support is provided at one end of bridge spans. The reason for providing roller support at one end is to allow contraction or expansion of bridge deck with respect to temperature differences in atmosphere. If roller support is not provided then it will cause severe damage to the banks of bridge. But this horizontal force should be resisted by at least one support to provide stability so, roller support should be provided at one end only not at both ends.

### **Pinned Supports**

A pinned support is same as hinged support. It can resist both vertical and horizontal forces but not a moment. It allows the structural member to rotate, but not to translate in any direction.

## Shear Force and Bending Moment


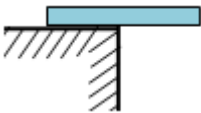
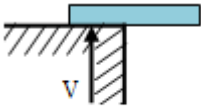

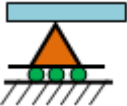
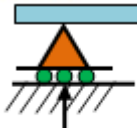
Many connections are assumed to be pinned connections even though they might resist a small amount of moment in reality. It is also true that a pinned connection could allow rotation in only one direction; providing resistance to rotation in any other direction. In human body knee is the best example of hinged support as it allows rotation in only one direction and resists lateral movements. Ideal pinned and fixed supports are rarely found in practice, but beams supported on walls or simply connected to other steel beams are regarded as pinned. The distribution of moments and shear forces is influenced by the support condition.

Best example for hinged support is door leaf which only rotates about its vertical axis without any horizontal or vertical movement.


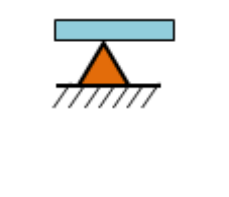
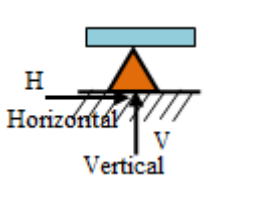

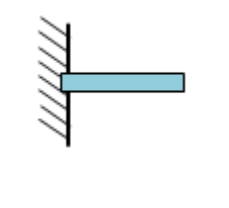
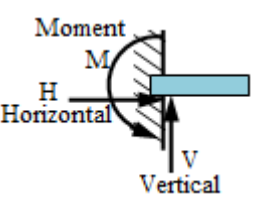
### Fixed Supports

Fixed support can resist vertical and horizontal forces as well as moment since they restrain both rotation and translation. They are also known as rigid support for the stability of a structure there should be one fixed support. A flagpole at concrete base is common example of fixed support In RCC structures the steel reinforcement of a beam is embedded in a column to produce a fixed support as shown in above image. Similarly all the riveted and welded joints in steel structure are the examples of fixed supports Riveted connection are not very much common now a days due to the introduction of bolted joints.

**Table 1. Idealized Structural Supports**

Types of supports	Real life Example	Symbol	Movement allowed and prevented	Unknown reactions
Frictionless or Simple support			Prevented: vertical translation Allowed: horizontal translation and rotation	 Reaction normal to plane of support
Roller support			Prevented: vertical translation Allowed: horizontal translation and rotation	 Reaction normal to plane of support

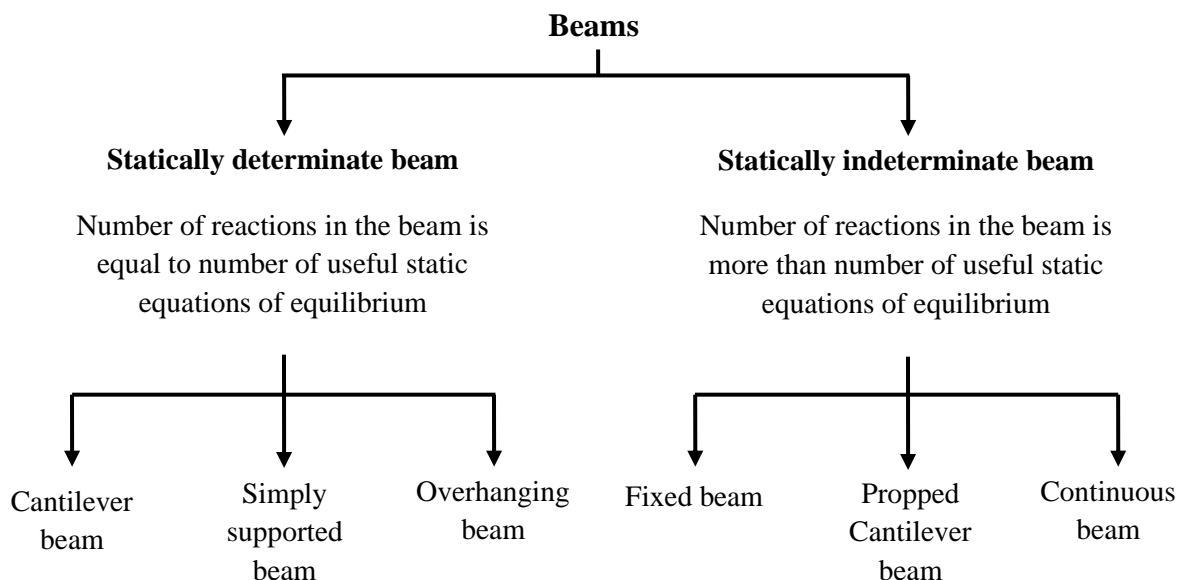
## Shear Force and Bending Moment

Pinned or hinged support			Prevented: horizontal translation and vertical translation Allowed: Rotation	
Fixed or Built-in support			Prevented: horizontal translation, vertical translation and rotation	

### BEAM:

A **Beam** is defined as a structural member subjected to transverse shear loads (load normal to the axis of the beam) during its functionality. Due to the transverse shear loads, a beam is subjected to variable shear force and bending moment. Beam is a flexural member, designed primarily for bending. Analysis of beam pertains to the calculations of shear forces and bending moments along the length of the beam and drawing of shear force diagram and bending moment diagram.

**TYPES OF BEAMS:** Depending upon the degrees of freedom and support conditions beams are of various types.



## *Shear Force and Bending Moment*

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### **Statically Determinate Beam**

A beam is said to be statically determinate if all its reaction components can be calculated by applying three conditions of static equilibrium.

### **Statically Indeterminate Beam**

When the number of unknown reaction components exceeds the static conditions of equilibrium, the beam is said to be statically indeterminate. To determine the unknown reactions additional equations of deformations are required.

### **The following are the important types of beam**

1. Cantilever beam,
2. Simply supported beam,
3. Overhanging beam,
4. Fixed beams, and
5. Continuous beam.

#### **1. Cantilever beam**

A beam which is fixed or built into a rigid support at one end and free at the other end is known as cantilever beam. Such beam is shown in Fig. The built-in support prevents displacements as well as rotations of the end of the beam. Cantilever is statically determinate.

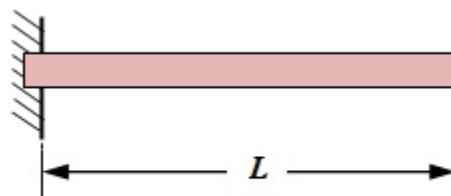


Fig.5 Cantilever beam

#### **2. Simply Supported beam**

A beam supported or resting freely on the supports at its both ends is known as simply supported beam. Such beam is shown in Fig. The end supports are free to rotate and have no moment of resistance. Simply supported beam is statically determinate beam.

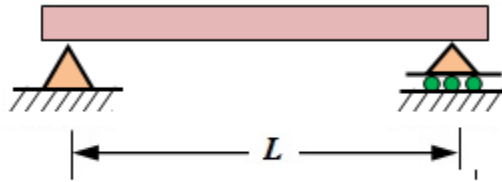


Fig.6 Simply supported beam

### 3. Overhanging Beam

A beam supported over two supports and extended beyond one or both the supports is known as overhanging beam. An *overhanging beam*, shown in Fig., is supported by a pin and a roller support, with one or both ends of the beam extending beyond the supports. It is a statically determinate beam.

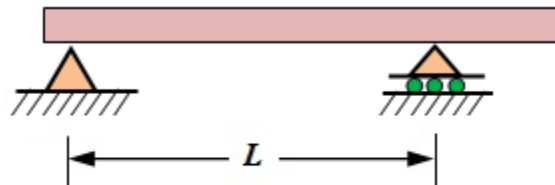


Fig.7 Overhanging beam

### 4. Fixed Beam

A beam with both ends fixed or built into the supports or walls, is known as fixed beam. Such beam is shown in Fig. A fixed beam is also known as a built-in or encasted beam. It is a statically indeterminate beam.

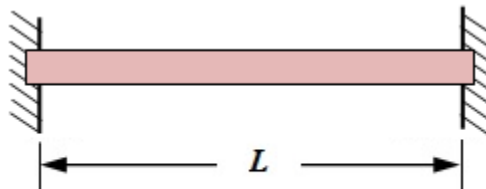


Fig.8 Fixed beam

### 5. Propped cantilever beam

A beam with one end fixed and the other end simply supported over a roller is known as propped cantilever beam or simply propped cantilever. Propped cantilever is statically indeterminate.

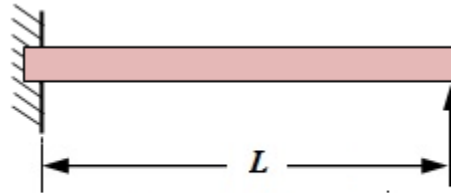


Fig.9 Propped cantilever

### 6. Continuous Beam

A beam which is supported over more than two supports is known as continuous beam. Continuous beam is also statically indeterminate.

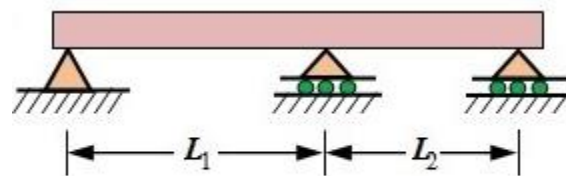


Fig.10 Continuous beam

### SHEAR FORCE AND BENDING MOMENT:

The beams transfer the transverse (vertical) loads to the supports. In the process of load transfer, they experience shear force and bending moments.

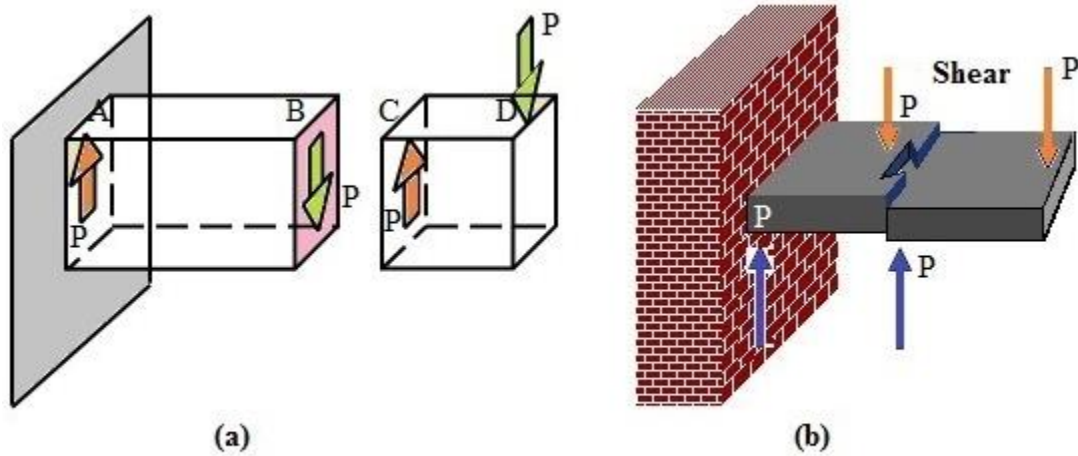


Fig.11 Shearing off beam

## Shear Force and Bending Moment

Shear force at any section of a beam is defined as the *net or unbalanced vertical force on either side of the section*. It is the algebraic sum of vertical components of all the forces acting on the beam on either left side or right side of the section. The effect of shear force is to shear off or cut the member at a section. It is similar to the effect of scissor cutting the page of paper.

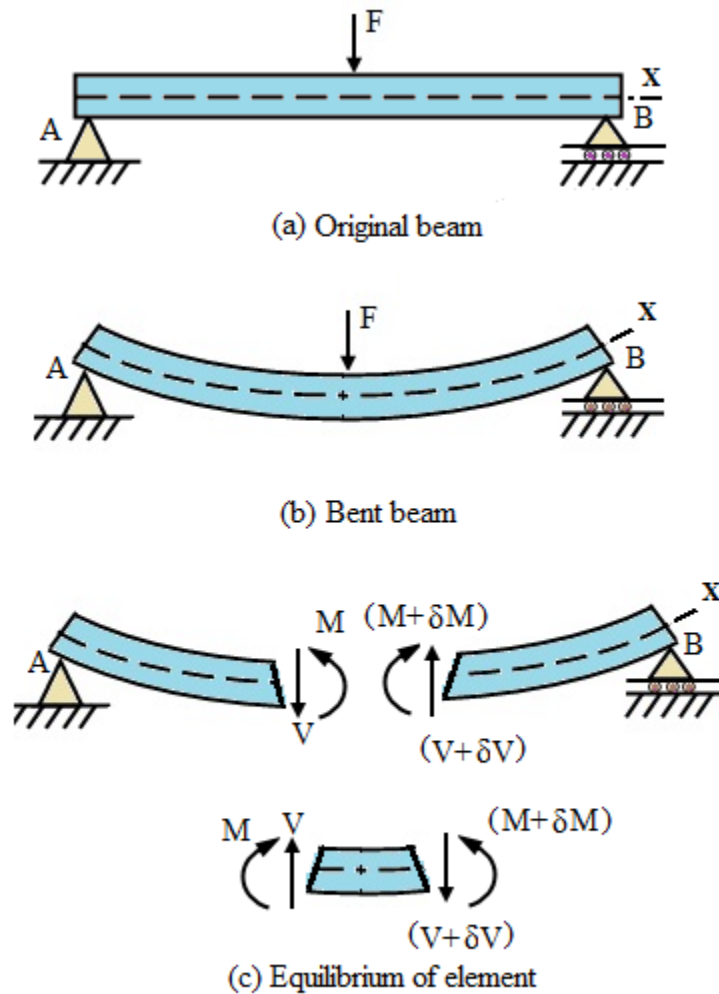


Fig.11 Bending of beam

The moment which tends to bend the beam in plane of load is known as bending moment. In other word bending moment at any section of a beam is the *net or unbalanced moment due to all forces on either side of the section*. Bending moment at any section is the algebraic sum of the moments due to all forces acting on the beam on either right or left side of the section. The effect of bending moment is to bend the element.



## Shear Force and Bending Moment

---

### Sign convention:

The shear force and bending moment are vector quantities and as a matter of convenience are assigned the following sign convention.

Shear force acting in the upward direction to the left hand side of the section and downward direction to the right hand side of the section is considered to be positive & vice-versa.

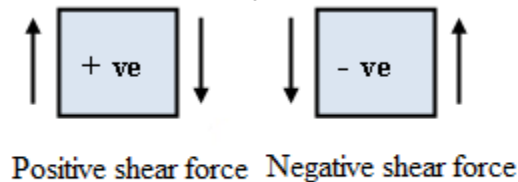


Fig.12

Bending moment is considered to be positive when it is acting in the clockwise direction on the left hand side of the section (L.H.S) (or) when it is acting in the counter-clockwise direction on the right hand side of the section (R.H.S) as the section & vice versa.

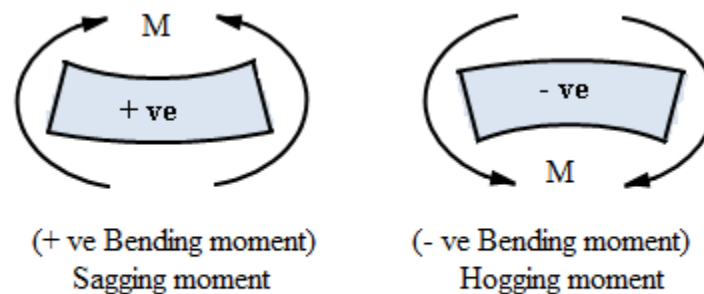


Fig. 13

### SHEAR FORCE AND BENDING MOMENT DIAGRAMS:

Graphical representation of variation of shear force along the length of the beam for any given loading condition is known as **shear force diagram** (SFD). If  $x$  denotes the length of the beam, then shear force ' $F$ ' is function of  $x$ , i.e.  $F(x)$ .

Similarly, graphical representation of variation of bending moment along the length of the beam for any given loading condition is known as **bending moment diagram** (BMD). If  $x$  denotes the length of the beam, then bending moment is function of  $x$ , and is denoted as  $M(x)$ .

## Shear Force and Bending Moment

---

Shear force diagram and bending moment diagram are helpful for further analysis and design of beam.

SFD and BMD of a beam reveal the following important information at salient points in the beam. These are maximum shear force, maximum bending moment, point of contraflexure or point of inflexion, etc.

### RELATIONS BETWEEN LOAD, SHEAR FORCE AND BENDING MOMENT

Consider a beam  $AB$  carrying generalized loading as shown in the figure. Take an element of infinitesimal length  $\delta x$  between section 1-1 and 2-2 at a distance of  $x$  from the left hand support  $A$ . The free body diagram of the element is drawn with positive sense of the shear forces and bending moments.

The intensity of loading over the length of the element may be taken as constant, i.e.,  $w$ . Considering equilibrium of the element,

Resolving the forces vertically,  $\sum V = 0$

$$F = w\delta x + F + \delta F$$

$$\delta F = -w\delta x$$

$$\frac{\delta F}{\delta x} = -w$$

In the limiting case, as  $\delta x \rightarrow 0$ ,  $\frac{dF}{dx} = -w$  (1)

So, the rate of change of shear force is equal to the intensity or rate of loading.

Taking moments of the forces and couples about the section 2-2,  $\sum M_2 = 0$

$$M + \delta M + w \frac{(\delta x)^2}{2} = M + F\delta x$$

Neglecting small quantities of higher order, we have

$$\frac{\delta M}{\delta x} = F$$

## Shear Force and Bending Moment

In the limiting case as  $\delta x \rightarrow 0$ ,  $\frac{dM}{dx} = F$  (2)

The above equation shows that the rate of change of bending moment is equal to the shear force at the section. Also bending moment would be maximum at a section where shear force is zero.

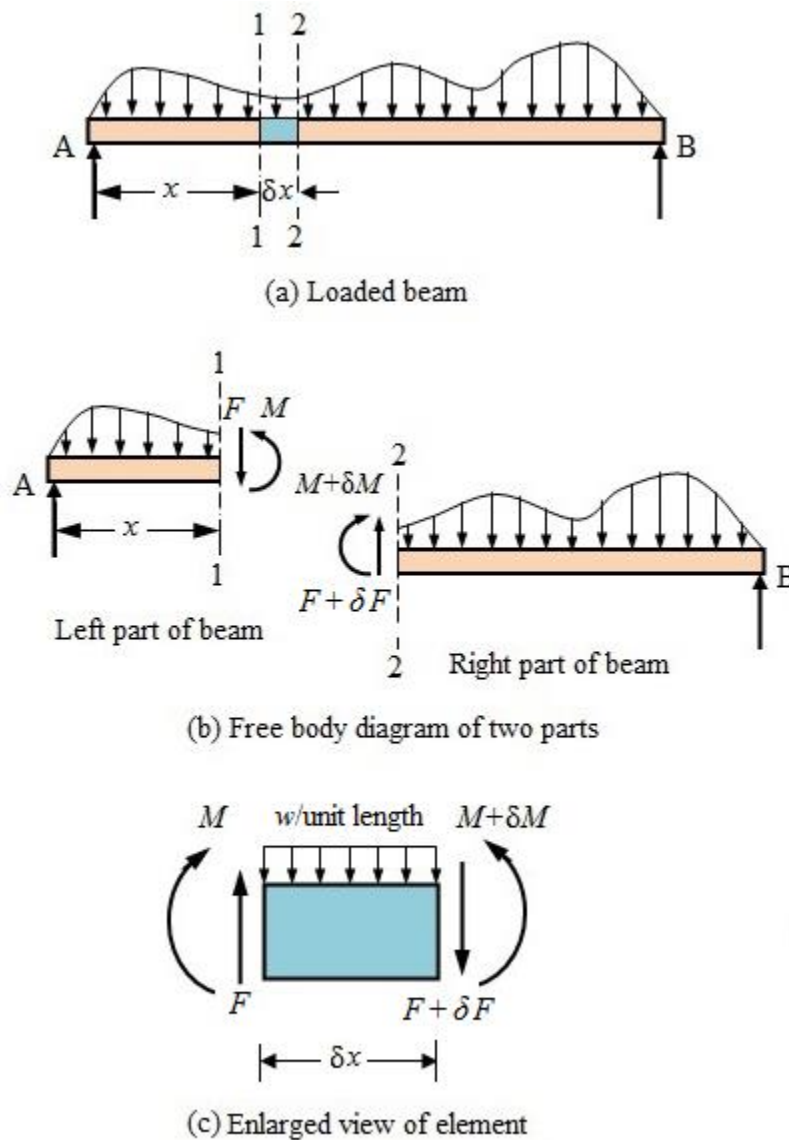


Fig. 14

### Evaluation of Shear Force and Bending Moment:

Thus analysis of beam for shear force and bending moment is carried out by the following process.

## Shear Force and Bending Moment

---

1. Determine the reactions at the supports by considering the entire beam as a rigid body and applying equations of equilibrium.
2. Take sections at different points on the beam near supports and load application points.
3. Apply equilibrium analyses on resulting free-bodies to determine internal shear forces and bending moments.
4. Draw shear force and bending moment diagram.
5. Identify the maximum shear and bending-moment from plots of their distributions.
6. Find the position of *point of contraflexure* or *point of inflexion*.

### Numerical

1. Draw the Shear force and bending moment diagram for a cantilever beam of length  $L$  carrying a point load  $W$  at its free end.

#### Solution:

#### Evaluation of support reactions:

Considering the equilibrium of the beam and applying static equations of equilibrium,

Sum of the vertical forces,  $\sum V = 0$ ,  $V_A = W$  ( $\uparrow$ )

Taking moment about A,  $\sum M_A = 0$ ,  $W \times L + M_A = 0$

$$M_A = -WL \text{ (counter - clockwise)}$$

#### Calculation of Shear force and bending moments:

Considering from the right hand side  $B$ , as the origin, take a section 1-1 at a distance of  $x$  from  $B$  between  $B$  and  $A$  ( $0 \leq x \leq L$ ).

Shear force at 1-1,  $F_x = W$

Shear force at  $B$ , i.e.,  $x = 0$   $F_B = W$

Shear force at  $A$ , i.e.,  $x = L$ ,  $F_A = W$

Bending moment at 1-1,  $M_x = -Wx$

Bending moment at  $B$ , i.e.,  $x = 0$ ,  $M_B = 0$

## Shear Force and Bending Moment

Bending moment at A, i.e.,  $x = L, M_A = -WL$

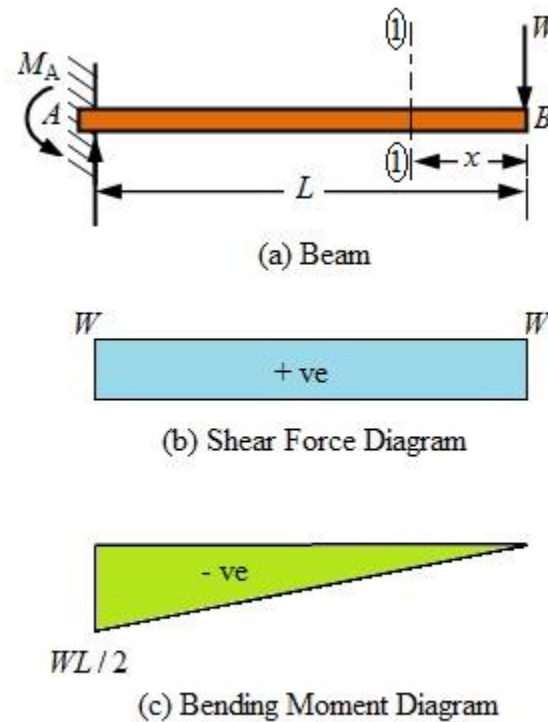


Fig.

2. Draw the Shear force and bending moment diagram for a cantilever beam of length  $L$  carrying uniformly distributed load of intensity  $w$  over the entire span.

**Solution:**

**Evaluation of support reactions:**

Considering the equilibrium of the beam and applying static equations of equilibrium,

Sum of the vertical forces,  $\sum V = 0, V_A = wL$

Taking moment about A,  $\sum M_A = 0, wL \times \frac{L}{2} + M_A = 0$

$$M_A = -\frac{wL^2}{2} \text{ (counter - clockwise)}$$

**Calculation of Shear force and bending moments:**

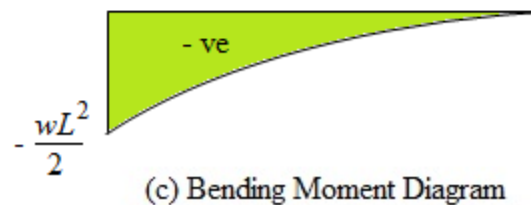
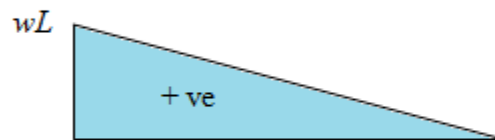
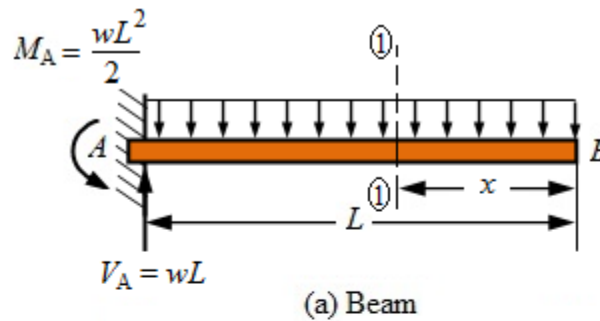
## Shear Force and Bending Moment

Considering from the right hand side  $B$ , as the origin, take a section 1-1 at a distance of  $x$  from  $B$  between  $B$  and  $A$  ( $0 \leq x \leq L$ ).

Shear force at 1-1,  $F_x = wx$

Shear force at  $B$ , i.e.,  $x = 0$   $F_B = 0$

Shear force at  $A$ , i.e.,  $x = L$ ,  $F_A = wL$



Bending moment at 1-1,  $M_x = -wx \times \left(\frac{x}{2}\right) = -\frac{wx^2}{2}$

Bending moment at  $B$ , i.e.,  $x = 0$ ,  $M_B = 0$

Bending moment at  $A$ , i.e.,  $x = L$ ,  $M_A = -\frac{wL^2}{2}$

3. Draw the Shear force and bending moment diagram for a cantilever beam of length  $L$  carrying uniformly distributed load of intensity  $w$  per unit length from the fixed support to the centre of the beam.

## Shear Force and Bending Moment

**Solution:**

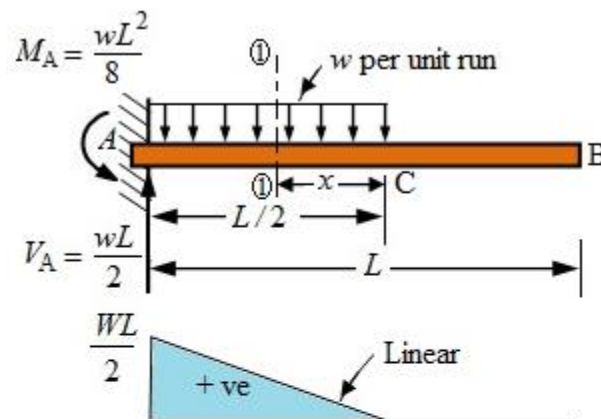
**Evaluation of support reactions:**

Considering the equilibrium of the beam and applying static equations of equilibrium,

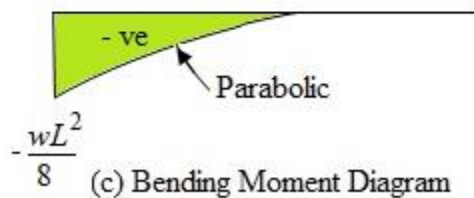
Sum of the vertical forces,  $\sum V = 0$ ,  $V_A = \frac{wL}{2}$

Taking moment about A,  $\sum M_A = 0$ ,  $\frac{wL}{2} \times \frac{L}{4} + M_A = 0$

$$M_A = -\frac{wL^2}{8} \text{ (counter - clockwise)}$$



(b) Shear Force Diagram



(c) Bending Moment Diagram

Fig.

**Calculation of Shear force and bending moments:**

Shear force and bending moment at the free end B,  $F_B = 0$ ;  $M_B = 0$

Shear force and bending moment anywhere between B and C is zero since there is no load on the beam in this portion when considered from right side.

## Shear Force and Bending Moment

---

Now, considering  $C$  as the origin, take a section 1-1 at a distance of  $x$  from  $C$  between  $C$  and  $A$  ( $0 \leq x \leq \frac{L}{2}$ ).

Shear force at 1-1,  $F_x = wx$

Shear force at  $A$ , i.e.,  $x = L$ ,  $F_A = w \frac{L}{2} = \frac{wL}{2}$

Bending moment at 1-1,  $M_x = -wx \times \left(\frac{x}{2}\right) = -\frac{wx^2}{2}$

Bending moment at  $A$ , i.e.,  $x = \frac{L}{2}$ ,  $M_A = -\frac{w\left(\frac{L}{2}\right)^2}{2}$   
 $= -\frac{wL^2}{8}$

4. Draw the Shear force and bending moment diagram for a cantilever beam of length  $L$  carrying uniformly distribute load of intensity  $w$  per unit length from the free end up to a distance of  $a$ .

### Solution:

#### Evaluation of support reactions:

Considering the equilibrium of the beam and applying static equations of equilibrium,

Sum of the vertical forces,  $\sum V = 0$ ,  $V_A = wa$

Taking moment about  $A$ ,  $\sum M_A = 0$ ,  $wa \times \left(L - \frac{a}{2}\right) + M_A = 0$

$$M_A = -\frac{wa}{2}(L - 2a) \quad (\text{counter - clockwise})$$

#### Calculation of Shear force and bending moments:

Shear force and bending moment at the free end  $B$ ,  $F_B = 0$ ;  $M_B = 0$

Now, considering  $B$  as the origin, take a section 1-1 at a distance of  $x$  from  $B$  between  $B$  and  $C$  ( $0 \leq x \leq a$ ).

Shear force at 1-1,  $F_x = wx$



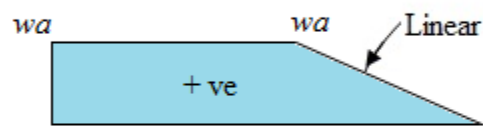
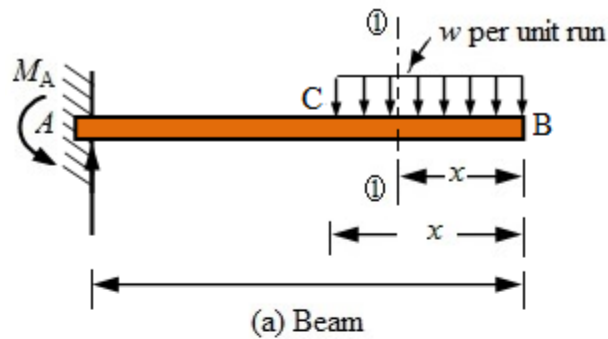
## Shear Force and Bending Moment

Shear force at B, i.e.,  $x = 0$ ,  $F_C = w \times 0 = 0$

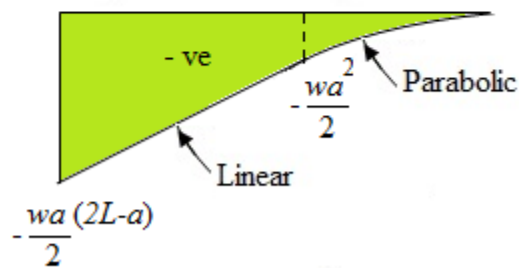
Shear force at C, i.e.,  $x = a$ ,  $F_C = wa$

Bending moment at 1-1,  $M_x = -wx \times \left(\frac{x}{2}\right) = -\frac{wx^2}{2}$

Bending moment at C, i.e.,  $x = a$ ,  $M_A = -\frac{wa^2}{2}$



(b) Shear Force Diagram



(c) Bending Moment Diagram

Fig.

Now, take a section 2-2 at a distance of  $x$  from B between C and A ( $a \leq x \leq L$ ).

Shear force at 2-2,  $F_x = F_C = wa$

Shear force will remain same as  $wa$  from C to A.

Bending moment at 2-2,  $M_x = -wa \times \left(x - \frac{a}{2}\right)$

## Shear Force and Bending Moment

$$\text{Bending moment at A, i.e., } x = L, M_x = -wa\left(L - \frac{a}{2}\right) = -\frac{wa}{2}(2L - a)$$

5. A cantilever of 3.5 m long carries point loads of 15 kN, 15 kN and 7.5 kN at 1 m, 1 m and 1.5 m respectively from the fixed end. Draw the Shear force and bending moment diagram for the beam.

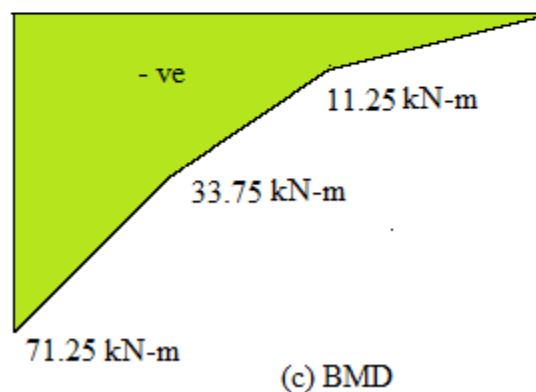
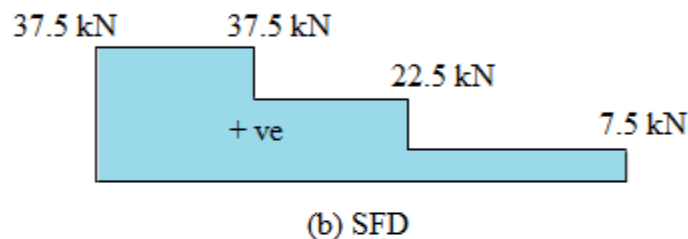
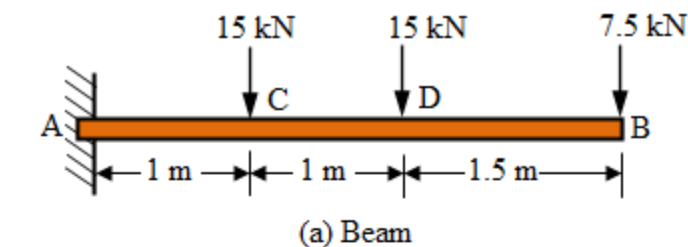
**Solution: Calculation of Shear force and bending moments:**

**Portion BD:** At section 1-1 at a distance  $x$  from  $B$  between  $B$  and  $D$  ( $0 \leq x \leq 1.5\text{m}$ )

Shear force at 1-1,  $F_x = 7.5\text{ kN}$  (constant from  $B$  to just right of  $D$ )

Shear force at  $B$ ,  $F_B = 7.5\text{ kN}$

Shear force just right of  $D$ ,  $F_{DL} = 7.5\text{ kN}$



**Fig.**

## Shear Force and Bending Moment

---

Bending moment at 1-1,  $M_x = -7.5x$

Bending moment at B, i.e., at  $x = 0$ ,  $M_B = -7.5 \times 0 = 0$

Bending moment at D, i.e., at  $x = 1.5$ ,  $M_D = -7.5 \times 1.5 = 11.25 \text{ kN-m}$

**Portion DC:** At section 2-2 at a distance  $x$  from B between D and C ( $1.5 \leq x \leq 2.5 \text{ m}$ )

Shear force at 2-2,  $F_x = 7.5 + 15$  (constant from D to just right of C)

Shear force at D,  $F_D = 22.5 \text{ kN}$

Shear force just right of C,  $F_{CL} = 22.5 \text{ kN}$

Bending moment at 2-2,  $M_x = -7.5x - 15(x - 1.5)$   
 $= -22.5x + 22.5$

Bending moment at C, i.e., at  $x = 2.5$ ,  $M_B = -22.5 \times 2.5 + 22.5 = -33.75 \text{ kN-m}$

**Portion CA:** At section 3-3 at a distance  $x$  from B between C and A ( $2.5 \leq x \leq 3.5 \text{ m}$ )

Shear force at 3-3,  $F_x = 7.5 + 15 + 15$  (constant from C to A)

Shear force at C,  $F_C = 37.5 \text{ kN}$

Shear force at A,  $F_A = 37.5 \text{ kN}$

Bending moment at 3-3,  $M_x = -7.5x - 15(x - 1.5) - 15(x - 2.5)$   
 $= -37.5x + 60$

Bending moment at A, i.e., at  $x = 3.5 \text{ m}$ ,  $M_B = -37.5 \times 3.5 + 60 = -71.25 \text{ kN-m}$

6. A cantilever of  $1.6 \text{ m}$  long carries a uniformly distributed load of intensity  $1.5 \text{ kN/m}$  over the entire span and a point load of  $2.5 \text{ kN}$  at the free end. Draw the Shear force and bending moment diagram for the beam.

**Solution:**

**Calculation of Shear force and bending moments:**

Considering from the right hand side B, as the origin, take a section 1-1 at a distance of  $x$  from B between B and A ( $0 \leq x \leq 1.6 \text{ m}$ ).

Shear force at 1-1,  $F_x = 2.5 + 1.5x$

Shear force at B, i.e.,  $x = 0$ ,  $F_B = 2.5 + 1.5 \times 0 = 2.5 \text{ kN}$

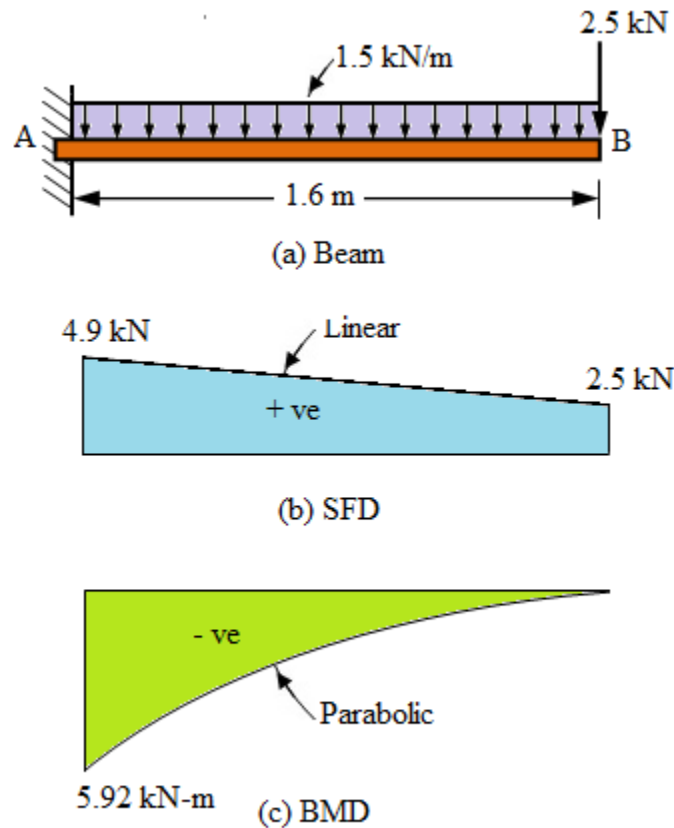
Shear force at A, i.e.,  $x = 1.6 \text{ m}$ ,  $F_x = 2.5 + 1.5 \times 1.6 = 4.9 \text{ kN}$

## Shear Force and Bending Moment

$$\begin{aligned} \text{Bending moment at 1-1, } M_x &= -2.5x - 1.5x\left(\frac{x}{2}\right) \\ &= -2.5x - 0.75x^2 \end{aligned}$$

$$\text{Bending moment at } B, \text{ i.e., } x = 0, M_B = 0$$

$$\begin{aligned} \text{Bending moment at } A, \text{ i.e., } x = 1.6, M_B &= -2.5 \times 1.6 - 0.75 \times 1.6^2 \\ &= -5.92 \text{ kN-m} \end{aligned}$$



**Fig.**

7. A cantilever of 1.5 m long is loaded with a uniformly distributed load of intensity 2 kN/m and a point load of 2.5 kN as shown in the figure. Draw the Shear force and bending moment diagram for the cantilever.

### Calculation of Shear force and bending moments:

Considering from the right hand side B, as the origin, take a section 1-1 at a distance of  $x$  from B between B and D ( $0 \leq x \leq 0.25 \text{ m}$ ).

$$\text{Shear force at 1-1, } F_x = 2x$$

## Shear Force and Bending Moment

Shear force at B, i.e.,  $x = 0$ ,  $F_B = 2 \times 0 = 0$

Shear force just right of D, i.e.,  $x = 0.25\text{m}$ ,  $F_D = 2 \times 0.25 = 0.5\text{kN}$

Shear force at D, i.e.,  $x = 0.25\text{m}$ ,  $F_D = 0.5 + 2.5 = 3\text{kN}$

Bending moment at 1-1,  $M_x = -2x\left(\frac{x}{2}\right) = -x^2$

Bending moment at D, i.e.,  $x = 0.25$ ,  $M_D = -(0.25)^2 = -0.0625\text{kN-m}$

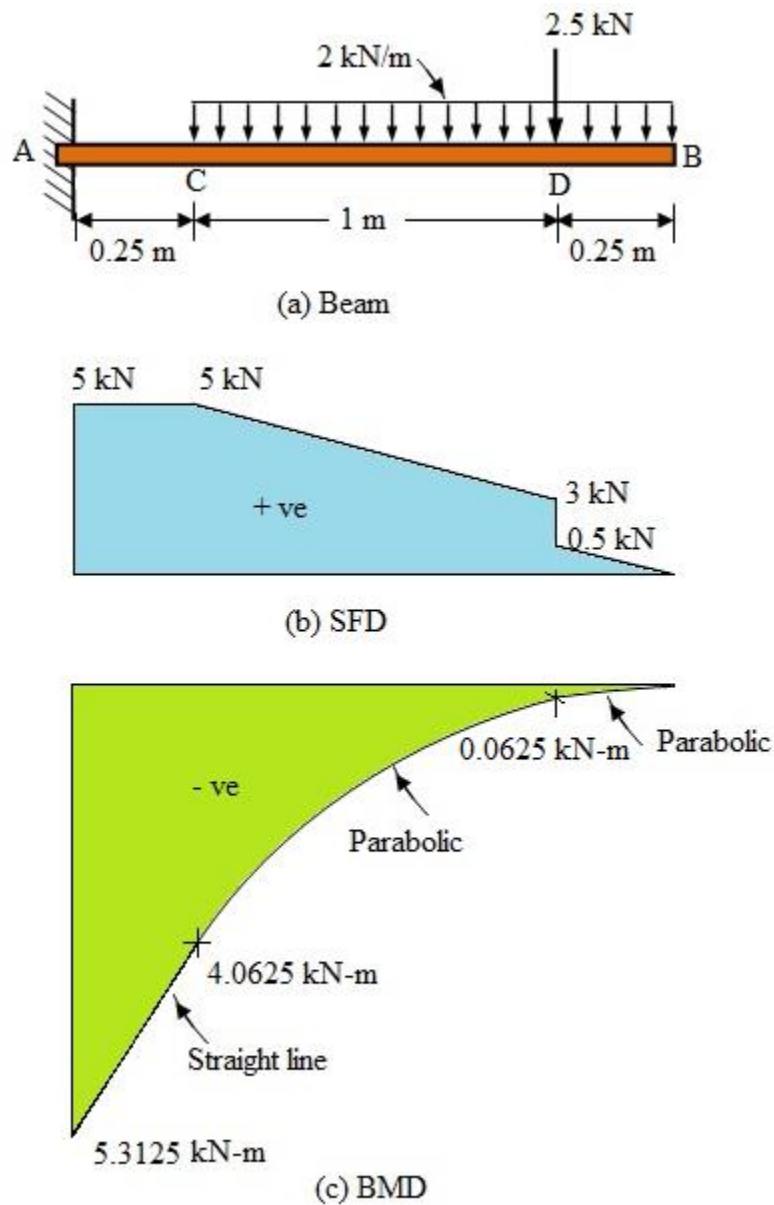


Fig.

## Shear Force and Bending Moment

Now, take a section 2-2 at a distance of  $x$  from  $B$  between  $D$  and  $C$  ( $0.25 \leq x \leq 1.25$ ).

$$\text{Shear force at 1-1, } F_x = 2x + 2.5$$

$$\text{Shear force at } D, \text{ i.e., } x = 0.25, F_B = 2 \times 0.25 + 2.5 = 3 \text{ kN}$$

$$\text{Shear force } C, \text{ i.e., } x = 1.25 \text{ m, } F_D = 2 \times 1.25 + 2.5 = 5 \text{ kN}$$

$$\begin{aligned} \text{Bending moment at 1-1, } M_x &= -2x \left( \frac{x}{2} \right) - 2.5(x - 0.25) \\ &= -x^2 - 2.5x + 0.625 \end{aligned}$$

$$\begin{aligned} \text{Bending moment at } C, \text{ i.e., } x = 1.25, M_C &= -1.25^2 - 2.5 \times 1.25 + 0.625 \\ &= -4.0625 \text{ kN-m} \end{aligned}$$

8. Calculate the shear force and bending moment for the beam subjected to a concentrated load of  $W$  as shown in the figure. Draw the shear force diagram (SFD) and bending moment diagram (BMD).

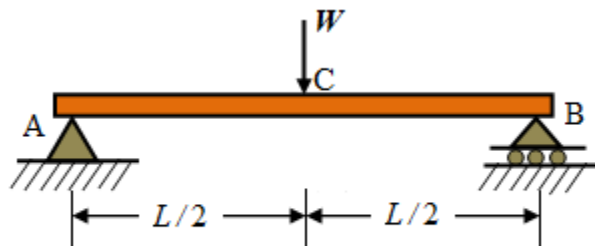


Fig.

**Solution:**

**Evaluation of support reactions:**

Considering the equilibrium of the beam and applying static equations of equilibrium,

$$\text{Taking moment about } B, \sum M_B = 0, V_A \times L = W \times \frac{L}{2}$$

$$V_A = \frac{W}{2}$$

$$\text{Sum of the vertical forces, } \sum V = 0, V_A + V_B = W$$

$$\text{Hence, } V_B = \frac{W}{2}$$

**Calculation of Shear force and bending moments:**

## Shear Force and Bending Moment

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Considering A as the origin, take a section 1-1 at a distance of  $x$  from A between A and C ( $0 < x < L/2$ ).

$$\text{Shear force at 1-1, } F_x = V_A = \frac{W}{2}$$

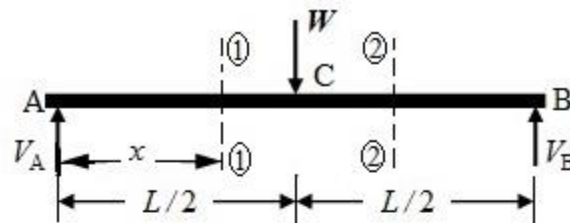
$$\text{Shear force at A, i.e., } x = 0 \quad F_A = \frac{W}{2}$$

$$\text{Shear force just left of C, i.e., } x = 0 \text{ i.e., } x = \frac{L}{2}, \quad F_{LC} = \frac{W}{2}$$

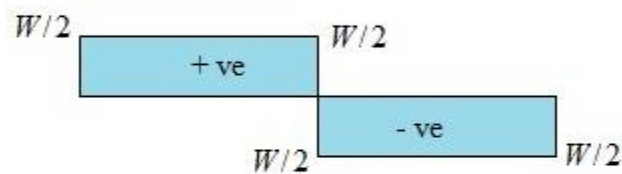
$$\text{Bending moment at 1-1, } M_x = V_A \times x = \frac{W}{2} x$$

$$\text{Bending moment at A, i.e., } x = 0, \quad M_A = \frac{Wx}{2} = 0$$

$$\text{Bending moment at C, i.e., } x = \frac{L}{2}, \quad M_C = \frac{W}{2} \times \frac{L}{2} = \frac{WL}{4}$$



(a) Beam



(b) Shear Force Diagram



(c) Bending Moment Diagram

Fig.

## Shear Force and Bending Moment

---

Take a section 2-2 at a distance of  $x$  from A between C and B  $\left(\frac{L}{2} \leq x \leq L\right)$ .

$$\text{Shear force at 2-2, } F_x = \frac{W}{2} - W = -\frac{W}{2}$$

$$\text{Shear force at C } \left(x = \frac{L}{2}\right), F_C = -\frac{W}{2}$$

$$\text{Shear force at B, } (x = L), F_B = -\frac{W}{2}$$

$$\begin{aligned} \text{Bending moment at 2-2, } M_x &= \frac{W}{2}x - W\left(x - \frac{L}{2}\right) \\ &= \frac{Wx}{2} - Wx + \frac{WL}{2} \\ &= -\frac{Wx}{2} + \frac{WL}{2} \end{aligned}$$

$$\text{Bending moment at B, i.e., } x = L, M_B = -\frac{WL}{2} + \frac{WL}{2} = 0$$

$$\text{Bending moment at C, i.e., } x = \frac{L}{2}, M_C = -\frac{W}{2}\left(\frac{L}{2}\right) + \frac{WL}{2} = \frac{WL}{4}$$

9. Draw the Shear force and bending moment diagram for a simply supported beam of length  $L$  carrying uniformly distributed load of intensity  $w$  per unit length over the entire span.

**Solution:**

**Evaluation of support reactions:**

The simply supported beam with uniformly distributed load over the entire span is symmetrically loaded symmetric beam. Hence, reactions at both supports are equal.

$$R_A = R_B = \frac{wL}{2}$$

**Calculation of Shear force and bending moments:**

In a symmetric beam, we need only to analyze half of the beam for shear force and bending moment. The other half will just be the mirror-image of the first half.



## Shear Force and Bending Moment

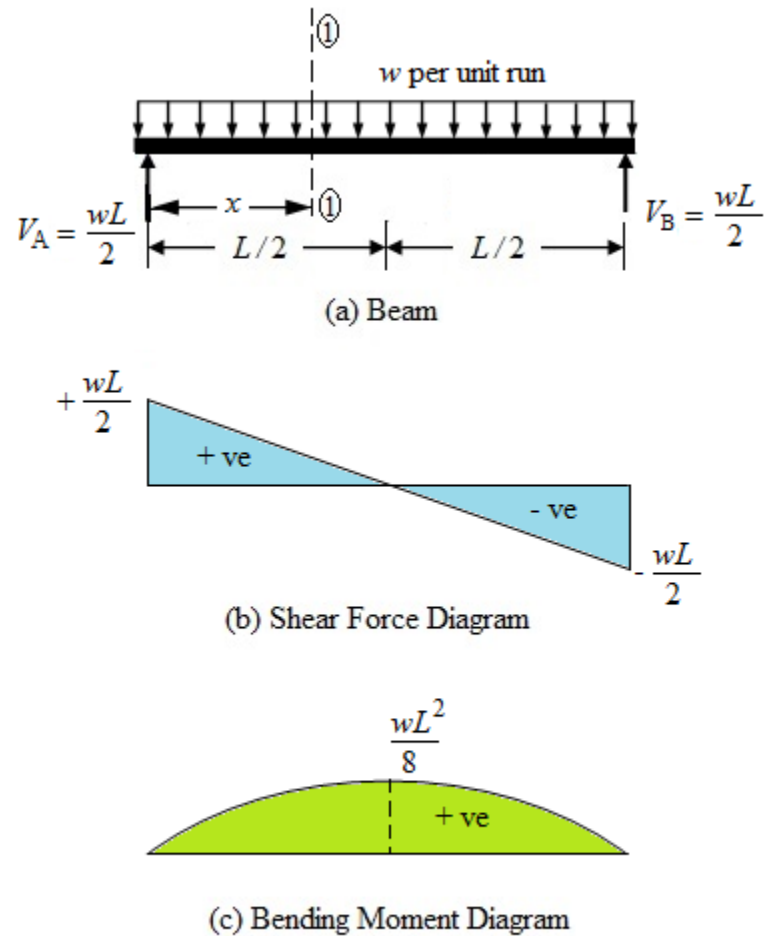


Fig.

Considering A as the origin, take a section 1-1 at a distance of  $x$  from A between A and C ( $0 \leq x \leq L/2$ ).

$$\text{Shear force at 1-1, } V_x = V_A - wx = \frac{wL}{2} - wx$$

$$\text{Shear force at A, i.e., } x = 0 \quad F_A = \frac{wl}{2}$$

$$\text{Shear force at C, i.e., } F_C = \frac{wL}{2} - w \times \frac{L}{2} = 0$$

$$\begin{aligned} \text{Bending moment at 1-1, } M_x &= V_A \times x - wx \left( \frac{x}{2} \right) = \frac{wL}{2} x - wx \left( \frac{x}{2} \right) \\ &= \frac{wL}{2} x - \frac{wx^2}{2} \end{aligned}$$

## Shear Force and Bending Moment

---

Bending moment at A, i.e.,  $x = 0$   $M_x = 0$

$$\begin{aligned}\text{Bending moment at C, } M_C &= \frac{wL}{2} \left( \frac{L}{2} \right) - \frac{w \left( \frac{L}{2} \right)^2}{2} = \frac{wL^2}{4} - \frac{wL^2}{8} \\ &= \frac{wL^2}{8}\end{aligned}$$

Bending moment equation is a quadratic in form, hence the bending moment diagram will be parabolic between A and B.

Due to symmetry, the bending moment and shear force for the other half at respective point of symmetry will be same as the first half AB.

10. A simply supported beam shown in the figure carries two concentrated loads and a uniformly distribute load. Analyze the beam for shear force and bending moment, and draw the SFD and BMD.

**Solution:**

**Evaluation of support reactions:**

Considering the equilibrium of the beam and applying static equations of equilibrium,

Taking moment about B,  $\sum M_B = 0$ ,  $V_A \times 8 = 25 \times 6 + 15 \times 4 + 7.5 \times 4 \times 2$

$$V_A = 33.75 \text{ kN}$$

Sum of the vertical forces,  $\sum V = 0$ ,  $33.75 + V_B = 25 + 15 + 7.5 \times 4$

Hence,  $V_B = 70 - 33.75 = 36.25 \text{ kN}$

**Calculation of Shear force and bending moments:**

Considering A as the origin, take a section 1-1 at a distance of  $x$  from A between A and C ( $0 \leq x \leq 2$ ).

Shear force at 1-1,  $F_x = V_A = 33.75 \text{ kN}$

Shear force at A, i.e.,  $x = 0$   $F_A = 33.75 \text{ kN}$

Shear force just left of C, i.e.,  $x = 2$ ,  $F_{LC} = 33.75 \text{ kN}$

Shear force at C, i.e.,  $x = 2$ ,  $F_C = 33.75 - 25 = 8.75 \text{ kN}$

Bending moment at 1-1,  $M_x = V_A \times x = 33.75x$

## Shear Force and Bending Moment

Bending moment at  $C$ , i.e.,  $x = 2$ ,  $M_C = 33.75 \times 2 = 67.5 \text{ kN-m}$

Take a section 2-2 at a distance of  $x$  from  $A$  between  $C$  and  $D$  ( $2 \leq x \leq 4$ ).

Shear force at 2-2,  $F_x = 33.75 - 25 = 8.75 \text{ kN}$

Shear force at  $C$ , i.e.,  $x = 2$ ,  $F_C = 8.75 \text{ kN}$

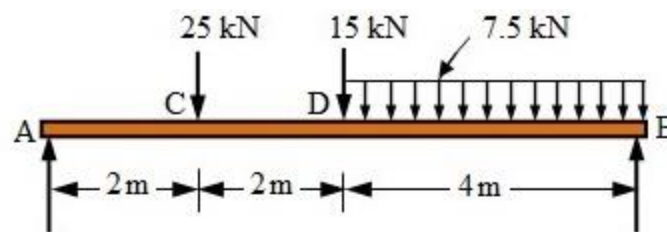
Shear force just left of  $D$ , i.e.,  $x = 4$ ,  $F_{D^-} = 8.75 \text{ kN}$

Shear force at  $D$ , i.e.,  $x = 4$ ,  $F_D = 8.75 - 15 = -6.25 \text{ kN}$

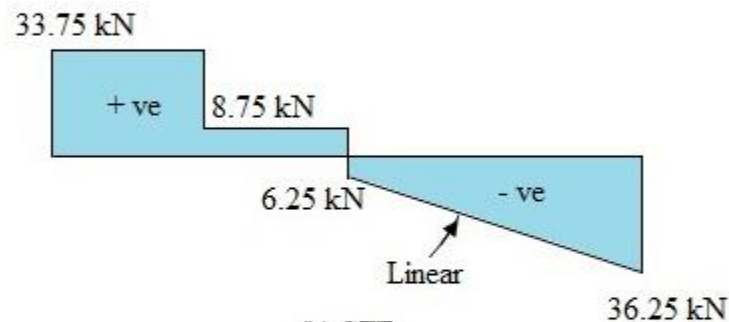
Bending moment at 2-2,  $M_x = 33.75x - 25(x - 2)$

$$M_x = 8.75x + 50$$

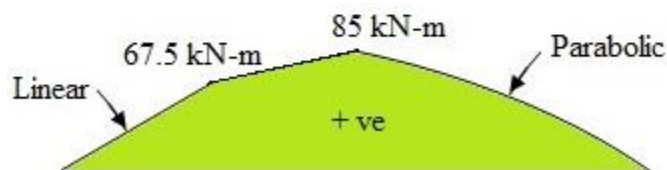
Bending moment at  $C$ , i.e.,  $x = 4$ ,  $M_D = 8.75 \times 4 + 50 = 85 \text{ kN-m}$



(a) Beam



(b) SFD



(c) BMD

Fig.

## Shear Force and Bending Moment

---

Now, considering from the right side and taking  $B$  as the origin, take a section 3-3 at a distance of  $x$  from  $B$  between  $B$  and  $D$  ( $0 \leq x \leq 4$ ).

$$\text{Shear force at 3-3, } F_x = -36.25 + 7.5x$$

$$\text{Shear force at } B, \text{ i.e., } x = 0 \quad F_B = 36.25 \text{ kN}$$

$$\text{Shear force just right of } D, \text{ i.e., at } x = 4, \quad F_x = -36.25 + 7.5 \times 4 = -6.25 \text{ kN}$$

$$\text{Shear force at } D, \text{ i.e., } x = 4, \quad F_D = -6.25 + 15 = 8.75 \text{ kN}$$

$$\text{Bending moment at 3-3, } M_x = 36.25x - \frac{7.5}{2}x^2 = 36.25x - 3.75x^2$$

$$\text{Bending moment at } D, \text{ i.e., } x = 4, \quad M_D = 36.25 \times 4 - 3.75 \times 4^2 = 85 \text{ kN-m}$$

11. Draw the shear force and bending moment diagram for the overhanging beam shown in the figure.

**Solution:**

**Evaluation of support reactions:**

Considering the equilibrium of the beam and applying static equations of equilibrium,

$$\text{Taking moment about } A, \sum M_A = 0, \quad V_D \times 4 = 20 \times 5 + 50 \times 2 + 20 \times 2 \times 1$$

$$V_D = 60 \text{ kN}$$

$$\text{Sum of the vertical forces, } \sum V = 0, \quad V_A + 60 = 20 \times 2 + 50 + 20$$

$$\text{Hence, } \quad V_A = 110 - 60 = 50 \text{ kN}$$

**Calculation of Shear force and bending moments:**

Considering  $A$  as the origin, take a section 1-1 at a distance of  $x$  from  $A$  between  $A$  and  $C$  ( $0 \leq x \leq 2$ ).

$$\text{Shear force at 1-1, } F_x = 50 - 20x$$

$$\text{Shear force at } A, \text{ i.e., } x = 0 \quad F_A = 50 \text{ kN}$$

$$\text{Shear force just left of } C, \text{ i.e., } x = 2, \quad F_{LC} = 50 - 20 \times 2$$

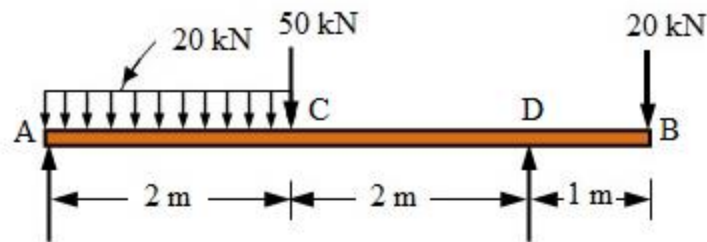
$$F_{LC} = 10 \text{ kN}$$

$$\text{Shear force at } C, \text{ i.e., } x = 2, \quad F_C = 10 - 50 = -40 \text{ kN}$$

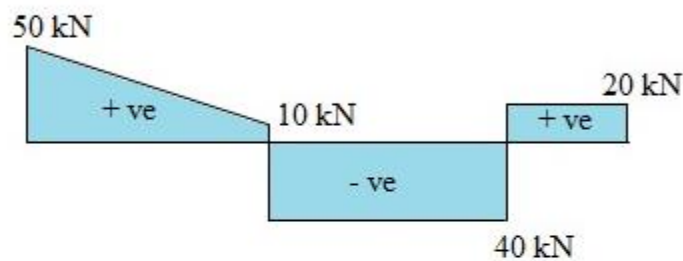
## Shear Force and Bending Moment

$$\begin{aligned} \text{Bending moment at 1-1, } M_x &= V_A \times x - 20 \times \frac{x^2}{2} \\ &= 50x - 10x^2 \end{aligned}$$

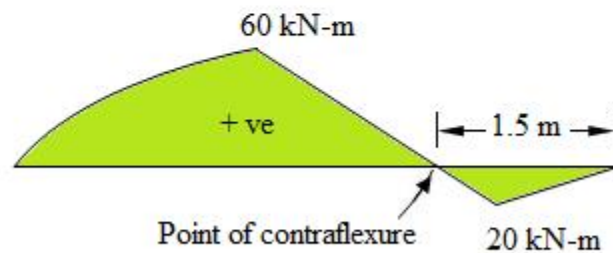
$$\text{Bending moment at C, i.e., } x = 2, M_C = 50 \times 2 - 10 \times 2^2 = 60 \text{ kN-m}$$



(a) Beam



(b) SFD



(c) BMD

Fig.

Now, considering from the right side and taking  $B$  as the origin, take a section 2-2 at a distance of  $x$  from  $B$  between  $B$  and  $D$  ( $0 \leq x \leq 4$ ).

$$\text{Shear force at 2-2, } F_x = 20 \text{ kN}$$

$$\text{Shear force at B, i.e., } x = 0 \quad F_B = 20 \text{ kN}$$

## Shear Force and Bending Moment

---

Shear force just right of  $D$ , i.e., at  $x = 1$ ,  $F_D = 20\text{ kN}$

Shear force at  $D$ , i.e.,  $x = 1$ ,  $F_D = 20 - 60 = -40\text{ kN}$

Bending moment at 2-2,  $M_x = -20x$

Bending moment at  $D$ , i.e.,  $x = 1$ ,  $M_B = 0$

Bending moment at  $D$ , i.e.,  $M_D = -20 \times 1 = -20\text{ kN-m}$

Take a section 3-3 at a distance of  $x$  from  $B$  between  $D$  and  $C$  ( $1 \leq x \leq 3$ ).

Shear force at 3-3,  $F_x = 20 - 60 = -40\text{ kN}$

Shear force at  $D$ , i.e.,  $x = 1$ ,  $F_D = -40\text{ kN}$

Bending moment at 3-3,  $M_x = -20x + 60(x - 1)$   
 $= 40x - 60$

Bending moment at  $C$ , i.e.,  $x = 3$ ,  $M_x = 40 \times 3 - 60 = 60\text{ kN-m}$

It is observed that bending moment changes sign between  $D$  and  $C$ . So, point of contraflexure exists between  $D$  and  $C$ .

Equating bending moment equation to zero, we get

$$40x - 60 = 0$$

$$x = 1.5\text{ m}$$

### Point of contraflexure:

A point of contraflexure is a point where the curvature of the beam changes signs. It is sometimes referred to as a *point of inflexion*. In other words, point of contraflexure is a point where bending moment changes its sign from positive to negative or from negative to positive through zero. This means, bending moment is zero at point of contraflexure.